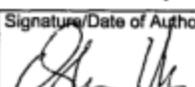
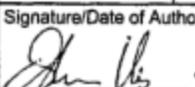


 US Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		OMB No. 2120-0020 Exp: 5/31/2018	Electronic Tracking Number
				For FAA Use Only
INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))				
1. Aircraft	Nationality and Registration Mark N727KB		Serial No. RK-260	
	Make Beech		Model 400A	Series
2. Owner	Name (As shown on registration certificate) PUMPJACK AVIATION LLC		Address (As shown on registration certificate) Address 1511 W 60th St. City Casper State WY Zip 82601-6203 Country USA	
	3. For FAA Use Only			
3. For FAA Use Only				
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	<u>Beech</u>	<i>(As described in Item 1 above)</i>
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name <u>Shaun Likens at Clemens Aviation</u> Address <u>14116 SW 32nd St</u> City <u>Benton</u> State <u>KS</u> Zip <u>67017</u> Country <u>USA</u>		<input checked="" type="checkbox"/> U. S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		
		C. Certificate No. <div style="font-size: 24pt; font-weight: bold; text-align: center;">3437625</div>		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual  8-14-2020		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> Approved <input type="checkbox"/> Rejected				
BY	FAA Fit. Standards Inspector	Manufacturer	Maintenance Organization	Persons Approved by Canadian Department of Transport
	FAA Designee	Repair Station	<input checked="" type="checkbox"/> Inspection Authorization	Other (Specify)
Certificate or Designation No. 3437625		Signature/Date of Authorized Individual  8-14-2020		

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N727KB

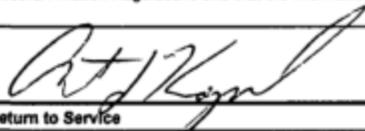
8/14/20

Nationality and Registration Mark

Date

Installed copilot windshield P/N: LA45AS31001-024, S/N: 10013 under STC no. ST018935WI.
Weight and balance revision no factor.

Additional Sheets Are Attached

 U.S. Department of Transportation Federal Aviation Administration		MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
For FAA Use Only					
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N727KB		Serial No RK-260		
	Make Raytheon Aircraft Company		Model 400A	Series	
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) _____ PKWY City _____ State OH Zip 44_____ USA		
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address Name _____ Address _____ City _____ State OH Zip _____			B. Kind of Agency <input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		
			<input type="checkbox"/> Manufacturer C. Certificate No. CRS # WC7R346J		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual Anton J. Koprivnik		 02/24/2020	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	<input type="checkbox"/> FAA Fit Standards Inspector	<input type="checkbox"/> Manufacturer	<input type="checkbox"/> Maintenance Organization	<input type="checkbox"/> Person Approved by Canadian Department of Transport	
	<input checked="" type="checkbox"/> FAA Designee	<input checked="" type="checkbox"/> Repair Station	<input type="checkbox"/> Inspection Authorization	Other (Specify) _____	
Certificate or Designation No. CRS # WC7R346J		Signature/Date of Authorized Individual Anton J. Koprivnik		 02/24/2020	

FAA Form 337 (10-06)

SDNY_GM_02758012

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244934

EFTA01261594

SDNY_GM_02758013

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244935

EFTA01261595

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N727KB

Nationality and Registration Mark

02/24/2020

Date

This 337 is to correct the original Dated 01/04/2013. Corrected block # 1, Registration number from N746TA to N787TA and fills in the serial number to RK-260. All other data is correct.

Part 1 of 3

The Following Equipment was removed along with Associated Hardware.

EFD-871	622-9345-203	Electronic Flight Display
EFD-871	622-9345-203	Electronic Flight Display
MFD-871	622-9434-213	Multi Function Display
ICU-85	622-6189-002	Internal Compensation Unit
ICU-85	622-6189-002	Internal Compensation Unit
AHC-85E	622-9336-400	Attitude/Heading Computer
AHC-85E	622-9336-400	Attitude/Heading Computer
SDU-640B	622-9735-001	Sensor Display Unit
SDU-640B	622-9735-001	Sensor Display Unit
SDU-640B	622-9735-001	Sensor Display Unit
DCP-5000	822-1028-011	Display Control Panel
DCP-5000	822-1028-011	Display Control Panel
ARP-851	622-9500-011	Altitude Reference Panel
ARP-851	622-9500-011	Altitude Reference Panel
AAP-851	822-0328-011	Altitude Awareness Panel
AAP-851	822-0328-011	Altitude Awareness Panel
CHP-850	622-7397-002	Course Heading Panel
DBU-4100	822-0014-002	Data Base Unit
DAU-650	622-9344-101	Data Acquisition Unit
SDD-640A	622-9347-001	Sensor Display Driver
IOC-4000 (4)	622-9814-514	Card Assembly
MDC-4000	622-9818-751	Maintenance Computer
CSU-4000 (2)	822-0049-002	Configuration Strapping Unit
FMC-5000 (2)	822-0891-008	Flight Management Computer
PWR-4000 (4)	622-9945-021	IAPS Power Card
GPS-4000	822-0931-003	GPS Receiver
Mounts	622-9430-002	EFD /MFD Mounts
EFB		Electronic Flight Bag Mounting

Additional Sheets Are Attached

Form 337 (10-06)

SDNY_GM_02758014

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244936

EFTA01261596

SDNY_GM_02758015

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244937

EFTA01261597

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N727KB

02/24/2020

Nationality and Registration Mark

Date

Page 2 of 3

Installed the Following Equipment along with Associated Hardware.

Model	Part Number	Description	Weight	Arm
CCP-3000	822-1746-002	Cursor Control Panel	2.60	90.0
CCP-3000	822-1746-002	Cursor Control Panel	2.60	90.0
DCU-3001C	822-2362-003	Data Concentrator Unit	5.0	325
DCU-3001C	822-2362-003	Data Concentrator Unit	5.0	325
ECU-3000	822-1200-209	External Compensation Unit	0.40	57.5
ECU-3000	822-1200-209	External Compensation Unit	0.40	57.5
FSU-5010	822-1543-101	File Server Unit (1 st IFIS)	6.5	60.83
ECU-3000	822-1200-998	File Server Unit (1st IFIS)	0.40	57.5
AFD-3010E	822-1753-416	Adaptive Flight Display	12.90	89.2
AFD-3010E	822-1753-416	Adaptive Flight Display	12.90	89.2
AFD-3010E	822-1753-416	Adaptive Flight Display	12.90	89.2
AFD-3010E	822-1753-416	Adaptive Flight Display	12.90	89.2
IMT-3010	822-1140-401	AFD Mounting Rack	4.40	85.0
IMT-3010	822-1140-401	AFD Mounting Rack	4.40	85.0
IMT-3010	822-1140-401	AFD Mounting Rack	4.40	85.0
IMT-3010	822-1140-401	AFD Mounting Rack	4.40	85.0
	10852B01Y00	Fuel Quantity Conditioner	0.90	324
	PC920-2A2000PH-2A1	Fuel Flow Conditioner	1.10	325
GPS-4000S	822-2189-004	GPS	6.0	37
	CI 429-200	GPS/XW Antenna	1.0	166.14
Amp Cluster	N241100-002	DC Voltmeter	1.5	97.4
	373-93-1501-1	Speed Switch x2	0.25	89.2
DBU-5010E	822-2215-202	Data Base Unit	1.60	134.5
PS-835D	501-1228-004	Power Supply	12.5	52.9
MMT-5000	822-1811-003	File Server Unit Rack	0.30	57.5
MMT-3010	822-1290-003	AHS Mounting Rack	1.80	57.5
MMT-3010	822-1290-003	AHS Mounting Rack	1.80	57.5
CHP-3010	822-1280-002	Course Heading Panel	1.30	92.5
OCM-4100	822-1463-228	Options Configuration Module WAAS/LPV	0.30	57.5
OCM-4100	822-1463-228	Options Configuration Module WAAS/LPV	0.30	57.5

 Additional Sheets Are Attached

Form 337 (10-06)

SDNY_GM_02758016

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244938

EFTA01261598

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N727KB
Nationality and Registration Mark

02/24/2020
Date

Page 3 of 3

Model	Part Number	Description	Weight	Arm
CSU-4100 (2)	822-1364-002	Configuration Strapping Unit	1.60	55.7
MDC-3110	822-1987-006	Maintenance Data Computer	0.80	55.7
FMC-6000 (2)	822-0868-123	Flight Management Computer	1.90	55.7
ADC-850D (2)	822-0389-468	Air Data Computer	5.30	60.83
AHC-3000A (2)	822-1378-001	Attitude Heading Computer	4.80	57.5
DCP-3030 (2)	822-1828-002	Display Control Panel	1.90	89.96
PWR-4000 (4)	622-9945-022	IAPS Power Card	1.3	55.0
IOC-4100 (4)	822-1362-511	Input/output Concentrator	0.8	55.0
XM WR-1000	822-2031-002	XM Weather	41.0	
	LT-4001-010	Master Caution Panel	1.3	87.0
Rev. Panel	373-91-3203-1	Pilot Reversion Panel	0.2	88.0
Rev. Panel	373-91-3203-3	Co-pilot Reversion Panel	0.2	88.0
9-PED	373-91-3203-9	9-Pedestal Panel	1.8	92.89
	345-6169	ELT Switch	0.2	88.0
Annunciator	2-F840233-I	AUX BATT (1/2) Annunciator	0.05	88.0
Annunciator (2)	2-F840232-I	Pull Up/ Terrain Annunciator	0.05	88.0
Annunciator	2-F840231-I	TERR INHBT/ TERR Fail	0.05	88.0

In Accordance with STC ST10959SC. See Attached Copy AFM Supplement, Nextant Aerospace Doc. No. 373-00-0023 Rev IR Dated 09/02/2011 or later FAA approved revision, installed on Aircraft.

Equipment Pilot Guides are Installed in the Aircraft as Portable Document Format (.pdf) via supplied I-Pad.
NOTE: Portable electronic devices used to store required aircraft records required by Part 91.9 are considered Electronic Flight Bags (EFB) per FAA Advisory Circular AC 120-76B par 4-g. Commercial operators are required to obtain authorization for their use from their managing Flight Standards District Office.

Instructions for Continued Airworthiness (ICA) Ref Nextant Aerospace Document 373-00-0002.Rev C or Later are included in the Aircraft Maintenance Records
Carried out System Ground Test Plan IAW Master Drawing List (MDL) 373-00-004 Rev C.

Carried out Hawker Beechcraft Service Bulletin 34-3431 Reduced Vertical Separation Minimum (RVSM) Airframe Inspection / Modification. Performed FAR 91.411 Inspection per Part 43 appendix "E" Pitot Static & FAR 91.413 Inspection per Part 43 appendix "F" Transponder System checks.

Aircraft Weighed see Report Dated 12/21/2012

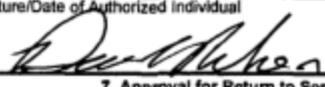
Additional Sheets Are Attached

SDNY_GM_02758019

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244941

EFTA01261601

 US Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 2/28/2011	Electronic Tracking Number
				For FAA Use Only
INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))				
1. Aircraft	Nationality and Registration Mark USA N727KB		Serial No. RK-260	
	Make Raytheon Aircraft Company		Model 400A	Series Beachief
2. Owner	Name (As shown on registration certificate) XT Leasing CO. LLC		Address [Redacted] City State <u>MN</u> Zip	
	3. For FAA Use Only			
The data identified herein complies with the applicable airworthiness requirements and is approved for the above described aircraft, subject to conformity inspection by a person authorized in 14CFR 43.7 <div style="text-align: right;"> DALE E DRENCKHAHN <small>Digitally signed by DALE E DRENCKHAHN Date: 2018.05.11 13:08:20 EDT</small> FAA Inspector MSP-FSDO-GL15 </div>				
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type Manufacturer	
6. Conformity Statement				
B. Kind of Agency <input type="checkbox"/> U. S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization				
C. Certificate No. KX5R005N				
I certify that the repair and/or alteration made to the unit(s) identified in Item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual  5-11-2018		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in Item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> Approved <input type="checkbox"/> Rejected				
BY	FAA Fit. Standards Inspector	Manufacturer	Maintenance Organization	Persons Approved by Canadian Department of Transport
	FAA Designee <input checked="" type="checkbox"/>	Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. KX5R005N		Signature/Date of Authorized Individual  5-11-2018		

FAA Form 337 (10-06)

SDNY_GM_02758020

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244942

EFTA01261602

SDNY_GM_02758021

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244943

EFTA01261603

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N7 27 KB

5-11-2018

Nationality and Registration Mark

Date

1. Installed Garmin Automatic Dependant Surveillance-Broadcast (ADS-B) system referencing Elliott Aviation Technical Products Development Inc. drawings 121-3761-M0560-60 revision IR, and 121-4070-E0560-000 revision PA. A satisfactory ground test of the ADS-B Out system was completed using a Aeroflex IFR-6000, checking for proper configuration and equipment performance requirements. The installed ADS-B Out system has been shown to meet the equipment requirements of 14CFR 91.227.

2. The following LRU's are installed and interfaced referencing the previously approved equipment pairing of the GTX-3000 Mode "S" transponders and the GDL-88 Universal Access Transceiver by STC ST04214CH.

a) Removed existing Rockwell Collins TDR-94D Mode "S" Transponders p/n 622-9210-004, s/n's 3MPD, and 6053 from right hand upper avionics shelf at sta. 48.

Installed two Garmin GTX-3000 Mode "S" Transponders (TSO-C112d), p/n 011-01997-00, on right hand upper avionics shelf, s/n 20S008042 at sta. 49.7, and s/n 20S008050 at sta. 52.95.

b) Installed a Garmin GDL-88 Universal Access Transceiver (TSO-C154c), p/n 010-00860-30, s/n 22U000700 on right hand upper avionics shelf at sta. 46.95. The GDL-88 is connected to a GA-36 Antenna (TSO-C144), p/n 013-00244-00, s/n 14145 at sta. 128.7 through the GPS port and a UAT Antenna (TSO-C66c) p/n C1110-40-30, s/n 522354 at sta. 35.1 for the FIS-B weather function.

c) Installed a Garmin Flight Stream 110 (TSO-C157), p/n 010-01194-01, s/n 3G0000731 on the co-pilots manual holder at sta. 136, interfaced to the GDL-88 UAT. The Flight Stream 110 provides FIS-B weather, ADS-B traffic, and GPS position data.

3. The FAA Approved Airplane Flight Manual Supplement for the Garmin Automatic Dependant Surveillance (ADS-B) Transponder installation Report Number FMS-4079-560 Revision IR, dated was placed in the Aircraft Flight Manual and must be available to the pilot at all times.

4. Structural work performed referencing the following: Elliott Aviation Technical Products Development Inc. drawing 121-3761-M0560-60 revision IR, Beechcraft Corporation Model 400/400A Maintenance Manual p/n 128-590001-9 Revision C29, Beechcraft Corporation Structural Repair Manual p/n 128-590001-17 Revision C16, and acceptable methods contained in AC43.13-1B, chapter 4, section 4, and AC43.13-2B, chapters 1,2,and 3.

5. The electrical interface of the Garmin Automatic Dependant Surveillance (ADS-B) Transponders was performed referencing Elliott Aviation Technical Products Development Inc. drawing 121-4070-E0560-000 revision PA. Wiring fabrication and interface performed in accordance with Beechcraft Corporation Beechjet 400/400A series wiring manual p/n 128-590001-79 Revision D, and acceptable practices contained in AC43.13-1B, change 1, chapter 11.

6. Weight and balance and equipment list have been revised to reflect this installation.

7. Instructions for Continued Airworthiness, Elliott Aviation Technical Products Development Inc. Report ICA-4079-560 Revision IR, provided in aircraft records.

END

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758022

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244944

EFTA01261604

SDNY_GM_02758023

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244945

EFTA01261605

Elliott Aviation Technical Products Development, Inc.
6601 74th Ave Building A
Milan, IL 61264



Report No.: ICA-4079-560
Revision: IR
Dated 05/01/2018
1 of 19

Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Instructions for Continued Airworthiness

For

Garmin ADS-B Out System Installation

(Type Certificate A16SW)

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SDNY_GM_02758024

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244946

EFTA01261606

SDNY_GM_02758025

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244947

EFTA01261607

Elliott Aviation Technical Products Development, Inc.
6601 74th Ave Building A
Milan, IL 61264



Report No.: ICA-4079-560
Revision: IR
Dated 05/01/2018
2 of 19

Instructions for Continued Airworthiness For Garmin ADS-B System Installation

REVISIONS PAGE

Rev	Description	Page (S)	Date	Approved By
IR	Initial Release	ALL		

SDNY_GM_02758026

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244948

EFTA01261608

1

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758027

EFTA_00244949

EFTA01261609



Instructions for Continued Airworthiness For Garmin ADS-B System Installation

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SDNY_GM_02758028

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244950

EFTA01261610

SDNY_GM_02758029

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00244951

EFTA01261611



Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Chapter 1 INTRODUCTION

1.1 Scope/Purpose/Applicability/Distribution

The Instructions for Continued Airworthiness (ICA) described herein are applicable only to RK-260 modified with Garmin ADS-B System. These ICA describe the recommended and required maintenance procedures for the installation of Garmin ADS-B System.

The maintenance information provided by this document must be included in the operator's Aircraft Maintenance Manual and the operator's Aircraft Scheduled Maintenance Program. All documents listed in the required documents section must be maintained with this document. FAA accepted revisions to this document will be distributed by mail to the applicable certifying authorities and operators of legally modified aircraft.

1.2 System Description

The features of this modification include the removal of both Collins TDR-94D transponders and replacement with Garmin GTX 3000 transponders, reusing the existing aircraft wiring. This modification also includes a Garmin GDL 88 GPS / UAT and optional Flight Stream 110 Bluetooth FIS-B installation. A GPS and UAT-SBS antenna is installed. Annunciation is added for ADS-B status. New circuit breakers are installed in the copilot's circuit breaker panel. New wiring is added to supply the GTX 3000 transponders with GPS position for the GDL 88. The Flight Stream 110 (if installed) supports properly equipped FIS-B capable Bluetooth devices

1.3 Required Data

All required data is located in Appendix A of this document.

1.4 Referenced Garmin documents:

Nomenclature	Drawing Number	Rev	Date
GTX 3000 Transponder Installation Manual	190-00926-01	L	Oct 2015
GDL 84/88 TSO Installation Manual	190-01122-00	H	Dec 2014
Flight Stream 110/210 TSO Installation Manual	190-01700-00	E	Dec 2015

1.5 Systems Control:

1.5.1 The GTX 3000 transponders replace the Collins TDR-94D transponders. All control is via the existing Collins Pro Line 21 cockpit Control Display Units. No changes to operational control are imposed by this STC.

1.5.2 The GDL 88 is a GPS sensor and FIS-B UAT receiver. The GDL 88 has no external control.

1.5.3 The Flight Stream 110 is a Bluetooth repeater for the GDL 88 FIS-B products and has no external control.

1.6 Document administration:

1.6.1 This document will be revised by Elliott Aviation Engineering when revisions to descriptive or approval data warrant.

1.6.2 When this document is revised, all pages will be revised to indicate the current level. Chapter 10 will have its own revision status embedded. FAA approved revisions to the document will be distributed by mail to the applicable certifying authorities and operations of legally modified aircraft.

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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

1.7 Abbreviations and Definitions:

1090ES	A transponder distinction. 1090 is the classic frequency of Secondary Surveillance Radar (SSR) transponders. The addition of "ES" at the end signifies that the transponder is equipped with Extended Squitter used by the ADS-B aircraft segment. Note that this class of transponder supports Class "A" airspace (altitudes above 18,000 feet) operations. Typically, no FIS-B or TIS-B functionality is included, compared to UAT.
AMM	Aircraft Maintenance Manual
DAH	Design Approval Holder. The FAA term that includes STC holders such as Elliott Aviation.
FIS-B	Flight Information System broadcast. A radio frequency service that transmits weather and aeronautical products, either by satellite or ground based systems. Refer to FAA Advisory Circular 20-149A for thorough explanation.
Latency	The total time between when the position is measured and when the position is transmitted by the aircraft. A 14 CFR §91.227 performance requirement for ADS-B broadcast systems. The collection of specific equipment, interface and testing represented by this STC comprises an approved Latency
OEM	Original Equipment Manufacturer.
SBS	Surveillance and Broadcast Service. Provided by the FAA. Ground based transmission at a frequency of 978 MHz.
SDA	System Design Assurance. A 14 CFR §91.227 performance requirement for ADS-B broadcast systems. The collection of specific equipment, interface and testing represented by this STC comprises an approved SDA.
UAT	Universal Access Transceiver. A new device that supports the ADS-B aircraft segment at lower altitudes (below 18,000 feet) Allows smaller aircraft that operate below Class "A" airspace a more

1.8 Precautions

Typical precautions are to be taken when accessing, removing, replacing, testing or inspecting these modifications. Specific precautions will be addressed in context throughout this document.

1.9 Units of measurement:

Non- metric. English units including Inches, feet, etc..

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Chapter 2 MAINTENANCE INSTRUCTIONS AND SCHEDULES:

2.1 Mandatory Replacements:

The GTX 3000 1090ES Transponder, GDL 88 GPS / UAT and optional Flight Stream 110 systems' electrical components and installation is "On Condition" as determined by routine surveillance and there is no mandatory replacement time for any aspect of this system.

Refer to 400A OEM AMM, Wiring Procedures and Repair – Description and Operations - chapter 20, for comprehensive wiring harness surveillance concerns.

See Chapter 10 Airworthiness Limitations for antenna installation inspections.

2.2 Recurring Maintenance:

2.2.1 Antenna Installations

Antenna installations require special inspection requirements and shall be maintained using Chapter 10 of this document.

2.2.2 Bulkhead Feedthrough Installation

The bulkhead feedthrough installation requires no special inspections requirements and shall be maintained using normal OEM maintenance manual procedures

2.2.3 Basic Mode S Transponder functions:

The installation of the GTX 3000 transponders must continue to meet the operational requirements of 14 CFR §§ 91.413, 91.215, and 91.217 and comply with the transponder system tests and inspections called out in 14 CFR § 43, appendix F.

The Collins Pro Line 21 altitude reporting equipment connected to the Garmin ADS-B transponders must comply with all applicable 14 CFR §§ 91.217, 91.411, and part 43 appendix E test and inspection requirements. If the altimetry system is compliant with the Reduced Vertical Separation Minimum (RVSM) standards, the requirements and tolerances stated in the approved RVSM maintenance program must be met. ADS-B installation does not alter these requirements.

2.2.4 ADS-B functions:

Compliance with aircraft equipage requirements imposed by the operational requirements of 14 CFR §§ 91.225 and 91.227 is achieved by these STC modifications. ADS-B Maintenance is "On Condition" as determined by routine surveillance during Mode S maintenance, by the flight crew as well the FAA Air Traffic Control system.

2.2.5 FIS-B functions:

This modification has an optional installation that supports ground based SBS FIS-B reception. UAT/FIS-B Maintenance is "On Condition".

2.3 Setup and Configuration in Service

2.3.1 Garmin GDL 88

2.3.1.1 Confirm Garmin GDL 88 line replaceable unit part number from the Elliott Aviation wiring diagram 121-3761-E0560-000 located in Appendix A.

2.3.1.2 Obtain Elliott Configuration Instructions located in Appendix A. Read Section 5 of Configuration Instructions before proceeding with reconfiguration of the GDL 88. Take special note of the specific instructions for laptop computer interface, software effectivity and maintenance record details.

The GDL-88 will not function properly until software setup and configuration is accomplished.

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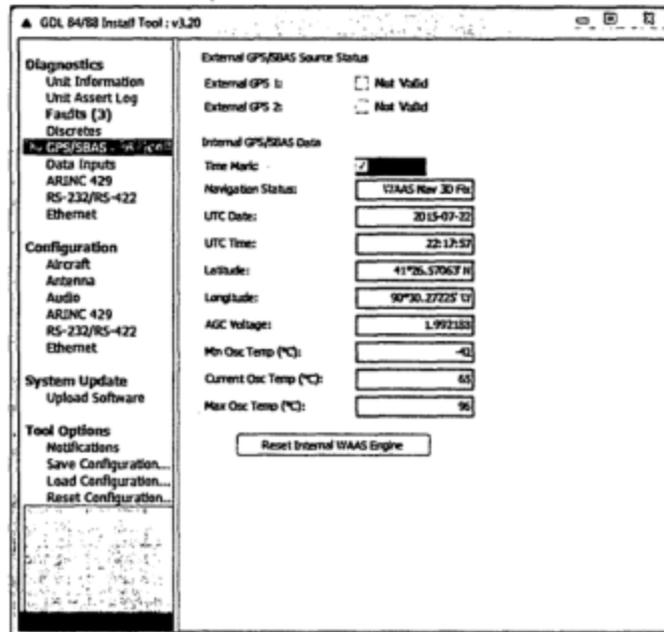
Instructions for Continued Airworthiness For Garmin ADS-B System Installation

PRECAUTION:

Part number, configuration and electrical interface has been evaluated and approved for ADS-B Latency and SDA requirements. Any deviations will require additional approval.

- 2.3.1.3 While using the Elliott Configuration Instructions Section 5, access to the GDL 88 Diagnostics section is available. Real time GPS information is presented.
- 2.3.1.4 For setup and configuration, GPS reception must be confirmed using the diagnostic page. Use known GPS position to validate the GDL 88 diagnostic position readout such as the aircrafts navigation GPS. This information is presented on the Pro Line 21 cockpit control display unit under the IDX pages. Sample page below show typical GDL 88 GPS data.
- 2.3.1.5 After GPS reception is verified aircraft maybe returned to service.
- 2.3.1.6 For troubleshooting an in-service system, GPS activity may be confirmed using this diagnostic page. Lack of GPS position may be due to disconnect antenna coax, faulty antenna or faulty GDL 88. Neither the GDL 88 nor GPS antenna is field serviceable. If faulty, the GDL 88 line replaceable unit may only be repaired by Garmin. Replace a faulty GPS antenna with new.

Sample page:



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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

2.3.2 GTX 3000 Transponder

- 2.3.2.1** Confirm Garmin GTX 3000 line replaceable unit part number from the Elliott Aviation wiring diagram 121-4079-E0560-000 located in Appendix A.
- 2.3.2.2** Obtain Elliott Configuration Instructions located in Appendix A. Read sections 2 and 4 of this document before proceeding with reconfiguring of the GTX 3000. Take special note of the requirements for laptop computer interface and software effectivity details.

The GTX 3000 will not function properly until software setup and configuration is accomplished.

PRECAUTION:

Part number, configuration and electrical interface has been evaluated and approved for ADS-B Latency and SDA requirements. Any deviations will require additional approval.

- 2.3.2.3** While using the Elliott Configuration Instructions, access to the GTX 3000 status is available. Real time information is presented.
- 2.3.2.4** For setup and configuration, the following items must be verified:
 - GPS Position
 - Magnetic
 - Weight on wheels is operating correctly.
 - ADS-B mode indicated.
 - 4096 code agrees with Pro Line 21 RTU ATC settings.
 - Flight ID agrees with Pro Line 21 RTU ATC settings. (note: Dynamic (pilot set) Flight ID not available on aircraft all.)
 - TCAS status correct for aircraft. Most aircraft will have TCAS installed.
- 2.3.2.5** GPS position is confirmed using the "Info / ADS-B Out" page. Confirm GPS position from the GDL 88. Sample page follows to show typical GTX 3000 GPS source information.
- 2.3.2.6** Heading is confirmed using "Info / EHS" page. Confirm heading from the Pro Line 21 system. Sample page follows to show typical GTX 3000 Heading source information.
- 2.3.2.7** WOW, ADS-B mode, 4096 Code, Flight ID, and TCAS status is confirmed using the "Status" page in the GTX 3000 configuration tool. Sample page follows to show typical GTX 3000 "Status" page information.
- 2.3.2.8** After these have been verified perform 14 CFR 43 App F ¶ A-J.
- 2.3.2.9** The aircraft may now be returned to service.

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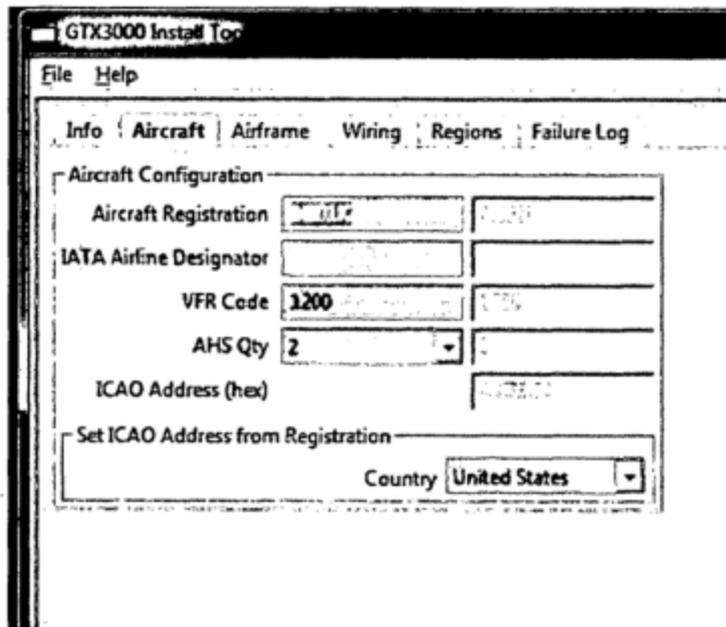
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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

2.3.3 GTX 3000 "N" number Registration change.

- 2.3.3.1** Obtain Elliott Configuration Instructions located in Appendix A. Using sections 2 and 4 of the Configuration Instructions Access the "Aircraft" tab on the GTX 3000 configuration tool. Modify the registration as needed and press enter to set the configuration. Shutdown the configuration tool and verify registration number change using IFR 6000 or similar transponder test set.

Sample page:



The screenshot shows the 'GTX3000 Install Tool' application window. The 'Aircraft' tab is selected in the top menu. The 'Aircraft Configuration' section contains the following fields:

Aircraft Registration	<input type="text" value="N1234"/>	<input type="text" value="1234"/>
IATA Airline Designator	<input type="text" value=""/>	<input type="text" value=""/>
VFR Code	<input type="text" value="1200"/>	<input type="text" value="1200"/>
AHS Qty	<input type="text" value="2"/>	<input type="text" value=""/>
ICAO Address (hex)	<input type="text" value=""/>	<input type="text" value=""/>
Set ICAO Address from Registration		<input type="checkbox"/>
Country	<input type="text" value="United States"/>	

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

2.4 GTX 3000 / GDL 88 / Pro Line 21 Troubleshooting

Reported Issue	Steps	
Transponder reporting: Altitude	Observe or determine Collins flight displays for air data indications.	If present and correct, GTX 3000 suspect wiring or configuration.
Transponder reporting: Heading	Observe or determine Collins flight displays for heading indications.	If present and correct, GTX 3000 suspect wiring or configuration.
Transponder reporting: ADS-B - Air	Observe or determine the ADS-B annunciator status.	If ADS-B annunciation OFF, suspect GTX 3000 in use. If ADS-B annunciation ON, suspect GDL 88 LRU, wiring or configuration.
Transponder reporting: ADS-B - Surface	Not seen by flight crew, report will come from ATC Ground.	Same as above for ADS-B annunciation status.
Transponder reporting: Mode C	Observe or determine the Collins CDU annunciations.	If alternate GTX 3000 can be controlled using Collins CDU, suspect the GTX 3000 LRU.
TCAS	Collins CDU: Navigate to ATC page and observe status.	Access the Collins MFD status page. TCAS faults with ATC faults point to transponder malfunction. TCAS faults alone point to TCAS.

Commercial Off The Shelf ADS-B receiver devices are suitable for GO / NO GO confirmation that the GTX 3000 transponders are transmitting GPS position. HOWEVER, only approved calibrated test devices, such as the IFR 6000, may be used to confirm transponder output power and minimum trigger levels as well as diversity isolation.

To conduct an in-depth as-installed complete functional check including UAT, qualified test equipment such as the IFR 6000 will be required. Refer to specific test set instructions for operation and testing.

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Chapter 3 ACCESS:

3.1 Overview:

The information contained herein supplements the basic Maintenance Manuals only in those areas listed, when the aircraft is modified in accordance with Elliott Aviation drawings and diagrams specified in Section 1.3 of this document.

- 3.1.1 GTX 3000s are located in the forward right avionics bay upper equipment shelf. A configuration module is also located inside the J3302 connectors.
- 3.1.2 GDL 88 is located in the forward left avionics bay lower equipment shelf. A configuration module is also located inside the J811 connector.
- 3.1.3 Flight Stream 110 (if installed) is located on the back of the Pilots chart holder cabinet.
- 3.1.4 ADS-B 1/2 annunciator is located in the cockpit panel
- 3.1.5 GA36 GPS Antenna is located on top of the aircraft fuselage above the cockpit headliner at FS 128.74 right of center line
- 3.1.6 UAT Antenna is located on the belly of the aircraft fuselage below the left forward avionics bay at FS 35.14
- 3.1.7 Wiring follows existing aircraft wire routes in the nose and cockpit area.

3.2 Equipment and wiring access:

No new access is required from the installation of this modification. Access to all equipment and wiring is gained using existing aircraft maintenance manual techniques.

3.2.1 GDL 88 and GTX 3000 rack mounted Line Replaceable Units (LRU):

- 3.2.1.1 No hand tools are required. The coax and wire harness connectors knurled and are intended to be removed and tightened by hand. Should assistance in the form of pliers be used, take care to not distort or over tighten.
- 3.2.1.2 Observe typical avionics systems precautions including:
 - Coax connectors. Do not bend or kink the coax, avoid bending the center pin, inspect for debris prior to reinstallation.
 - Wire harness connectors. Disengage associated circuit breakers first. Use care when handling connectors to ensure shield drain integrity, alternate locking screw tightening sequence and ensure proper alignment during re-installation.
- 3.2.1.3 Each LRU rack uses a lock down mechanism to secure the LRU in the rack.
 - To remove, pull while twisting counter clockwise until the LRU hook is clear. The LRU should now be free.
 - To install, align the LRU in the rack. Slide the LRU until the foot is firmly engaged in the rack. Lift the lockdown mechanism over the LRU hook. Tighten firmly until secure.
 - Conduct necessary checkout procedures described elsewhere in this document.
- 3.2.1.4 Configuration Modules are located inside the J881 GDL 88 connector and inside each J3302 GTX 3000 connector. Hand tools are required to gain access inside the backshell. Retain hardware after removal. To install secure retained hardware and Reconnect to applicable unit. Conduct necessary checkout procedure described elsewhere in this document.

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

3.2.3 Flight Stream 110 (if installed) Line Replaceable Unit (LRU), GPS and UAT antennas:

Hand tools are required. These components are secured with screws and nuts. To remove, remove this hardware and retain. Disconnect the electrical / coax connector. To install, secure with the retained hardware. Reconnect electrical / coax connector.

3.2.4 ADS-B 1/2 Annunciator:

Hand tools are required. This component is secured with locking paws. To gain access to these locking paws gently pull the annunciator lens out. Screws located inside loosen the upper and lower screw. Once locking paw has been loosened pull annunciator body out of the annunciator housing and instrument panel. To install insert annunciator body back through the instrument panel and annunciator housing. Tighten locking paws until annunciator body and housing are tight against the instrument panel, then press annunciator lens firmly into the annunciator body. Conduct necessary checkout procedures described elsewhere in this document.

Chapter 4

LIFTING AND SHORING:

Not applicable this modification.

Chapter 5

LEVELING AND WEIGHING:

Not applicable this modification.

Chapter 6

TOWING AND TAXIING:

Not applicable this modification.

Chapter 7

PARKING AND MOORING:

Not applicable this modification.

Chapter 8

PLACARDS AND MARKINGS:

Not applicable this modification.

Chapter 9

SERVICING:

Not applicable this modification.

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Chapter 10 AIRWORTHINESS LIMITATIONS

10.1 FAA Approval of Airworthiness Limitations

This chapter of this supplement must be attached to the Airplane Maintenance Manuals. The information contained herein supplements the basic Maintenance Manuals only in those areas listed, when the aircraft is modified in accordance with FAA Form 337 for the installation of Garmin ADS-B System. For limitations and procedures not contained in this supplement, consult the basic Airplane Maintenance Manuals.

The Airworthiness Limitations Section is FAA approved and specifies maintenance required and under 14CFR §§ 43.16 and 91.403, unless an alternative program has been FAA approved.

FAA APPROVED: _____

Principal Avionics Inspector
Minneapolis – St. Paul FSDO

FAA Approved Date: _____

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Chapter 10 Revisions Status

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IR	INITIAL RELEASE	ALL		

FAA Approved Date: _____

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

10.1 GPS & UAT Antenna

The area of the GPS and UAT Antenna installations must be inspected visually for damage, wear, and corrosion in accordance with Hawker/Beechcraft general inspection criteria in addition use eddy current inspection techniques in accordance with MIL-STC-271 using templates defined in section 10.3.

Structural Element (Skin)	Threshold (Flights)	Recurring (Flights)
GPS Antenna, Connector Hole	9,058	9,058
GPS Antenna, Fastener Hole	4,146	4,146
UAT Antenna, Connector Hole	21,139	21,139
UAT Antenna Fastener Hole	7,572	7,572

10.2 Inspection Method

The connector and fastener holes should be inspected using eddy current inspection techniques, according to the intervals outlined in tables below. POD curves from the NTIAC "Nondestructive Evaluation Capabilities Data Book" for lap splice joint specimens that a 0.10" crack has a 90% probability of detection using eddy current techniques. The threshold crack length for each location indicates that a crack, if present should be detectable at the time of the threshold inspection. Cracks longer than 0.10" must be repaired prior to the next flight. The inspection should indicate the following details;

Modification	Affected Aircraft System	Structural Detail	Inspection Notes
GPS Antenna UAT Antenna	Aircraft Skin	Connector Hole	Inspect edge of hole in skin.
		Doubler Fastener Holes	Inspect fastener holes in skin. Cracks are most likely around fastener holes closest to the edges of the doubler.

10.3 Required Circle Templates

Modification	Rivets	0.201"	0.75
GPS Antenna UAT Antenna	X	X	X

FAA Approved Date: _____

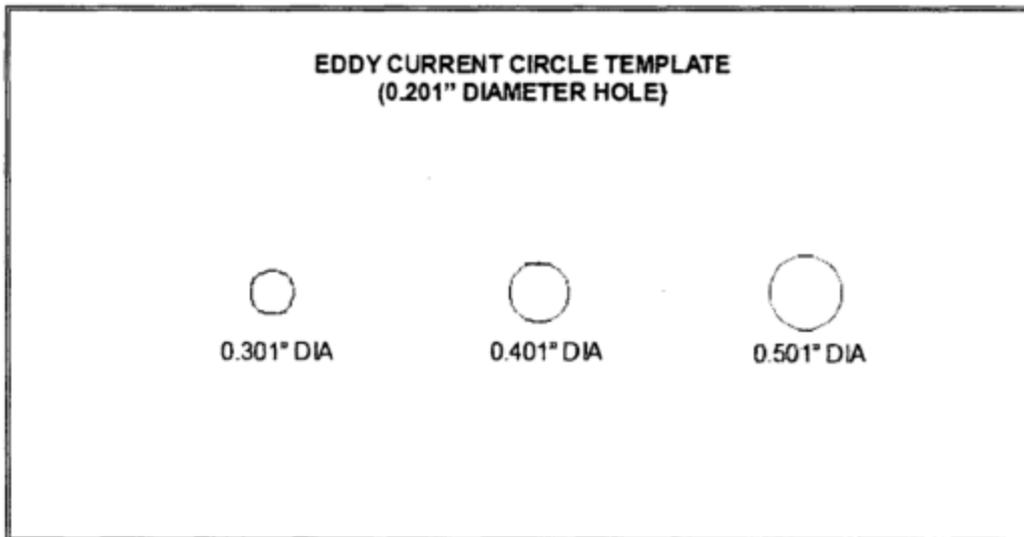
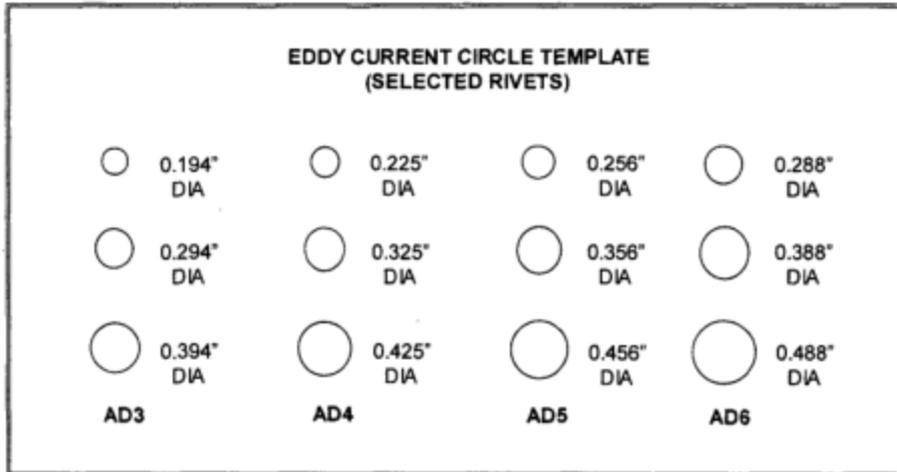
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10.4 Eddy Current Templates



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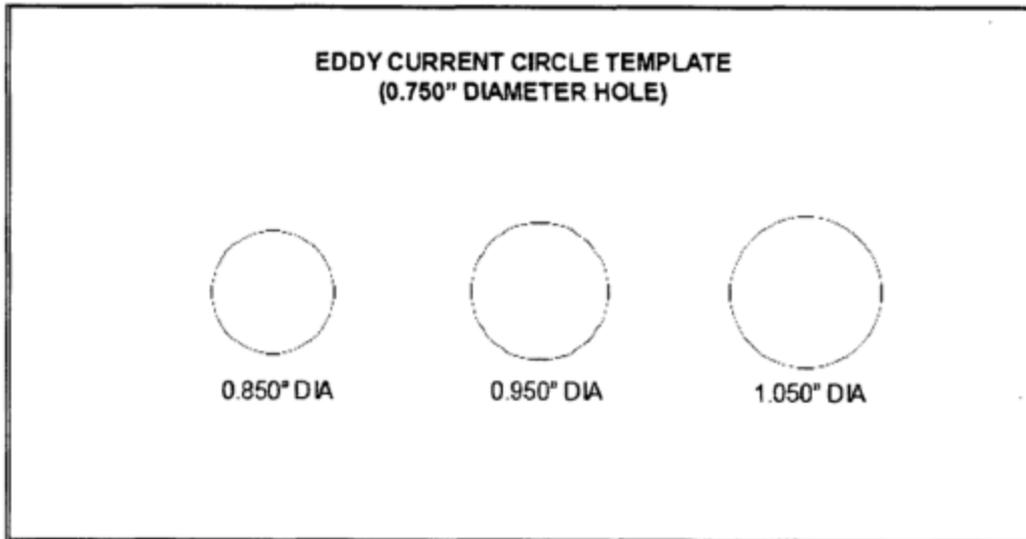
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Revision: IR
Dated 05/01/2018
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A1

Instructions for Continued Airworthiness For Garmin ADS-B System Installation

**APPENDIX A
ATTACHMENTS**

All documents listed in the table below must be attached to this document and maintained in accordance with Chapter one.

ATTACHMENTS			
DOCUMENT NO.	TITLE	REV	PAGES
CFG-3761-560	Configuration Instructions for Garmin ADS-B Out System Installation	IR	14
121-4079-E0560-000	GTX3000 Transponders and GDL88 ADS-B System	IR	6
121-3761-M0560-050	Wire Routing Installation	IR	3

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Instructions for Continued Airworthiness

For

Garmin ADS-B Out System Installation

(Type Certificate A16SW)

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Dated 05/01/2018
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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

REVISIONS PAGE

Rev	Description	Page (S)	Date	Approved By
IR	Initial Release	ALL		

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Chapter 1 INTRODUCTION

1.1 Scope/Purpose/Applicability/Distribution

The Instructions for Continued Airworthiness (ICA) described herein are applicable only to RK-260 modified with Garmin ADS-B System. These ICA describe the recommended and required maintenance procedures for the installation of Garmin ADS-B System.

The maintenance information provided by this document must be included in the operator's Aircraft Maintenance Manual and the operator's Aircraft Scheduled Maintenance Program. All documents listed in the required documents section must be maintained with this document. FAA accepted revisions to this document will be distributed by mail to the applicable certifying authorities and operators of legally modified aircraft.

1.2 System Description

The features of this modification include the removal of both Collins TDR-94D transponders and replacement with Garmin GTX 3000 transponders, reusing the existing aircraft wiring. This modification also includes a Garmin GDL 88 GPS / UAT and optional Flight Stream 110 Bluetooth FIS-B installation. A GPS and UAT-SBS antenna is installed. Annunciation is added for ADS-B status. New circuit breakers are installed in the copilot's circuit breaker panel. New wiring is added to supply the GTX 3000 transponders with GPS position for the GDL 88. The Flight Stream 110 (if installed) supports properly equipped FIS-B capable Bluetooth devices

1.3 Required Data

All required data is located in Appendix A of this document.

1.4 Referenced Garmin documents:

Nomenclature	Drawing Number	Rev	Date
GTX 3000 Transponder Installation Manual	190-00926-01	L	Oct 2015
GDL 84/88 TSO Installation Manual	190-01122-00	H	Dec 2014
Flight Stream 110/210 TSO Installation Manual	190-01700-00	E	Dec 2015

1.5 Systems Control:

1.5.1 The GTX 3000 transponders replace the Collins TDR-94D transponders. All control is via the existing Collins Pro Line 21 cockpit Control Display Units. No changes to operational control are imposed by this STC.

1.5.2 The GDL 88 is a GPS sensor and FIS-B UAT receiver. The GDL 88 has no external control.

1.5.3 The Flight Stream 110 is a Bluetooth repeater for the GDL 88 FIS-B products and has no external control.

1.6 Document administration:

1.6.1 This document will be revised by Elliott Aviation Engineering when revisions to descriptive or approval data warrant.

1.6.2 When this document is revised, all pages will be revised to indicate the current level. Chapter 10 will have its own revision status embedded. FAA approved revisions to the document will be distributed by mail to the applicable certifying authorities and operations of legally modified aircraft.

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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

1.7 Abbreviations and Definitions:

1090ES	A transponder distinction. 1090 is the classic frequency of Secondary Surveillance Radar (SSR) transponders. The addition of "ES" at the end signifies that the transponder is equipped with Extended Squitter used by the ADS-B aircraft segment. Note that this class of transponder supports Class "A" airspace (altitudes above 18,000 feet) operations. Typically, no FIS-B or TIS-B functionality is included, compared to UAT.
AMM	Aircraft Maintenance Manual
DAH	Design Approval Holder. The FAA term that includes STC holders such as Elliott Aviation.
FIS-B	Flight Information System broadcast. A radio frequency service that transmits weather and aeronautical products, either by satellite or ground based systems. Refer to FAA Advisory Circular 20-149A for thorough explanation.
Latency	The total time between when the position is measured and when the position is transmitted by the aircraft. A 14 CFR §91.227 performance requirement for ADS-B broadcast systems. The collection of specific equipment, interface and testing represented by this STC comprises an approved Latency
OEM	Original Equipment Manufacturer.
SBS	Surveillance and Broadcast Service. Provided by the FAA. Ground based transmission at a frequency of 978 MHz.
SDA	System Design Assurance. A 14 CFR §91.227 performance requirement for ADS-B broadcast systems. The collection of specific equipment, interface and testing represented by this STC comprises an approved SDA.
UAT	Universal Access Transceiver. A new device that supports the ADS-B aircraft segment at lower altitudes (below 18,000 feet) Allows smaller aircraft that operate below Class "A" airspace a more

1.8 Precautions

Typical precautions are to be taken when accessing, removing, replacing, testing or inspecting these modifications. Specific precautions will be addressed in context throughout this document.

1.9 Units of measurement:

Non- metric. English units including Inches, feet, etc..

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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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Chapter 2 MAINTENANCE INSTRUCTIONS AND SCHEDULES:

2.1 Mandatory Replacements:

The GTX 3000 1090ES Transponder, GDL 88 GPS / UAT and optional Flight Stream 110 systems' electrical components and installation is "On Condition" as determined by routine surveillance and there is no mandatory replacement time for any aspect of this system.

Refer to 400A OEM AMM, Wiring Procedures and Repair – Description and Operations - chapter 20, for comprehensive wiring harness surveillance concerns.

See Chapter 10 Airworthiness Limitations for antenna installation inspections.

2.2 Recurring Maintenance:

2.2.1 Antenna Installations

Antenna installations require special inspection requirements and shall be maintained using Chapter 10 of this document.

2.2.2 Bulkhead Feedthrough Installation

The bulkhead feedthrough installation requires no special inspections requirements and shall be maintained using normal OEM maintenance manual procedures

2.2.3 Basic Mode S Transponder functions:

The installation of the GTX 3000 transponders must continue to meet the operational requirements of 14 CFR §§ 91.413, 91.215, and 91.217 and comply with the transponder system tests and inspections called out in 14 CFR § 43, appendix F.

The Collins Pro Line 21 altitude reporting equipment connected to the Garmin ADS-B transponders must comply with all applicable 14 CFR §§ 91.217, 91.411, and part 43 appendix E test and inspection requirements. If the altimetry system is compliant with the Reduced Vertical Separation Minimum (RVSM) standards, the requirements and tolerances stated in the approved RVSM maintenance program must be met. ADS-B installation does not alter these requirements.

2.2.4 ADS-B functions:

Compliance with aircraft equipage requirements imposed by the operational requirements of 14 CFR §§ 91.225 and 91.227 is achieved by these STC modifications. ADS-B Maintenance is "On Condition" as determined by routine surveillance during Mode S maintenance, by the flight crew as well the FAA Air Traffic Control system.

2.2.5 FIS-B functions:

This modification has an optional installation that supports ground based SBS FIS-B reception. UAT/FIS-B Maintenance is "On Condition".

2.3 Setup and Configuration in Service

2.3.1 Garmin GDL 88

2.3.1.1 Confirm Garmin GDL 88 line replaceable unit part number from the Elliott Aviation wiring diagram 121-3761-E0560-000 located in Appendix A.

2.3.1.2 Obtain Elliott Configuration Instructions located in Appendix A. Read Section 5 of Configuration Instructions before proceeding with reconfiguration of the GDL 88. Take special note of the specific instructions for laptop computer interface, software effectivity and maintenance record details.

The GDL-88 will not function properly until software setup and configuration is accomplished.

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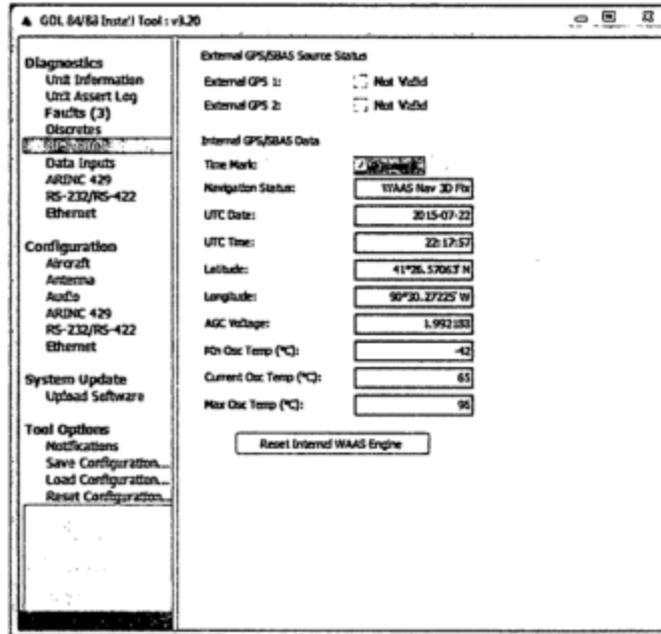
Instructions for Continued Airworthiness For Garmin ADS-B System Installation

PRECAUTION:

Part number, configuration and electrical interface has been evaluated and approved for ADS-B Latency and SDA requirements. Any deviations will require additional approval.

- 2.3.1.3 While using the Elliott Configuration Instructions Section 5, access to the GDL 88 Diagnostics section is available. Real time GPS information is presented.
- 2.3.1.4 For setup and configuration, GPS reception must be confirmed using the diagnostic page. Use known GPS position to validate the GDL 88 diagnostic position readout such as the aircrafts navigation GPS. This information is presented on the Pro Line 21 cockpit control display unit under the IDX pages. Sample page below show typical GDL 88 GPS data.
- 2.3.1.5 After GPS reception is verified aircraft maybe returned to service.
- 2.3.1.6 For troubleshooting an in-service system, GPS activity may be confirmed using this diagnostic page. Lack of GPS position may be due to disconnect antenna coax, faulty antenna or faulty GDL 88. Neither the GDL 88 nor GPS antenna is field serviceable. If faulty, the GDL 88 line replaceable unit may only be repaired by Garmin. Replace a faulty GPS antenna with new.

Sample page:



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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

2.3.2 GTX 3000 Transponder

- 2.3.2.1 Confirm Garmin GTX 3000 line replaceable unit part number from the Elliott Aviation wiring diagram 121-4079-E0560-000 located in Appendix A.
- 2.3.2.2 Obtain Elliott Configuration Instructions located in Appendix A. Read sections 2 and 4 of this document before proceeding with reconfiguring of the GTX 3000. Take special note of the requirements for laptop computer interface and software effectivity details.

The GTX 3000 will not function properly until software setup and configuration is accomplished.

PRECAUTION:

Part number, configuration and electrical interface has been evaluated and approved for ADS-B Latency and SDA requirements. Any deviations will require additional approval.

- 2.3.2.3 While using the Elliott Configuration Instructions, access to the GTX 3000 status is available. Real time information is presented.
- 2.3.2.4 For setup and configuration, the following items must be verified:
 - GPS Position
 - Magnetic
 - Weight on wheels is operating correctly.
 - ADS-B mode indicated.
 - 4096 code agrees with Pro Line 21 RTU ATC settings.
 - Flight ID agrees with Pro Line 21 RTU ATC settings. (note: Dynamic (pilot set) Flight ID not available on aircraft all.)
 - TCAS status correct for aircraft. Most aircraft will have TCAS installed.
- 2.3.2.5 GPS position is confirmed using the "Info / ADS-B Out" page. Confirm GPS position from the GDL 88. Sample page follows to show typical GTX 3000 GPS source information.
- 2.3.2.6 Heading is confirmed using "Info / EHS" page. Confirm heading from the Pro Line 21 system. Sample page follows to show typical GTX 3000 Heading source information.
- 2.3.2.7 WOW, ADS-B mode, 4096 Code, Flight ID, and TCAS status is confirmed using the "Status" page in the GTX 3000 configuration tool. Sample page follows to show typical GTX 3000 "Status" page information.
- 2.3.2.8 After these have been verified perform 14 CFR 43 App F ¶ A-J.
- 2.3.2.9 The aircraft may now be returned to service.

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

GTX 3000 Configuration GPS Sample page:

The screenshot displays the 'GTX 3000 Configuration GPS Sample page'. It features a top navigation bar with 'File' and 'Help' options. Below this, there are tabs for 'ADS-B Data' and 'Pre-Load'. The main area is divided into several sections:

- Required Parameters:** A grid of input fields for GPS Latitude (deg), Longitude (deg), Ground Speed (kt), Heading (deg), and Altitude (ft).
- Required Parameters (one of the following):** Fields for True Heading (deg) and Magnetic Heading (deg).
- Required Parameters (one of the following):** Fields for MCK/DCI Selected Altitude (ft) and RW Selected Altitude (ft).
- Other Parameters:** Fields for GPS Horizontal Protection Limit (ft), GPS Horizontal Figure of Merit (ft), GPS Vertical Figure of Merit (ft), Penetration Offset Applied by Sensor, GPS Height Above Ground (ft), Vertical Rate (ft/min), and Barometric Pressure Setting (hPa).
- Buttons:** A 'Next Timeout Page' button is located at the bottom center.
- Bottom Section:** A 'Status' section with 'Operating Mode' (ADS-B, Cessna), 'GNSS State' (GPS, WAAS, etc.), 'Status On' (various system checks), 'Degraded (Pre-Load only)', and 'PC Serial Interface' (Port, Baud Rate).

GTX 3000 Configuration Heading Sample page:

The screenshot displays the 'GTX 3000 Configuration Heading Sample page'. It features a top navigation bar with 'File' and 'Help' options. Below this, there are tabs for 'ADS-B Data' and 'Pre-Load'. The main area is divided into several sections:

- Required Parameters:** Fields for Ground Track (deg), Ground Speed (kt), Magnetic Heading (deg), Indicated Airspeed (kt), Mach, and Selected Altitude (ft).
- Required Parameters (one of the following):** Fields for Air Data Vertical Rate (ft/min) and Vertical Vertical Rate (ft/min).
- Required Parameters (one of the following):** Fields for Track Angle Rate (deg/s) and True Airspeed (kt).
- Other Parameters:** Fields for Barometric Pressure Setting (hPa) and Roll Angle (deg).
- Buttons:** A 'Next Timeout Page' button is located at the bottom center.
- Bottom Section:** A 'Status' section with 'Operating Mode' (ADS-B, Cessna), 'GNSS State' (GPS, WAAS, etc.), 'Status On' (various system checks), 'Degraded (Pre-Load only)', and 'PC Serial Interface' (Port, Baud Rate).

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

GTX 3000 Configuration Status Sample Page:

- 2.3.2.10 For troubleshooting an in-service system, GPS activity may be confirmed using this "Info / ADS-B Out" page. Lack of GPS position may be due to GDL 88 issues such as disconnect antenna coax, faulty antenna or faulty GDL 88. See 2.5 above. If the GDL 88 is not suspect, suspect RS 232 interface, configuration or GTX 3000 transponder. The GTX 3000 is not field serviceable. If faulty, the GTX 3000 line replaceable unit may only be repaired by Garmin.
- 2.3.2.11 For troubleshooting an in-service system, heading activity may be confirmed using this "Info / EHS" page. Lack of heading may be due to suspect 429 interface or configuration. The GTX 3000 heading source is the same as is used for the Flight Crew presentations. The GTX 3000 is not field serviceable. If faulty, the GTX 3000 line replaceable unit may only be repaired by Garmin.
- 2.3.2.12 For troubleshooting an in-service system, the above may be confirmed using this "Status" page:
 - Weight on wheels is operating correctly. (squat switch, wiring and config).
 - ADS-B mode indicated. (GDL 88, valid 429 data, wiring and config).
 - 4096 code agrees with Pro Line 21 CDU ATC settings. Access the Pro Line control display unit and exercise the squawk codes, the GTX 3000 should indicate correctly almost immediately. (wiring and config).
 - Flight ID agrees with Pro Line 21 CDU ATC settings. Select reversionary mode on the associated instrument panel to enable off-side control display unit GTX 3000 control. (CDU, wiring and config).
 - TCAS status correct for aircraft. Most aircraft will have TCAS installed. Confirm TCAS by visual observation of directional antennas and / or Pro Line control display unit TCAS page. Confirm no faults if TCAS installed and the ADS-B system is interfaced properly. (wiring and config).
- 2.3.2.13 The GTX 3000 is not field serviceable. If faulty, the GTX 3000 line replaceable unit may only be repaired by Garmin.

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

2.3.3 GTX 3000 "N" number Registration change.

- 2.3.3.1** Obtain Elliott Configuration Instructions located in Appendix A. Using sections 2 and 4 of the Configuration Instructions Access the "Aircraft" tab on the GTX 3000 configuration tool. Modify the registration as needed and press enter to set the configuration. Shutdown the configuration tool and verify registration number change using IFR 6000 or similar transponder test set.

Sample page:

A screenshot of the GTX3000 Install Tool software interface. The window title is "GTX3000 Install Tool". It has a menu bar with "File" and "Help". Below the menu bar are tabs: "Info", "Aircraft", "Airframe", "Wiring", "Regions", and "Failure Log". The "Aircraft" tab is selected. Under "Aircraft Configuration", there are several input fields:

- Aircraft Registration: []
- IATA Airline Designator: []
- VFR Code: 1200
- AHS Qty: 2 (dropdown)
- ICAO Address (hex): []
- Set ICAO Address from Registration: []
- Country: United States (dropdown)

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

2.4 GTX 3000 / GDL 88 / Pro Line 21 Troubleshooting

Reported Issue	Steps	
Transponder reporting: Altitude	Observe or determine Collins flight displays for air data indications.	If present and correct, GTX 3000 suspect wiring or configuration.
Transponder reporting: Heading	Observe or determine Collins flight displays for heading indications.	If present and correct, GTX 3000 suspect wiring or configuration.
Transponder reporting: ADS-B - Air	Observe or determine the ADS-B annunciator status.	If ADS-B annunciation OFF, suspect GTX 3000 in use. If ADS-B annunciation ON, suspect GDL 88 LRU, wiring or configuration.
Transponder reporting: ADS-B - Surface	Not seen by flight crew, report will come from ATC Ground.	Same as above for ADS-B annunciation status.
Transponder reporting: Mode C	Observe or determine the Collins CDU annunciations.	If alternate GTX 3000 can be controlled using Collins CDU, suspect the GTX 3000 LRU.
TCAS	Collins CDU: Navigate to ATC page and observe status.	Access the Collins MFD status page. TCAS faults with ATC faults point to transponder malfunction. TCAS faults alone point to TCAS.

Commercial Off The Shelf ADS-B receiver devices are suitable for GO / NO GO confirmation that the GTX 3000 transponders are transmitting GPS position. HOWEVER, only approved calibrated test devices, such as the IFR 6000, may be used to confirm transponder output power and minimum trigger levels as well as diversity isolation.

To conduct an in-depth as-installed complete functional check including UAT, qualified test equipment such as the IFR 6000 will be required. Refer to specific test set instructions for operation and testing.

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Chapter 3 ACCESS:

3.1 Overview:

The information contained herein supplements the basic Maintenance Manuals only in those areas listed, when the aircraft is modified in accordance with Elliott Aviation drawings and diagrams specified in Section 1.3 of this document.

- 3.1.1 GTX 3000s are located in the forward right avionics bay upper equipment shelf. A configuration module is also located inside the J3302 connectors.
- 3.1.2 GDL 88 is located in the forward left avionics bay lower equipment shelf. A configuration module is also located inside the J811 connector.
- 3.1.3 Flight Stream 110 (if installed) is located on the back of the Pilots chart holder cabinet.
- 3.1.4 ADS-B 1/2 annunciator is located in the cockpit panel
- 3.1.5 GA36 GPS Antenna is located on top of the aircraft fuselage above the cockpit headliner at FS 128.74 right of center line
- 3.1.6 UAT Antenna is located on the belly of the aircraft fuselage below the left forward avionics bay at FS 35.14
- 3.1.7 Wiring follows existing aircraft wire routes in the nose and cockpit area.

3.2 Equipment and wiring access:

No new access is required from the installation of this modification. Access to all equipment and wiring is gained using existing aircraft maintenance manual techniques.

3.2.1 GDL 88 and GTX 3000 rack mounted Line Replaceable Units (LRU):

- 3.2.1.1 No hand tools are required. The coax and wire harness connectors knurled and are intended to be removed and tightened by hand. Should assistance in the form of pliers be used, take care to not distort or over tighten.
- 3.2.1.2 Observe typical avionics systems precautions including:
 - Coax connectors. Do not bend or kink the coax, avoid bending the center pin, inspect for debris prior to reinstallation.
 - Wire harness connectors. Disengage associated circuit breakers first. Use care when handling connectors to ensure shield drain integrity, alternate locking screw tightening sequence and ensure proper alignment during re-installation.
- 3.2.1.3 Each LRU rack uses a lock down mechanism to secure the LRU in the rack.
 - To remove, pull while twisting counter clockwise until the LRU hook is clear. The LRU should now be free.
 - To install, align the LRU in the rack. Slide the LRU until the foot is firmly engaged in the rack. Lift the lockdown mechanism over the LRU hook. Tighten firmly until secure.
 - Conduct necessary checkout procedures described elsewhere in this document.
- 3.2.1.4 Configuration Modules are located inside the J881 GDL 88 connector and inside each J3302 GTX 3000 connector. Hand tools are required to gain access inside the backshell. Retain hardware after removal. To install secure retained hardware and Reconnect to applicable unit. Conduct necessary checkout procedure described elsewhere in this document.

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

3.2.3 Flight Stream 110 (if installed) Line Replaceable Unit (LRU), GPS and UAT antennas:

Hand tools are required. These components are secured with screws and nuts. To remove, remove this hardware and retain. Disconnect the electrical / coax connector. To install, secure with the retained hardware. Reconnect electrical / coax connector.

3.2.4 ADS-B 1/2 Annunciator:

Hand tools are required. This component is secured with locking paws. To gain access to these locking paws gently pull the annunciator lens out. Screws located inside loosen the upper and lower screw. Once locking paw has been loosened pull annunciator body out of the annunciator housing and instrument panel. To install insert annunciator body back through the instrument panel and annunciator housing. Tighten locking paws until annunciator body and housing are tight against the instrument panel, then press annunciator lens firmly into the annunciator body. Conduct necessary checkout procedures described elsewhere in this document.

Chapter 4

LIFTING AND SHORING:

Not applicable this modification.

Chapter 5

LEVELING AND WEIGHING:

Not applicable this modification.

Chapter 6

TOWING AND TAXIING:

Not applicable this modification.

Chapter 7

PARKING AND MOORING:

Not applicable this modification.

Chapter 8

PLACARDS AND MARKINGS:

Not applicable this modification.

Chapter 9

SERVICING:

Not applicable this modification.

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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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Elliott Aviation Technical Products Development, Inc.
6601 74th Ave Building A
Milan, IL 61264



Report No.: ICA-4079-560
Revision: IR
Dated 05/01/2018
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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Chapter 10 AIRWORTHINESS LIMITATIONS

10.1 FAA Approval of Airworthiness Limitations

This chapter of this supplement must be attached to the Airplane Maintenance Manuals. The information contained herein supplements the basic Maintenance Manuals only in those areas listed, when the aircraft is modified in accordance with FAA Form 337 for the installation of Garmin ADS-B System. For limitations and procedures not contained in this supplement, consult the basic Airplane Maintenance Manuals.

The Airworthiness Limitations Section is FAA approved and specifies maintenance required and under 14CFR §§ 43.16 and 91.403, unless an alternative program has been FAA approved.

FAA APPROVED: _____
Principal Avionics Inspector
Minneapolis – St. Paul FSDO

FAA Approved Date: _____

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Dated 05/01/2018
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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

Chapter 10 Revisions Status

Rev	Description	Page (S)	Date	FAA Approved By
IR	INITIAL RELEASE	ALL		

FAA Approved Date: _____

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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

10.1 GPS & UAT Antenna

The area of the GPS and UAT Antenna installations must be inspected visually for damage, wear, and corrosion in accordance with Hawker/Beechcraft general inspection criteria in addition use eddy current inspection techniques in accordance with MIL-STC-271 using templates defined in section 10.3.

Structural Element (Skin)	Threshold (Flights)	Recurring (Flights)
GPS Antenna, Connector Hole	9,058	9,058
GPS Antenna, Fastener Hole	4,146	4,146
UAT Antenna, Connector Hole	21,139	21,139
UAT Antenna Fastener Hole	7,572	7,572

10.2 Inspection Method

The connector and fastener holes should be inspected using eddy current inspection techniques, according to the intervals outlined in tables below. POD curves from the NTIAC "Nondestructive Evaluation Capabilities Data Book" for lap splice joint specimens that a 0.10" crack has a 90% probability of detection using eddy current techniques. The threshold crack length for each location indicates that a crack, if present should be detectable at the time of the threshold inspection. Cracks longer than 0.10" must be repaired prior to the next flight. The inspection should indicate the following details;

Modification	Affected Aircraft System	Structural Detail	Inspection Notes
GPS Antenna UAT Antenna	Aircraft Skin	Connector Hole	Inspect edge of hole in skin.
		Doubler Fastener Holes	Inspect fastener holes in skin. Cracks are most likely around fastener holes closest to the edges of the doubler.

10.3 Required Circle Templates

Modification	Rivets	0.201"	0.75
GPS Antenna UAT Antenna	X	X	X

FAA Approved Date: _____

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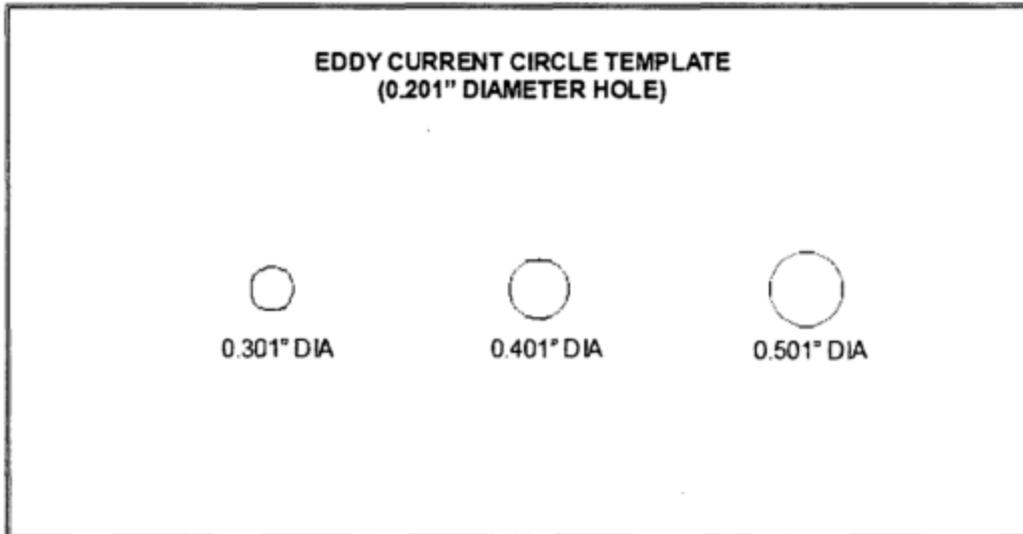
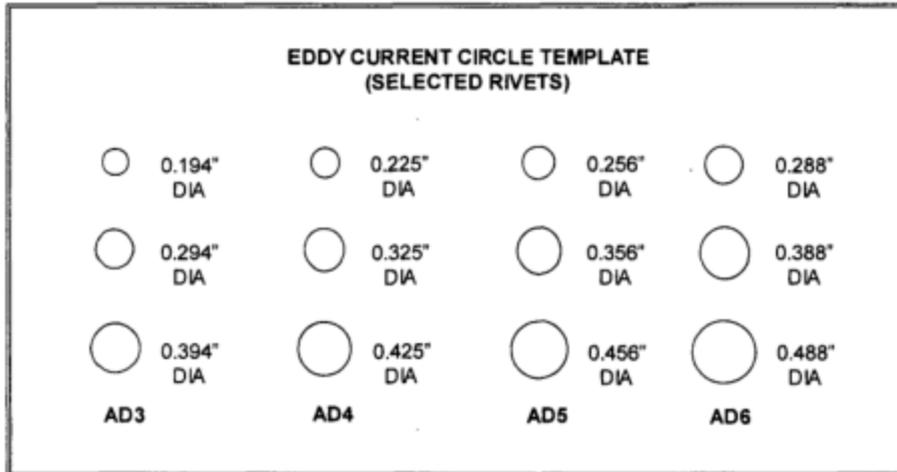
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10.4 Eddy Current Templates



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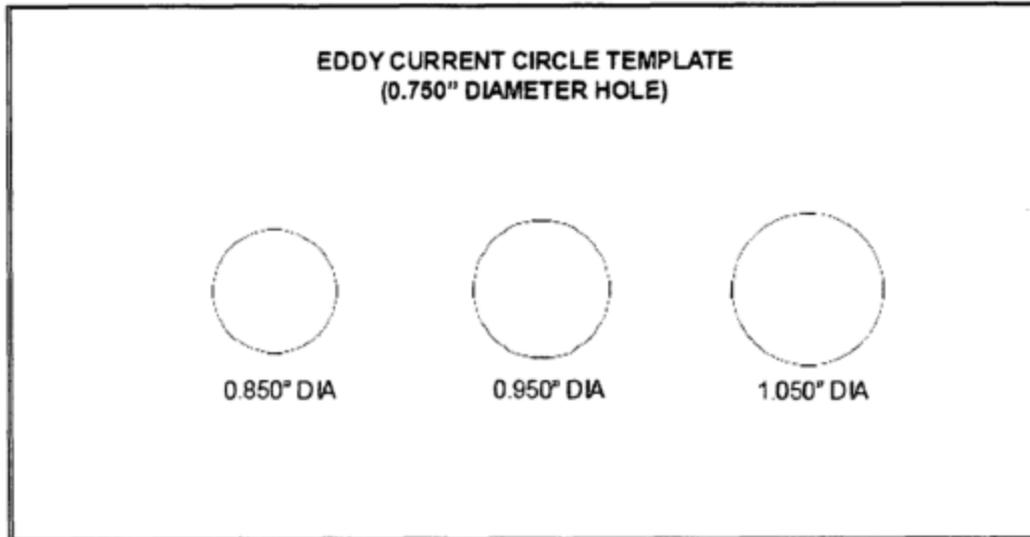
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Elliott Aviation Technical Products Development, Inc.
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Report No.: ICA-4079-560
Revision: IR
Dated 05/01/2018
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Instructions for Continued Airworthiness For Garmin ADS-B System Installation



FAA Approved Date: _____

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Instructions for Continued Airworthiness For Garmin ADS-B System Installation

**APPENDIX A
 ATTACHMENTS**

All documents listed in the table below must be attached to this document and maintained in accordance with Chapter one.

ATTACHMENTS			
DOCUMENT NO.	TITLE	REV	PAGES
CFG-3761-560	Configuration Instructions for Garmin ADS-B Out System Installation	IR	14
121-4079-E0560-000	GTX3000 Transponders and GDL88 ADS-B System	IR	6
121-3761-M0560-050	Wire Routing Installation	IR	3

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SDNY_GM_02758103

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Flight Manual Supplement For Garmin ADS-B System Installation

**FAA Approved
Aircraft Flight Manual Supplement
For
Garmin ADS-B Out System Installation
In
Beechcraft Corporation Model 400A (Beechjet)
(Type Certificate A16SW)**

Registration: N727KB

Serial Number: RK-260

This supplement must be attached to the Approved Airplane Flight Manual when the Garmin ADS-B Out System is installed in accordance with FAA Form 337 dated 5-2-2018. The information contained in this document supplements or supersedes the basic Airplane Flight Manual only for the areas listed. For Limitations, Procedures, and Performance not contained in this supplement, consult the basic Airplane Flight Manual

DALE E
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FAA Approved MSP-FSDO-GL15

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FAA APPROVED DATE: _____

SDNY_GM_02758104

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245026

EFTA01261686

SDNY_GM_02758105

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245027

EFTA01261687

Elliott Aviation Technical Products Development, Inc.
6601 74th Ave Building A
Milan, IL 61264



Report No.: FMS-4079-560
Revision: IR
Page 2 of 5

Flight Manual Supplement For Garmin ADS-B System Installation

REVISIONS PAGE

Rev	Description	Page (S)	Date	Approved By
IR	Initial Release	ALL		

FAA APPROVED DATE: _____

SDNY_GM_02758106

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245028

EFTA01261688

SDNY_GM_02758107

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245029

EFTA01261689



Flight Manual Supplement For Garmin ADS-B System Installation

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FAA APPROVED DATE: _____

SDNY_GM_02758108

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758109

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245031

EFTA01261691



Flight Manual Supplement For Garmin ADS-B System Installation

1 General:

The Garmin GTX 3000 Mode S Transponders meet FAA TSO C112d and are digitally interfaced into the Collins Pro Line 21 system. This includes existing Air Data Systems (ADS) and control systems. There is no change to the Collins Pro Line 21 Control Display Unit (CDU) "ATC CONTROL" page operations or indications. There is no change to the existing transponder circuit breaker assignment, value, labeling or operation.

The GTX 3000 Transponders also meet FAA TSO C166b and therefore incorporate by design ADS-B OUT 1090ES functionality. No separate control is required to enable ADS-B functionality.

The GTX 3000 Transponders are digitally interfaced to the Attitude Heading System (AHS) and TCAS (if installed) to support ADS-B functionality.

The Garmin GDL 88 GPS / UAT meets FAA TSO C145c/B2 and is interfaced with the GTX 3000 Transponders to serve as ADS-B required position source. No separate control is available and may only be disabled by pulling the circuit breaker.

ADS-B GPS required position source failure is indicated by the amber **ADSB 1** or **ADSB 2** annunciators. The "1" and "2" annunciators lights illuminate based on the active transponder. ADS-B position monitoring is only available when the selected transponder is ON.

ADS-B device failure is indicated by the ATC FAIL annunciation displayed on the ProLine 21 RTU.

The GDL 88 GPS / UAT also meets FAA TSO C154c which allows reception of FAA TSO C157a/1 ADS-B FIS-B information for distribution to Portable Electronic Devices via a Garmin Flight Stream 110 Bluetooth FAA TSO C157 transceiver. No separate control is available and may only be disabled by pulling the circuit breaker.

The GDL 88 GPS / UAT and optional Flight Stream 110 Bluetooth units are protected by the following circuit breakers:

C/B Name	Amps	Location	Bus
ADS-B GPS	1	Pilot CB Panel	28vdc Right Main Bus
FIS-B BL TH	1/2	Pilot CB Panel	28vdc Right Main Bus

The installed ADS-B OUT system has been shown to meet the equipment requirements of 14 CFR 91.227.

FAA APPROVED DATE: _____

SDNY_GM_02758111

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245033

EFTA01261693



Flight Manual Supplement For Garmin ADS-B System Installation

2 Limitations:

This STC does not constitute airworthiness approval for portable electronic devices whether used to display FIS-B information or any other aviation application. This STC approves FIS-B provisions which may be displayed on a portable electronic device as allowed by regulation. An operational approval may be required.

The GDL88 GPS/UAT is not approved as a navigation position source.

3 Emergency:

No change to basic AFM.

4 Abnormal:

- 4.1 ADS-B position loss: In the event the **ADSB 1** or **ADSB 2** annunciator (amber) illuminates, notify ATC.
- 4.2 ADS-B loss of function: In the event an ATC FAIL message on the RTU, select alternate transponder.

5 Normal:

- 5.1 Before Taxi – Select transponder ON (this enables ADS-B ground functions for ATC monitoring)
- 5.2 Shutdown – After parking brake is set, select transponder to STBY.
- 5.3 If the optional Flight Stream 110 is installed FIS-B information is intended to enhance pilot awareness of weather and airspace conditions. It does not replace positive two way communication when making safety critical weather or routing decisions. Use FIS-B weather and NAS status information as follows:
 - a) To aid pilot awareness of hazardous meteorological conditions and awareness of the regulatory status of the airspace.
 - b) Changes in hazardous meteorological conditions and/or airspace status the pilot should communicate with the ATC controller, FSS specialist, or operator dispatch for more information about the current meteorological conditions or regulatory airspace status.
 - c) FIS-B information is meant to enhance flight planning only. It lacks sufficient resolution and updating necessary for tactical maneuvering.

6 Performance:

No change to basic AFM.

7 Weight and Balance / Equipment List:

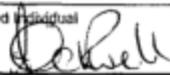
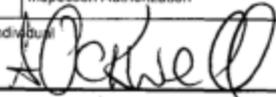
See basic Airplane Flight Manual for current Weight and Balance

FAA APPROVED DATE: _____

SDNY_GM_02758113
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245035

EFTA01261695

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)			Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260		
	Make Raytheon Aircraft Company 400A		Model 400A	Series	
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) 26180 Curtiss Wright PKWY		
			City Richmond Heights	State OH	
		Zip 44143-1453	Country USA		
3. For FAA Use Only					
3. For FAA Use Only					
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name Nextant Aerospace			<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Manufacturer		
Address 355 Richmond Road			<input type="checkbox"/> Foreign Certificated Mechanic C. Certificate No.		
City Cleveland State OH			<input checked="" type="checkbox"/> Certificated Repair Station		
Zip 44143-1453 Country USA			<input type="checkbox"/> Certificated Maintenance Organization CRS # 25NR667B		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A.Ockwell 		01/04/2013	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport	
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)	
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A.Ockwell 		01/04/2013	

FAA Form 337 (10-06)

SDNY_GM_02758114

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245036

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SDNY_GM_02758115

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245037

EFTA01261697

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Installed the following In-Flight Entertainment System in accordance with STC # ST03960AT and in accordance with Nextant Aerospace Master Drawing List DB1002067. Revision E. Dated January 26 2012.

The Emteq LED cabin lights provide high reliability low power consumption Indirect Cabin Lighting, Galley Task Lighting and Lavatory Vanity Lighting. Installed LED Passenger Safety Signs The cabin lights are interfaced to the existing cabin lighting power distribution and control switches.

Installed Cabin LED Lighting System without the Venue System in accordance with DB1107014. Rev IR Dated January 26,2012.

Installed LED Passenger Safety Signs in accordance with Wiring Diagram DB 1108001 Rev IR January 26 2012

Scheduled Inspection and Maintenance.

Inspect System Installation Component including wire Bundles, Cables Racks and Connector Condition and security in conjunction with the B Check (every 400 hours after Initial 200 Hours "I" inspection) tasks and "C" Check (every 1200 after initial 200 Hr "I" Inspection) Per Hawker Beechjet 400A Maintenance Manual 5.20.02 requirements listed within Scheduled Inspection Section RK-1 and above

ICA for LED Cabin Lighting System Ref to Nextant Aerospace Document No DB1108019 Rev IR Dated September 22 2011 or later FAA Accepted Revisions

Installed Iridium Sat Comm and Broadband Internet System with Aircell Axxess II and ATG 4000 in Accordance with DB 1001013. Rev D

ICA for Aircell System Ref to Nextant Aerospace Document No DB1105003. Rev A dated September 22 2011 or later FAA Accepted Revisions

The following manuals contain complete detailed maintenance instructions and should be consulted for all maintenance activities not covered within this ICA:

Aircell Axxess II Installation Manual D12004 Revision E, dated March 2010 or later applicable revision

Aircell ATG-4000 Installation Manual D13485 Revision B Dated November 2009 or later applicable revision

Aircell Axxess II WLAN AFM Supplement DB1002072 Rev IR dated March 22 2012 for the has been Installed in the AFM

Installed the Astronics DC-AC 115 VAC Power Supply in Accordance with the following Nextant Aerospace Drawings This is Located under the Seat Divan

NX252-10007-03. Rev IR Dated June 04 2012 Channel Hat Inverter Mount

NX252-10007-04. Rev IR Dated June 12, 2012. Assy Inverter 115 VAC, W/Heatsink and Hat.

NX252-10007-05 Rev IR. Dated June 14 2012 Mount Plate ,Single Inverter. Under Divan

NX252-10007-06 Rev IR Dated. June 13. 2012. Assy. Inverter, Divan Mount

NX252-10007-12. Rev IR. Dated June 14 . 2012 Assy ,Bracket Plate Mount Inverter ,Under Divan

NX252-10007-13. Rev IR, Dated June 14 2012. Angle.1x1" . Inverter Mount

ICA for Astronics 115 VAC Cabin Inverter Ref to Nextant Aerospace Report No DB1108018 Rev IR Date 9/22/2011 or later FAA Accepted Revision.

Aircraft Flight Supplement DB1109012 has been installed on the Aircraft

See Current Weight and Balance Dated 12/21/2012

End

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758116

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245038

EFTA01261698

SDNY_GM_02758117

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245039

EFTA01261699

United States of America
Department of Transportation — Federal Aviation Administration

Supplemental Type Certificate

Number ST03960AT

This certificate issued to Nextant Aerospace LLC
3800 Southern Blvd.
West Palm Beach, FL 33406

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified herein meets the airworthiness requirements of Part 25 of the Federal Aviation Regulations.

Original Product - Type Certificate Number : A16SW
Make : Hawker Beechcraft Corporation
Model : 400A

Description of Type Design Change:

Installation of Rockwell Collins Venue In-Flight Entertainment(IFE), Aircell Axxess Cabin II Iridium Phone and High Speed Internet System with Wireless Local Area Network (WLAN) Astronics 115VAC 60HZ Cabin Inverter, and Emteq LED Cabin Lighting System in accordance with Nextant Aerospace LLC, Master Drawing List DB1002067, Revision E, dated January 26, 2012, or later FAA approved revision.

Limitations and Conditions: This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated, unless it is determine by the installer that the interrelationship between this change and any other previously approved modifications will produce no adverse effect upon the airworthiness of that airplane. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application : July 28, 2012

Date reissued :

Date of issuance : March 22, 2012

Date amended :



By direction of the Administrator

[Signature]
(Signature)

Melvin D. Taylor
Manager
Atlanta Aircraft Certification Office

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAX Form B110-2(10-06)

PAGE 1 OF 2 PAGES

This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758118

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245040

EFTA01261700

SDNY_GM_02758119

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245041

EFTA01261701

United States of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate
(Continuation Sheet)

Number ST03960AT

When the Rockwell Collins Venue system is installed Airplane Flight Manual Supplement, DB1106015, Revision A, dated March 22, 2012, or later FAA approved revision, is required.

When the Aircell Axxess II system is installed Airplane Flight Manual Supplement, DB1002072, Revision IR, dated March 22, 2012, or later FAA approved revision, is required.

When the Astronics Cabin Inverter system is installed Airplane Flight Manual Supplement, DB1109012, Revision IR, dated March 22, 2012, or later FAA approved revision, is required.

This system is intended to provide internet connection and email services to the airplane's cabin using portable electronic devices (PEDs). Any other intended function of this equipment will require a re-examination of the certification basis.

When the Rockwell Collins Venue system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1002068, Rev IR, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

When the Aircell Axxess II system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1105003, Rev A, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

When the Astronics Cabin Inverter system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1108018, Rev IR, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

When the Emteq LED Cabin Lighting system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1108019, Rev IR, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.
FAA Form 8110-2-1(10-09) PAGE 2 of 3 PAGES This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758120

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245042

EFTA01261702

SDNY_GM_02758121

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245043

EFTA01261703

United States of America
Department of Transportation - Federal Aviation Administration
Supplemental Type Certificate
(Continuation Sheet)

Number ST03960AT

Certification Basis:

The Certification Basis for this installation is per Type Certificate Data Sheet A16SW original Certification Basis, Federal Aviation Regulation (FAR), Part 25 as amended by Amendment 25-1 through 25-40 plus FAR 25.1335, 25.1351(d), 25.1353(c)(5), and 25.1447 of Amendment 25-41; FAR 25.29, FAR 25.255, and FAR 25.1353(c)(6) of Amendment 25-42; and FAR 25.361(b) and 25.1329(h) of Amendment 25-46.

In addition the following specified rules amendment levels were used for this installation.

25.571 [25-54]	25.1353(a)(c) [25-123]	25.1357 [25-123]	25.1431 [25-113]	25.1529 [25-54]
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[.] Indicates amendment level

-----END-----

SDNY_GM_02758123

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245045

EFTA01261705

UNITED STATES OF AMERICA
DEPARTMENT OF TRANSPORTATION-FEDERAL AVIATION ADMINISTRATION

STANDARD AIRWORTHINESS CERTIFICATE

1 NATIONALITY AND REGISTRATION MARKS N727KB	2 MANUFACTURER AND MODEL RAYTHEON AIRCRAFT CO. BE-400A	3 AIRCRAFT SERIAL NUMBER RK-260	4 CATEGORY TRANSPORT
--	--	--	--------------------------------

5 AUTHORITY AND BASIS FOR ISSUANCE
This airworthiness certificate is issued pursuant to 49 U.S.C. § 44704 and certifies that, as of the date of issuance, the aircraft to which
issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been
shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex E to the
Convention on International Civil Aviation, except as noted herein.
Exemptions:

NONE

6 TERMS AND CONDITIONS
Unless sooner amended, suspended, revoked, or a limitation date is otherwise established by the FAA, this airworthiness certificate
is effective as long as the maintenance, inspection, alterations, and alterations are performed in accordance with Parts 21, 43, and
39 of the Federal Aviation Regulations and the aircraft is registered in the United States.

DATE OF ISSUANCE R 11 DEC 1999	FAA REPRESENTATIVE DANIEL J. MICHAELSEN	DESIGNATION NUMBER CE-01
--	---	------------------------------------

Any alteration, reproduction, or reuse of this certificate may be punishable by a fine not exceeding \$1,000 or imprisonment not exceeding 1 year, or both.
THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.
FAA Form 8100-2 (04-11) Supersedes Previous Edition

SDNY_GM_02758124

T TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15,

EFTA_00245046

SDNY_GM_02758125

T TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15,

EFTA_00245047

UNITED STATES OF AMERICA
DEPARTMENT OF TRANSPORTATION-FEDERAL AVIATION ADMINISTRATION
STANDARD AIRWORTHINESS CERTIFICATE

1 NATIONALITY AND REGISTRATION MARKS N727KC	2 MANUFACTURER AND MODEL RAYTHEON AIRCRAFT CO BE-400A	3 AIRCRAFT SERIAL NUMBER RK-260	4 CATEGORY TRANSPORT
---	--	---	--------------------------------

5 AUTHORITY AND BASIS FOR ISSUANCE
This airworthiness certificate is issued pursuant to 49 U.S.C. § 44704 and certifies that, as of the date of issuance, the aircraft to which it has been inspected and found to conform to the type certificate number, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein.
Exemptions:

NONE

6 TERMS AND CONDITIONS
Unless sooner suspended, suspended, revoked, or a termination date is otherwise established by the FAA, this airworthiness certificate is effective as long as the requirements, prescriptive maintenance, and alterations are performed in accordance with Parts 21, 43, and 89 of the Federal Aviation Regulations, as applicable, and the aircraft is registered in the United States.

DATE OF ISSUANCE R 12/11/1999	FAA REPRESENTATIVE DEREK H. JACKSON 	DESIGNATION NUMBER AEA-FSDD-68
---	---	--

Any alteration, reproduction, or reuse of this certificate may be punishable by a fine not exceeding \$1,000 or imprisonment not exceeding 3 years or both.
THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.
FAA Form 8130-2 (04-11) Supersedes Previous Edition

SDNY_GM_02758126

↑ TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15,

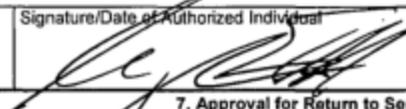
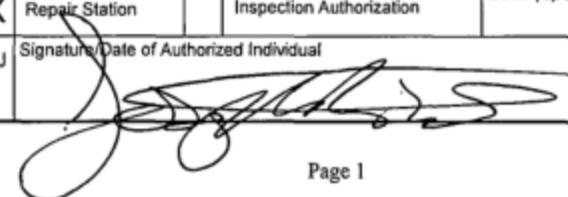
EFTA_00245048



SDNY_GM_02758127

Γ TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15.

EFTA_00245049

 US Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		OMB No. 2120-0020 Exp: 8/31/2014	Electronic Tracking Number
				For FAA Use Only
INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))				
1. Aircraft	Nationality and Registration Mark N727KG		Serial No. RK-260	
	Make RAYTHEON AIRCRAFT COMPANY		Model 400A	Series
2. Owner	Name (As shown on registration certificate) STONY POINT I LLC		Address (As shown on registration certificate) Address 1 W PACK SQ STE 305 City ASHEVILLE State NORTH CAROLINA Zip 28801-3419 Country UNITED STATES	
	3. For FAA Use Only			
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name CONSTANT AVIATION Address 8211 SECONDARY RD. City CLEVELAND State OHIO Zip 44135 Country UNITED STATES		<input type="checkbox"/> U. S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Manufacturer <input type="checkbox"/> Certified Maintenance Organization		
		C. Certificate No. WC7R346J		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual  3/12/2015		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> Approved <input type="checkbox"/> Rejected				
BY	<input type="checkbox"/> FAA Flt. Standards Inspector	<input type="checkbox"/> Manufacturer	<input type="checkbox"/> Maintenance Organization	<input type="checkbox"/> Persons Approved by Canadian Department of Transport
	<input checked="" type="checkbox"/> FAA Designee	<input checked="" type="checkbox"/> Repair Station	<input type="checkbox"/> Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS#WC7R346J		Signature/Date of Authorized Individual  3/12/2015		

FAA Form 337 (10/06)

Page 1

SDNY_GM_02758128

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245050

EFTA01261710

SDNY_GM_02758129

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245051

EFTA01261711

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N727KG

3/12/2015

Nationality and Registration Mark

Date

INSTALLED BEECH 400/400A/400T SERIES CABIN INTERIOR NOISE REDUCTION SYSTEM IN ACCORDANCE WITH ENGINEERING DRAWING N23011100-501 APPROVED BY EDWARD KATS ON FAA FORM 8110-3 DATED MARCH 10, 2015.

COMPLETED WEIGHT AND BALANCE OF AIRCRAFT

INSTRUCTIONS FOR CONTINUED AIRWORTHINESS: INSPECT AIRCRAFT IN ACCORDANCE WITH MANUFACTURES SPECIFICATIONS, NO SPECIAL INSPECTION REQUIRED

Additional Sheets Are Attached

SDNY_GM_02758131

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245053

EFTA01261713

DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION				I. DATE
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				March 10, 2015
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (<i>Airplane, Radio, Helicopter, etc.</i>) Airplane	5. NAME OF APPLICANT Constant Aviation	
LIST OF DATA				
6. IDENTIFICATION		7. TITLE		
N2301, Revision D Dated 06/30/2014		Master Data List		
D10095, Revision (-) Dated 09/05/2013		Beech 400/400A/400T Series Cabin Interior Noise Reduction System		
D10081, Revision A Dated 09/05/2013 -----end-----		Structural Substantiation Report for N23011100-501 Installation		
		N23011100-501 Weight and Balance Analysis		
		NOTES:		
		1. This approval indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements". Compliance to the additional regulations not listed here may be required. This form does not constitute FAA approval of all the data necessary for the substantiation of compliance to the necessary requirements for the entire alteration		
		2. This approval is applicable to Beechcraft 400A aircraft, S/N RK-260, N727KG only		
		3. This approval covers structural aspects only		
		4. This approval covers engineering data only and is not an installation approval		
		End-----end-----end-----end		
8. PURPOSE OF DATA To show compliance with applicable Federal Airworthiness Regulations for the alteration performed on Beechcraft 400A aircraft, S/N RK-260, N727KG only.				
9. APPLICABLE REQUIREMENTS (<i>List specific sections</i>) 14 CFR Part 25 §§ 25.23(b), Amdt 25-0; 25.29(b), Amdt 25-0; 25.301, Amdt 25-23; 25.303, Amdt 25-23; 25.305(a)(b), Amdt 25-86; 25.307(a), Amdt 25-23; 25.561(a)(b)(c), Amdt 25-23; 25.601, Amdt 25-0; 25.603, Amdt 25-38; 25.605, Amdt 25-0; 25.609, Amdt 25-0; 25.611, Amdt 25-23; 25.613, Amdt 25-0; 25-619, Amdt 25-23; 25-623, Amdt 25-0; 25-625, Amdt 25-23				
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and attached sheets numbered <u>none</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed.				
I (We) Therefore		<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data		
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBER(S)	13. CLASSIFICATION (S)	
Edward Kats <i>Edward Kats</i>		DERT-605279-NM	Structures	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758132

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245054

EFTA01261714

SDNY_GM_02758133

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245055

EFTA01261715



MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
2/28/2011

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))

1. Aircraft	Nationality and Registration Mark N727KG	Serial No. RK-260	
	Make RAYTHEON AIRCRAFT COMPANY	Model 400A	Series

2. Owner	Name (As shown on registration certificate) STONY POINT 1 LLC	Address (As shown on registration certificate) Address 1 W PACK SQ STE 305	
		City ASHEVILLE	State NC
		Zip 28801-3419	Country UNITED STATES

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency	
Name <u>Constant Aviation</u>		<input type="checkbox"/> U.S. Certificated Mechanic	<input type="checkbox"/> Manufacturer
Address <u>5211 Secondary Rd.</u>		<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.
City <u>Cleveland</u>	State <u>Ohio</u>	<input checked="" type="checkbox"/> Certificated Repair Station	CRS WC7R346J
Zip <u>44135</u>	Country <u>USA</u>	<input type="checkbox"/> Certificated Maintenance Organization	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>	Signature/Date of Authorized Individual <u>MARK D. SANDER JR</u>	Date: <u>3/12/15</u>
--	---	----------------------

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	<input type="checkbox"/> FAA Fit Standards Inspector	<input type="checkbox"/> Manufacturer	<input type="checkbox"/> Maintenance Organization	<input type="checkbox"/> Person Approved by Canadian Department of Transport
	<input type="checkbox"/> FAA Designee	<input checked="" type="checkbox"/> Repair Station	<input type="checkbox"/> Inspection Authorization	Other (Specify)

Certificate or Designation No. CRS WC7R346J	Signature/Date of Authorized Individual <u>[Signature]</u>	Date: <u>3-17-15</u>
--	---	----------------------

FAA Form 337 (10-06)

SDNY_GM_02758134

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245056

EFTA01261716

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N727KG

03/12/2015

Nationality and Registration Mark

Date

Accomplished installation of a Standby Instrument in accordance with Nextant Aerospace STC# ST10959SC per Master Drawing List STC Document (Report: 373-00-0001 Rev EF.13, 06/19/2014) with approved deviations as follows:

Equipment removed:

Smith-Industries-Standby-Altimeter-(Part-Number-1611AM10)-Smith-Industries-Standby-Airspeed-Indicator-(Part-Number-276AS1)-and-L3-Standby-Altitude-Indicator (Part Number: 501-1412-03).

Equipment installed:

Mid-Continent MD302 Standby Instrument (Part Number: 6420302-5).

Structural Deviations:

"Install, MD-302 STBY Instrument Upgrade" Drawing (NX311-10010 Rev IR, 02/20/2015) and "LH FWD Circuit Breaker Panel Fabrication" Engineering Order (CA-EO-0315-02 Rev IR, 03/04/2015) approved by Paul Hedding (DERT-410115-CE) on FAA Form 8110-3 dated: 03/04/2015.

Performed satisfactory Ground Test Procedure in accordance with STC Document (373-00-0051 Rev C, 12/09/2013).

Provided Instructions for Continued Airworthiness STC Document (373-00-0055 Rev B, 11/18/2013).

Provided Airplane Flight Manual Supplement STC Document (373-00-0048 Rev IR, 12/16/2013).

Aircraft Equipment List updated and Weight & Balance revised to reflect these changes.

END

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758135

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245057

EFTA01261717



**MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)**

Form Approved
OMB No. 2120-0020
2/28/2011

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))

1. Aircraft	Nationality and Registration Mark N727KG	Serial No. RK-260	
	Make RAYTHEON AIRCRAFT COMPANY	Model 400A	Series
2. Owner	Name (As shown on registration certificate) STONY POINT 1 LLC		Address (As shown on registration certificate) 1 W PACK SQ STE 305
			City ASHEVILLE State NC
			Zip 28801-3419 Country UNITED STATES

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency	
Name Constant Aviation		<input type="checkbox"/> U.S. Certificated Mechanic	<input type="checkbox"/> Manufacturer
Address 5211 Secondary Rd.		<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.
City Cleveland	State Ohio	<input checked="" type="checkbox"/> Certificated Repair Station	CRS WC7R346J
Zip 44135	Country USA	<input type="checkbox"/> Certificated Maintenance Organization	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>	Signature/Date of Authorized Individual <i>Mark D. Smydee Jr.</i> Date: 3/12/15
--	---

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	<input type="checkbox"/> FAA Fit Standards Inspector	<input type="checkbox"/> Manufacturer	<input type="checkbox"/> Maintenance Organization	<input type="checkbox"/> Person Approved by Canadian Department of Transport
	<input type="checkbox"/> FAA Designee	<input checked="" type="checkbox"/> Repair Station	<input type="checkbox"/> Inspection Authorization	Other (Specify)

Certificate or Designation No. CRS WC7R346J	Signature/Date of Authorized Individual <i>[Signature]</i> Date: 3-12-15
---	--

FAA Form 337 (10-06)

SDNY_GM_02758136

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245058

EFTA01261718

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N727KG

03/12/2015

Nationality and Registration Mark

Date

Accomplished modification of Glareshield to incorporate LED Annunciators in accordance with Nextant Aerospace STC# ST10959SC per Master Drawing List STC Document (Report: 373-00-0001 Rev EF.13, 06/19/2014).

Performed satisfactory Ground Test Procedure in accordance with STC Document (373-00-0058 Rev A, 07/25/2013).

No change to Equipment List or Weight & Balance.

END

Additional Sheets Are Attached

FAA Form 337 (10-08)

SDNY_GM_02758137

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245059

EFTA01261719

nextant aerospace

//REIMAGINED //REBUILT //REBORN

Date: March 12, 2015

To: Constant Aviation,LLC

From: Nextant Aerospace,LLC

Subject: Supplemental Type Certificate and Nextant Aerospace Sound Kit authorization for 400A serial number RK-260 (N727KG)

This letter serves to provide evidence that the listed Supplemental Type Certificate and Nextant Aerospace Sound Kit on a single installation basis by entities listed in this letter. This also approves Constant Aviation to purchase all articles needed to complete the installation and modifications documented in the Master Drawing Lists. All change requests, invoices, and amendments to the purchase agreement must be approved by Nextant Aerospace. Nextant Aerospace Supplier quality provisions 1, 4, 5, 8 apply for this sub contracted maintenance event.

Supplemental Type Certificate (s)

1. ST1059SC (Proline 21) for Installation of MD-302 System
 - a. Master Drawing List: 373-00-0001 MDL (Master Drawing List) Proline 21 Rev EF.13 later approved by Nextant Aerospace
2. ST1059SC (Proline 21) for installation of Glareshield modification and installation
 - a. Master Drawing List: 373-00-0001 MDL (Master Drawing List) Proline 21 Rev EF.13 later approved by Nextant Aerospace

Nextant Aerospace Sound Kit

1. N23011100-501-1
 - a. Master Kiting listing N23011100-501KL dated 2014-01-17
 - b. FAA 8110-3 dated March 10, 2015
2. N23011100-501-2
 - a. Master Kiting listing N23011100-501KL dated 2014-01-17
 - b. FAA 8110-3 dated March 10, 2015

Retain a copy of this Authorization in your aircraft records.

A copy of this authorization letter is to be retained in all aircraft records, as applicable.

Agreement acceptance By:

Aircraft Details

Name: _____

Registration Number : _____

Company: _____

Aircraft hours: _____

Date: _____

Aircraft Cycles: _____

Note: A signed copy must be returned to Nextant Aerospace

SDNY_GM_02758138

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245060

EFTA01261720

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758139

EFTA_00245061

EFTA01261721

United States of America
Department of Transportation -- Federal Aviation Administration
Supplemental Type Certificate

Number ST10959SC

This certificate issued to Nextant Aerospace LLC
26180 Curtiss-Wright Pkwy.
Richmond Heights, OH 44143

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified herein meets the airworthiness requirements of Part 25 of the Federal Aviation Regulations. (See Pages 3 and 4 for complete certification basis.)

Original Product - Type Certificate Number A16SW
Make Hawker Beechcraft Corp.
Model 400A

Description of Type Design Change Installation of Rockwell Collins Pro Line 21 Electronic Flight Instrument System (EFIS) with Rockwell Collins FMS-6100, Localizer Performance with Vertical Guidance (LPV) approach capability and Universal Avionics Systems Company (UASC) Terrain Awareness Warning System (TAWS) in accordance with Nextant Aerospace LLC Master Drawing List Doc. No. 373-00-0001 Revision W, dated March 29, 2012, or later FAA approved version.

Limitations and Conditions:

- 1) The installer must determine whether this design change is compatible with previously approved modifications.
- 2) If the holder agrees to permit another person to use this certificate to alter a product, the holder must give that person written evidence of that permission.
- 3) For aircraft equipped with Pratt & Whitney JT15D Engines FAA approved Airplane Flight Manual Supplement, Nextant Aerospace LLC Doc. No. 373-00-0008 Revision IR, dated 10/19/2009, or later FAA approved version, is required on board the modified aircraft.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: June 26, 2008

Date issued:

Date of issuance: October 19, 2009

Date amended: September 26, 2011; April 05, 2012



By directing of the Administrator

(Signature)

Steven L. Lardinois
Manager, Systems and Flight Test Branch
Chicago Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2(10-08) PAGE 1 of 4 PAGES

This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758140

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245062

EFTA01261722

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758141

EFTA_00245063

EFTA01261723

United States of America
Department of Transportation - Federal Aviation Administration
Supplemental Type Certificate
(Continuation Sheet)

Number ST10959SC

Date of issuance: October 19, 2009

Date Amended: September 26, 2011; April 05, 2012

Limitations and Conditions (continued):

- 4) For aircraft equipped with Williams FJ44-3AP Engines, those engines must be installed under STC ST02371LA.
- 5) For aircraft equipped with Williams FJ44-3AP Engines FAA approved Airplane Flight Manual Supplement, Nextant Aerospace LLC Doc. No. 373-00-0023 Revision IR, dated September 2, 2011, or later FAA approved version, is required on board the modified aircraft.
- 6) Aircraft must have previously qualified for Operations in Reduced Vertical Airspace via:
RK-1 thru RK-117 and RK-119 thru RK-139, Hawker Beechcraft Service Bulletin No. 34-3431.
RK-118, RK-140 thru RK-224, Hawker Beechcraft Service Bulletin Nos. 34-3228 and 34-3431.
RK-225 thru RK-299, Hawker Beechcraft Service Bulletin No. 34-3431.
RK-300 and after as original equipped from Hawker Beechcraft.
- 7) For aircraft equipped with Universal Avionics Systems Corporation (UASC) TAWS FAA approved Airplane Flight Manual Supplement, Nextant Aerospace LLC Doc. No. 373-00-0036 Revision IR, approved March 28, 2012, or later FAA approved version, is required on board the modified aircraft.
- 8) For aircraft with Collins Proline 21 FMS-6100 LPV Approach enabled FAA approved Airplane Flight Manual Supplement, Nextant Aerospace LLC Doc. No. 373-00-0032 Revision IR, approved March 28, 2012, or later FAA approved version, is required on board the modified aircraft.

Certification Basis:

Based on 14 CFR §§21.115 and 21.101, and the FAA policy for significant changes in FAA Order 8110.48, the certification basis for the Hawker Beechcraft 400A aircraft is as follows:

- a. The type certification basis for the Hawker Beechcraft 400A aircraft is shown on TCDS A16SW for parts not changed or not affected by this change.
- b. The certification basis for the parts changed or affected by this change since the reference date of application, September 29, 2009, is based on TCDS A16SW and §§ 14 CFR Part 25 as shown on page 4 of 4 of this STC.

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8130-2-1(10-69)

PAGE 3 OF 4 PAGES

This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758142

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245064

EFTA01261724

SDNY_GM_02758143

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245065

EFTA01261725

United States of America
 Department of Transportation - Federal Aviation Administration
Supplemental Type Certificate
 (Continuation Sheet)

Date of issuance: October 19, 2009
 Date Amended: September 26, 2011; April 05, 2012

Based on 14 CFR §§ 21.115 and 21.101, and the FAA policy for significant changes in FAA Order 8110.48, the certification basis for the baseline installation of Rockwell Collins Pro Line 21 Electronic Flight Instrument System (EFIS) with Rockwell Collins FMS-6100 modification was determined to be TCDS A16SW and the following later regulations:

Amdt.[24-4] 25.771 (a,e)	Amdt.[25-23] 25.301; 25.303; 25.581; 25.607; 25.611, 25.773 (a)(1)(2)(d)	Amdt.[25-38] 25.1309 (n)(b)(c)(d)(e)(g); 25.1322	Amdt.[25-40] 25.901 (b)(1)(i); 25.1549 (a)(b)(c)	Amdt.[25-41] 25.1321 (a)(b)(c)(e); 25.1331; 25.1333
Amdt.[25-42] 25.1501	Amdt.[25-46] 25.603; 25.605; 25.613; 25.1329 (h)	Amdt.[25-54] 25.1529	Amdt.[25-72] 25.307; 25.571, 25.1307 (e)(e); 25.1351 (a)(1)(d); 25.1381; 25.1521 (a)(c); 25.1543 (b); 25.1581; 25.1583 (a)	Amdt.[25-80] 25.1316
Amdt.[25-86] 25.305	Amdt.[25-90] 25.1303	Amdt.[25-91] 25.561	Amdt.[25-102] 25.981 (a)(b)	Amdt.[25-105] 25.1585 (a)
Amdt.[25-108] 25.1325 (d)	Amdt.[25-109] 25.1323 (a)(f)(g)	Amdt.[25-113] 25.869 (a)(4); 25.1431	Amdt.[25-122] 25.1317 (a)(b)(c)	Amdt.[25-123] 25.1353 (a)(c)

Based on 14 CFR §§ 21.115 and 21.101, and the FAA policy for significant changes in FAA Order 8110.48, the certification basis for the Collins Proline 21 FMS-6100 with LPV Approach enabled and UASC TAWS was determined to be TCDS A16SW and the following later regulations:

Amdt.[Original] 25.601, 25.605, 25.609, 25.613, 25.1301, 25.1357, 25.1381, 25.1525, 25.1541, 25.1581	Amdt.[24-4] 25.771 (a)	Amdt.[25-23] 25.301; 25.303; 25.305, 25.307, 25.561(c); 25.607; 25.611, 25.1307	Amdt.[25-38] 25.603, 25.1322, 25.1583(e)	Amdt.[25-40] 25.1585 (a)
Amdt.[25-41] 25.1309 (a)(b)(c)(d)(e)(g) 25.1351 (a)(1)	Amdt.[25-42] 25.1501	Amdt.[25-46] 25.777(a)(c)	Amdt.[25-54] 25.1529	Amdt.[25-72] 25.869(a)(1)
Amdt.[25-102] 25.981(a)(b)	Amdt.[25-113] 25.1431(a)	Amdt.[25-123] 25.1353 (a)(c)		

- END -

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.
 FAA FORM 8110-2-1(10-69) PAGE 4 OF 4 PAGES This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758144

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245066

EFTA01261726

SDNY_GM_02758145

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245067

EFTA01261727

Nextant Aerospace LLC
355 Richmond Rd.
Richmond Heights OH 44143

Report No.: 373-00-0001
Revision EF.13

Master Drawing List

Document Number: 373-00-0001

Revision: EF.13 2/24/2015

Upgrade of Collins Proline 4 to Proline 21 Avionics System

Hawker Beech 400A Aircraft

FAA STC ST10959SC

Approved By



Date 6-19-2014

Page 1 of 17

SDNY_GM_02758146

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245068

EFTA01261728

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

REVISIONS			
LEVEL	DESCRIPTION	APPROVED	DATE
IR	Initial Release	TT	-
A	Minor Changes	TT	2/25/09
B	Major Rewrite	DLB	6/29/09
C	Minor Changes	DLB	7/7/09
D	Minor Changes	DLB	9/2/09
E	Minor Changes	DLB	9/18/09
F	Project Clean up	DLB	10/12/09
G	Revised SSA/CCA/DCU Config documents.	DLB	10/16/09
H	Project Clean up	DLB	11/24/09
I	Drawing Revisions, Aircraft Serial Number Applicability added	DLB	9/7/10
J	Addition of FJ44-3AP Engine Instrument display capability	DLB	9/23/10
K	Corrected FJ44-3AP drawing callouts	DLB	10/27/10
L	Revised Certification Plan 373-00-0028 Revised drawings: Table 1: 373-91-5000 Table 8A: 373-77-4100 373-77-4100-01 373-77-4200 373-77-4200-01 373-34-2515 373-34-2535 373-34-2555 373-34-2545 Added drawing 373-33-5005 to Table 8A	DLB	11/5/10
M	Updated MDL to indicate new and revised documents and drawings for amendment SA7464CH-T.	DLB	12/15/10
N	Revised DCU Config 373-00-0026 Added CSU-4100 drawing 373-42-0001.	DLB	1-24-2011

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

REVISIONS (CONTINUED)			
LEVEL	DESCRIPTION	APPROVED	DATE
O	Revised EIS GTP 373-00-0024 Revised DCU Config 373-00-0026 Revised RVSM addendum 373-00-0029 Revised drawings: Table 1: 373-42-0001 Table 8A: 373-77-4100 373-77-4100-01 373-77-4200 373-77-4200-01 373-34-2515 373-34-2535 373-34-2555 373-34-2545	DLB	5-31-2011
P	Revised EI Analysis 373-00-0025 Revised ELA 373-00-0022 Revised SSA 373-00-0027 Revised drawings: Table 1: 373-34-2510 373-34-2510-01 373-34-2530 373-34-2530-01 373-34-2540 373-34-2540-01 373-34-2550 373-34-2550-01 Table 8A: 373-34-2515 373-34-2535 373-34-2555 373-34-2545	DLB	7-26-2011
Q	Added software and part number note to Table 8A. Added Software documents to Supporting Data table. Minor Change submittal of drawings 373-77-4001-01 and 373046-2002-01.	DLB	9-20-2011
R	Re-formatted MDL for aircraft effectivity.	DLB	9-26-2011

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

REVISIONS (CONTINUED)			
LEVEL	DESCRIPTION	APPROVED	DATE
S	Added STC Amendment Project SA7659CH-T documents and revisions	DLB	10-27-2011
T	Revised documents: 373-00-0031 Cert Plan 373-00-0040 Struct Anal 373-91-6000 TAWS Installation	DLB	12-29-2011
U	Added Table 11.	DLB	1-5-2012
V	Revised Drawings: 373-34-2510 373-34-2510-01 373-34-5101 373-34-5102 373-34-4650 Revised Reports: 373-00-0034 373-00-0033 373-00-0038 373-00-0037 373-00-0043 Added Report: 373-00-0046	DLB	2-8-2012
W	Revised Reports: 373-00-0039 373-00-0044	DLB	3-29-2012
X	Minor Drawing Changes	DG	5/2/2012
Y	Added 373-22-1101, 373-22-1102, 373-22-1200, 373-22-1200-01, Minor drawing changes	DG	7/5/2012
Z	Minor Drawing Changes Amendment to add MD-302, TS835, Glareshield Annunciators.	DLB	6/20/2013
AA	Removed TS835 EPS.	DLB	7/1/2013
BB	Revised MD302 and Glareshield Annun GTPs.	DLB	7/30/2013
CC	Removed MD-302 Standby Attitude Module.	DLB	8/15/2013
DD	Added MD-302 Standby Attitude Module.	DLB	11/18/2013
EE	Revised MD302 GTP 373-00-0051.	DLB	12/9/2013

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

EF	Revised ICA 373-00-0002, RVSM Report 373-00-0029. Revised/added drawings marked at Issue EF.	<i>[Signature]</i>	6-19-14
EF.01	Chg 373-22-1200-01 - WD APP, 373-34-5021 - WD NO1 ADF		9/8/2014
EF.02	Chg 373-34-4510, 373-34-4510-01 - WD TCAS, 373-34- 4511-TCAS-RELOCATION		9/9/2014
EF.03	Chg 373-00-0004 Production GTP,		9/17/2014
EF.04	Add NX311-10003, NX311-10003-01 - Nutplate Spacers, Instrument Panel		9/23/2014
EF.05	Chg WD #1 PFD chg GND to Float GND (ECR 1865)		9/26/2014
EF.06	CHG 373-34-5022 WD #2 ADF (ECR 1907)		10/1/2014
EF.07	Chg 373-42-1007, 373-42-1007-01		10/2/2014
EF.08	Chg 373-23-5001, 373-23-5001-01		10/7/2014
EF.09	Chg NX316-10001-01, add NX110-10004, NX110-10005, moved NX315-10005 to table 12.		10/30/2014
EF.10	Chg NX110-10005		10/31/2014
EF.11	Chg 373-77-4200-01		1/15/2015
EF.12	Chg 373-34-7001 and 373-46-2001-01		1/27/2015
EF.13	Chg 373-42-1002, 373-42-1002-01, 373-42-1007, 373-42- 1007-01		2/24/2015

The latest revision letter of this document will be shown in the upper right hand corner of each page below the document number. The entire document will be repaginated and re-released each time revisions are made.

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

SUBSTANTIATION DATA

Documents applicable to all aircraft:

373-00-0001	EF.13	Master Drawing List
373-00-0002	D.01	Instructions for Continued Airworthiness
373-00-0045	A	Ground Test Plan, EMI

Documents applicable to Pratt & Whitney JT-15D engine equipped aircraft:

373-00-0003	IR	Electrical Load Analysis
373-00-0004	C	Systems Ground Test Plan
373-00-0005	A	Collins Proline 4 to 21 Function Hazard Assessment
373-00-0006	C	Collins Proline 4 to 21 System Safety Assessment
373-00-0007	A	Collins Proline 4 to 21 Common Cause Analysis
373-00-0008	IR	Airplane Flight Manual Supplement
373-00-0009	B	Flight Test Plan
373-00-0013	B	Ground Test Plan, Engine Instrument System
373-00-0014	IR	Engine Instrument Comparison Analysis
373-00-0015	B	Collins Proline 21 DCU Configuration Report
373-00-0016	IR	Structural Analysis - Upgrade of Collins Proline 4 to Proline 21 Avionics System
373-00-0017	IR	Airspeed Analysis
373-00-0018	IR	RVSM Substantiation Report
373-00-0019	E	Project Specific Certification Plan
373-00-0020	IR	DCU Structural Load Test Plan
373-93-7000-3-A	A	Structural Analysis - Transponder Antennas Installation
373-93-7000-7-A	IR	Structural Analysis - TCAS Antenna Installation
373-93-7000-9-A	IR	Structural Analysis - GPS/XM Antenna Installation
373-91-4000-21-A	IR	Structural Analysis - Forward Pressure Bulkhead Feed Through Plate Installation

Documents applicable to Williams FJ44-3AP engine equipped aircraft:

373-00-0022	A	Electrical Load Analysis– Williams FJ44-3AP Equipped
373-00-0023	A	Airplane Flight Manual Supplement– Williams FJ44-3AP Equipped
373-00-0024	F	Ground Test Plan, Engine Instrument System – Williams FJ44-3AP Equipped
373-00-0025	B	Engine Instrument Comparison Analysis– Williams FJ44-3AP Equipped
373-00-0026	C	Engine Data Concentrator Configuration Report – Williams FJ44-3AP Equipped
373-00-0027	A	System Safety Assessment Addendum – Williams FJ44-3AP Equipped
373-00-0028	D	Certification Plan/Compliance Checklist Report – Amendment to add Williams FJ44-3AP Engine Instrument Display Option
373-00-0029	B	Addendum to RVSM Substantiation Report 373-00-0018
373-00-0030	IR	Structural Analysis – Addendum to 373-00-0016
373-00-0058	A	Ground Test Plan, Glareshield Annunciators
373-00-0061	IR	Instructions for Continued Airworthiness – Glareshield Annunciators

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

SUBSTANTIATION DATA (cont.)

Aircraft with FMS-6100 LPV Installed

373-00-0031	F	Certification Plan, LPV/TAWS
373-00-0032	IR	Airplane Flight Manual Supplement, LPV
373-00-0033	A	Flight Test Plan, LPV
373-00-0034	A	Ground Test Plan, LPV
373-00-0035	IR	Collins Proline 21 Function Hazard and System Safety Assessment, LPV
373-00-0046	IR	GPS-4000S Change Impact Analysis

Aircraft with Universal Avionics TAWS A Installed

373-00-0031	F	Certification Plan, LPV/TAWS
373-00-0036	IR	Airplane Flight Manual Supplement, TAWS
373-00-0037	A	Flight Test Plan, TAWS
373-00-0038	A	Ground Test Plan, TAWS
373-00-0039	A	Collins Proline 21 System Safety Assessment, TAWS
373-00-0040	A	Structural Analysis – TAWS Installation
373-00-0041	IR	Instructions for Continued Airworthiness – TAWS Installation
373-00-0042	IR	Electrical Load Analysis– TAWS Installation
373-00-0043	A	Configuration Matrix– TAWS Installation
373-00-0044	A	Collins Proline 21 Function Hazard Assessment, TAWS

Aircraft with Upgraded Glareshield Annunciator Panel

373-00-0047	C	Certification Plan
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PITOT TUBE REPLACEMENT

373-00-0059	IR	Ground Test Plan, Pitot Tube Replacement
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Aircraft with MD-302

SAM Installed

373-00-0062	IR	Certification Plan
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Aircraft with MD-302 Standby Instrument

373-00-0062	IR	Certification Plan
373-00-0048	IR	Airplane Flight Manual Supplement, MD-302
373-00-0050	IR	Flight Test Plan, MD-302
373-00-0051	C	Ground Test Plan, MD-302
373-00-0060	IR	Function Hazard Assessment, MD-302
373-00-0053	IR	System Safety Assessment, MD-302
373-00-0055	B	Instructions for Continued Airworthiness – MD-302
373-00-0057	A	Electrical Load Analysis–MD-302 Installation

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

DESIGN DATA

This project consists of a upgrading the existing Collins Proline 4 CRT Flight Displays to Proline 21 LCD Flight Displays. As part of this STC most of the existing remaining avionics wiring is removed and replaced with new wire. Those wiring diagrams are listed in Table 9. The remaining existing avionics interface remains as originally TC'd.

The existing AHS-85E Attitude Heading Reference System (AHRS) can be upgraded to Collins AHS-3000A per Collins STC ST02173LA in conjunction with this STC. Reference Table 6 for wiring interface drawings to be used with the Proline 4 to 21 upgrade.

The existing Collins RTA-444 Radar can be upgraded to Collins RTA-854 Turbulence Detection Radar per Collins STC ST01474WI-D in conjunction with this STC. Reference Table 7 for wiring interface drawings to be used with the Proline 4 to 21 upgrade.

The aircraft must have dual Collins AMS-5000 Avionics Management Systems installed prior to the incorporation of the Proline 4 to 21 STC. It can be upgraded per Collins STC ST00485WI-D in conjunction with this STC.

The base STC Installation consists of removing the existing Collins Pro Line 4 Integrated Displays (7x7 CRTs) system and installing:

- Three Pro Line 21 Flat Panel Liquid Crystal Displays (8X10 LCD)
- Engine Instrument Display Integration (Pratt & Whitney JT15D-5 (See Table 1A))
- Fuel Quantity Display Integration
- Upgrade of the existing Collins FMC5000 to the FMC6000
- Replacement of Master Warn/Master Caution Panel

These systems will be installed IAW data listed in Table 1

STC Options include:

- Fourth LCD panel as #2 MFD. This system will be installed IAW data listed in Table 5.
- Collins TCAS II. This system will be installed IAW data listed in Table 2.
- XM Satellite receiver. This system will be installed IAW data listed in Table 3.
- Single or Dual IFIS-5000 Integrated Flight Information System (IFIS) This system will be installed IAW data listed in Table 4.
- Williams FJ44-3AP Engine Instrument Display Integration. This system will be installed IAW data listed in Table 8. Engines must be installed under STC ST02371LA.
- Universal Avionics TAWS A. This system will be installed IAW data listed in Table 10.
- FMS-6100 WAAS/LPV Approaches. This system will be installed IAW data listed in Table 11.
- Glareshield Annunciators. This system will be installed IAW data listed in Table 12.
- Midcontinent Instruments MD302. This system will be installed IAW data listed in Table 13.

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

Aircraft Applicability

Do to minor factory wiring variances of aircraft serial number ranges the installer must observe the Aircraft Applicability column for determination of applicable drawings to use.
 The applicability breakdown is as follow:

All
 RK-48 and Below
 RK-49 and Above

All drawings are applicable to either JT15D or FJ44-3AP engine equipped aircraft excluding Table 1A (JT15D aircraft only) and Table 8 (FJ44-3AP aircraft only).

For baseline installation the following drawings in Table 1 should be used with both JT15D and FJ44-3AP engine equipped aircraft:

Table 1				
Drawings	Rev	Issue	Description	Aircraft Applicability
Wiring Diagrams				
373-34-2001	E		W/D NO.1 CCP	ALL
373-34-2510	K	EF.05	W/D NO.1 PFD	RK-48 AND BELOW
373-34-2510-01	J	EF.05	W/D NO.1 PFD	RK-49 AND ABOVE
373-34-2511	D		W/D NO.1 DCP	RK-48 AND BELOW
373-34-2511-01	C		W/D NO.1 DCP	RK-49 AND ABOVE
373-34-2521	E		W/D NO.2 DCP	ALL
373-34-2530	J	Z	W/D NO.1 MFD	RK-48 AND BELOW
373-34-2530-01	E	Z	W/D NO.1 MFD	RK-49 AND ABOVE
373-34-2550	J	Z	W/D NO.2 PFD	RK-48 AND BELOW
373-34-2550-01	F	Z	W/D NO.2 PFD	RK-49 AND ABOVE
373-24-4000	K	EF	W/D AUX BATT TEST	ALL
		EF	SINGLE AUX BAT (PS-835) WIRING	ALL
373-24-4002	A		DIAGRAM	
373-24-5000	G		W/D CKT BKR PNL	RK-48 AND BELOW
373-24-5000-01	E	Z	W/D CKT BKR PNL	RK-49 AND ABOVE
373-73-5001	C		W/D FUEL FLOW CONV	RK-48 AND BELOW
373-73-5001-01	D	EF	W/D FUEL FLOW CONV	RK-49 AND ABOVE
373-28-6001	G		W/D FUEL QTY CONV	RK-48 AND BELOW
373-28-6001-01	B	EF	W/D FUEL QTY CONV	RK-49 AND ABOVE
373-33-1206	H	EF	W/D MC/MW INTERFACE	ALL
373-34-3004	E	EF	REVERSION	ALL
373-39-1301	E		W/D CHP	ALL
373-39-3101	E		W/D NOSE RJB MOD	RK-48 AND BELOW
373-39-3101-01	D		W/D NOSE RJB MOD	RK-49 AND ABOVE
373-91-0000	C		WIRE ROUTING DIAGRAM	ALL
33-11-02-01	IR		COCKPIT LIGHTING ECO	RK-48 AND BELOW
Drawings				
	Rev	Issue	Description	Aircraft Applicability
373-24-3100	E		W/D VOLT AMMETER EXT	RK-48 AND BELOW
373-24-3100-01	A	EF	W/D VOLT AMMETER	RK-49 AND ABOVE

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

			EXT	
373-39-3001	H	EF	W/D SWITCH PANELS	ALL
DB1006007	D	Z	STBY GYRO LIGHTING MOD	RK-049 AND ABOVE
Structural Drawings				
373-91-3203	G	Z	SW PNL ASSY	ALL
373-91-3213	G	Z	SW PNL FAB	ALL
373-91-4110	L	EF	SHELF FAB	ALL
373-91-4100	M	EF	SHELF ASSY	ALL
373-91-4000	K	EF	SHELF INSTL	ALL
373-92-4110	G	Z	INST PNL FAB	ALL
373-92-4100	C		INST PNL ASSY	ALL
373-92-4110-3	H	EF	ANGLE, INST PNL	ALL
373-92-4110-5	H	EF	ANGLE, INST PNL	ALL
373-92-4110-7	H	EF	ANGLE, INST PNL	ALL
373-92-4110-8	B	EF	PNL, INST, FO STD	RK-49 AND ABOVE, OPTIONAL
373-92-4110-10	B	EF	PNL, INST, XT-DOM	RK-49 AND ABOVE, OPTIONAL
373-92-4110-14	A	EF	PNL, INST, XT-INT	RK-49 AND ABOVE, OPTIONAL
373-92-4110-21	J	EF	ANGLE, INST PNL	ALL
373-92-4110-23	H	EF	RACK, TJ BLOCK	ALL
NX311-10003	A	EF.04	ASSY, NUTPLATE SPACER	RK-1 THRU RK-999
NX311-10003-01	A	EF.04	PLATE, NUTPLATE SPACER	RK-1 THRU RK-999
NX311-10004	A	EF	MOD, AFD-3010	ALL
373-92-4000	F		INST PNL INSTL	ALL
373-93-1501	C		SPEED SWITCH ASSY	ALL
373-93-1511	D		SPEED SWITCH FAB	ALL
373-93-1511-9	IR		IAS/VS SWITCH OVERLAY	ALL
373-94-2110	IR		OVHD PNL MOD FAB	ALL
373-94-2000	A		OVHD PNL MOD ASSY & INSTL	ALL
373-94-1110	D	EF	PEDESTAL EXT FAB	ALL
373-94-1100	C	EF	PEDESTAL EXT ASSY	ALL
373-94-1000	D	EF	PEDESTAL EXT INSTL	ALL
373-97-1000	IR		FWD PEDESTAL MODIFICATION	ALL
373-91-5000	B	EF	ALTERNATE GPS/STBY BATT SHELF INSTL	ALL
373-91-5100	A	EF	ALTERNATE GPS/STBY BATT SHELF ASSY	ALL
NX230-10001-03	B	EF	PLATE, PILOTS MIC JACK PNL	ALL
NX311-10002-01	B	EF	WEDGE, INST PNL	ALL
EO 024-11	IR		PITOT PROBE	ALL, OPTIONAL.

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

373-24-4003	IR	EF	STANDBY BATTERY CIRCUIT BREAKER INSTALLATION	ALL
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Table 1A should be used with both JT15D engine equipped aircraft:

Table 1A

Drawings applicable to Pratt & Whitney JT15D engine equipped aircraft only:

Drawings	Rev	Issue	Description	Aircraft Applicability
373-77-4001	G	Y	W/D NO.1 DCU	RK-48 AND BELOW
373-77-4001-01	B		W/D NO.1 DCU	RK-49 AND ABOVE
373-77-4002	F		W/D NO.2 DCU	RK-48 AND BELOW
373-77-4002-01	A		W/D NO.2 DCU	RK-49 AND ABOVE
373-77-4003	E		W/D NO.1 EDC	RK-48 AND BELOW
373-77-4003-01	B		W/D NO.1 EDC	RK-49 AND ABOVE
373-77-4004	E		W/D NO.2 EDC	RK-48 AND BELOW
373-77-4004-01	A		W/D NO.2 EDC	RK-49 AND ABOVE
373-42-0000	IR		CSU-4100 STRAPPING	ALL
373-77-7001	IR		OIL PRESS. TRANS. INSTL	ALL

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

For any installation including optional installation TCAS II the following drawings in Table 2 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 2				
Drawings	Rev		Description	Aircraft applicability
Wiring Diagrams				
373-34-4510	F	EF.02	W/D TCAS	RK-48 AND BELOW
373-34-4510-01	B	EF.02	W/D TCAS	RK-49 AND ABOVE
373-34-4550	D	EF	W/D TCAS INT (TTR-920 OR TTR-4000)	ALL
373-34-5010	E		W/D NO.1 TDR	RK-48 AND BELOW
373-34-5010-01	B		W/D NO.1 TDR	RK-49 AND ABOVE
373-34-5011	E		W/D NO.2 TDR	ALL
Structural Drawings				
373-34-4511	IR	EF.02	TCAS INSTL/RELOCATE	ALL
373-93-7110	D	Z	ANTENNA FAB	ALL
373-93-7000	E	Z	ANTENNA INSTL	ALL

For any installation including optional installation XM satellite receiver the following drawings in Table 3 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 3				
Drawings	Rev		Description	Aircraft applicability
Wiring Diagrams				
373-34-7001	G	EF.12	W/D XM SAT REC	ALL
Structural Drawings				
373-93-7110	D	Z	ANTENNA FAB	ALL
373-93-7000	E	Z	ANTENNA INSTL	ALL

For any installation including optional installation Single or Dual IFIS-5000 the following drawings in Table 4 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 4				
Drawings	Rev		Description	Aircraft applicability
Wiring Diagrams				
373-46-2001	J	Z	W/D NO.1 FSU	RK-48 AND BELOW
373-46-2001-01	E	EF.12	W/D NO.1 FSU	RK-49 AND AFTER
373-46-2002	H	Z	W/D NO.2 FSU	RK-48 AND BELOW
373-46-2002-01	D	Z	W/D NO.2 FSU	RK-49 AND ABOVE

For any installation including optional installation of #2 MFD the following drawings in Table 5 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 5				
Drawings	Rev		Description	Aircraft applicability
Wiring Diagrams				
373-34-2540	J	Z	W/D 2MFD	RK-48 AND BELOW
373-34-2540-01	E	Z	W/D 2MFD	RK-49 AND ABOVE
373-34-2002	E		W/D NO.2 CCP	ALL

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For any installation upgrading the existing AHS-85E Attitude Heading Reference System (AHRS) to Collins AHS-3000A per Collins STC ST02173LA in conjunction with this STC the following drawings in Table 6 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 6

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-34-2003	D		W/D NO.1 AHC	RK-48 AND BELOW
373-34-2003-01	B		W/D NO.1 AHC	RK-49 AND ABOVE
373-34-2004	D		W/D NO.2 AHC	RK-48 AND BELOW
373-34-2004-01	B		W/D NO.2 AHC	RK-49 AND ABOVE

For any installation upgrading the existing Collins RTA-444 Radar to Collins RTA-854 Turbulence Detection Radar per Collins STC ST01474WI-D in conjunction with this STC the following drawings in Table 7 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 7

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-34-4801	C		W/D WX RADAR	ALL

For any aircraft equipped with Williams FJ44-3AP Engines installed under STC ST02371LA the following drawings in Table 8 should be used. Table 8 drawings should be used in addition to the drawings included in the Table 1 and any other optional installations:

NOTE: The below design data in Table 8 is approved for use with Collins Proline 21 AFD-3010(E) MR 8.2 software only. This software is defined as AFD-3010 part number 822-1084-416 and AFD-3010E part number 822-1753-416 only.

The below design data in Table 8 is approved for use with Collins Proline 21 DCU-3001C V7.0 software only. This software is defined as DCU-3001C part number 822-2362-003.

The below design data in Table 8 is approved for use with Collins Proline 21 IOC-4100 MR V8.0 software only. This software is defined as IOC-4100 part number 822-1362-511.

Table 8

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-77-4100	F	EF	W/D LH FADEC INT. AND NO. 1 DCU	RK-48 AND BELOW
373-77-4100-01	K	EF	W/D LH FADEC INT. AND NO. 1 DCU	RK-49 AND ABOVE
373-77-4200	E		W/D RH FADEC INT. AND NO. 2 DCU	RK-48 AND BELOW
373-77-4200-01	L	EF.11	W/D RH FADEC INT. AND NO. 2 DCU	RK-49 AND ABOVE
373-34-2515	F	Z	W/D NO.1 PFD TO FADEC INTERFACE	ALL
373-34-2535	F	Z	W/D NO.1 MFD TO FADEC INTERFACE	ALL

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

Table 8 (Continued)

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-34-2555	F	Z	W/D NO.2 PFD TO FADEC INTERFACE	ALL
373-42-0001	A		CSU-4100 STRAPPING	ALL
373-34-2545	F	Z	W/D NO.2 MFD TO FADEC INTERFACE	ALL

As an option the remaining existing avionics wiring replacement may be done IAW the drawings listed below in Table 9.

Table 9

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-22-1101	D		W/D NO.1 MSP	ALL
373-22-1102	D		W/D NO.2 MSP	ALL
373-22-1200	E		W/D APP	RK-048 AND BELOW
373-22-1200-01	D	EF.01	W/D APP	RK-049 AND ABOVE
373-22-1400	D	EF	W/D SPEED SWITCH	ALL
373-23-1001	F	EF	W/D NO.1 VHF	ALL
373-23-1002	E	EF	W/D NO.2 VHF	ALL
373-23-5001	I	EF.08	W/D AUD	RK-48 AND BELOW
373-23-5001-01	H	EF.08	W/D AUD	RK-49 AND ABOVE
373-23-6000	D	Z	W/D ELT	ALL
373-25-1301	D		W/D NO 1 CLOCK	RK-48 AND BELOW
373-25-1301-01	B		W/D NO 1 CLOCK	RK-49 AND ABOVE
373-25-1302	E	Z	W/D NO 2 CLOCK	RK-48 AND BELOW
373-25-1302-01	C	Z	W/D NO 2 CLOCK	RK-49 AND ABOVE
373-34-1111	G	EF	W/D NO.1 ADC	RK-48 AND BELOW
373-34-1111-01	E	EF	W/D NO.1 ADC	RK-49 AND ABOVE
373-34-1112	E	EF	W/D NO.2 ADC	RK-48 AND BELOW
373-34-1112-01	E	EF	W/D NO.2 ADC	RK-49 AND ABOVE
373-34-3001	E		W/D NO.1 VIR	ALL
373-34-3002	E		W/D NO.2 VIR	ALL
373-34-4001	F	EF	W/D RADIO ALT	ALL
373-34-4400	G	EF	W/D TAWS 8000	RK-48 AND BELOW
373-34-4400-01	G	EF	W/D TAWS 8000	RK-49 AND ABOVE
373-34-4501	D		W/D NO.1 DME	ALL
373-34-4502	D		W/D NO.2 DME	ALL
373-34-5001	H	EF	W/D NO.1 GPS	RK-48 AND BELOW
373-34-5002	H	EF	W/D NO.2 GPS	ALL
373-34-5021	E	EF.01	W/D NO.1 ADF	ALL
373-34-5022	E	EF.06	W/D NO.2 ADF	ALL
373-39-1201	E	Y	W/D NO.1 RTU	RK-48 AND BELOW
373-46-2003	E	EF	W/D DATA BASE	ALL

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

Table 9 (Continued)

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-39-1201-01	E	EF	W/D NO.1 RTU	RK-49 AND ABOVE
373-39-1202	D	Y	W/D NO.2 RTU	RK-48 AND BELOW
373-39-1202-01	E	EF	W/D NO.2 RTU	RK-49 AND ABOVE
373-39-4001	D		W/D DAU	P&W JT15D aircraft only.
373-39-4001-01	A		W/D DAU	RK-48 AND BELOW P&W JT15D aircraft only.
373-42-1001	F	EF	W/D IAPS LP1	ALL
373-42-1002	G	EF.13	W/D IAPS LP2	RK-48 AND BELOW
373-42-1002-01	G	EF.13	W/D IAPS LP2	RK-49 AND ABOVE
373-42-1003	D		W/D IAPS LP3	ALL
373-42-1004	F		W/D IAPS LP4	ALL
373-42-1006	E		W/D IAPS RP1	ALL
373-42-1007	G	EF.13	W/D IAPS RP2	RK-48 AND BELOW
373-42-1007-01	D	EF.13	W/D IAPS RP2	RK-49 AND ABOVE
373-42-1008	D		W/D IAPS RP3	RK-48 AND BELOW
373-42-1008-01	A		W/D IAPS RP3	RK-49 AND ABOVE
373-42-1009	F		W/D IAPS RP4	ALL
373-34-6001	G	EF	W/D NO.1 CDU	ALL
373-34-6002	H	EF	W/D NO.2 CDU	ALL

For installation Universal Avionics TAWS A in conjunction with this STC the following drawings in Table 10 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 10

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-34-4650	D	EF	W/D Universal TAWS	ALL
373-91-3446	IR		TAWS Wire Routing Diagram	ALL
Structural Drawings				
373-91-6000	A		TAWS Installation	

For FMS-6100 WAAS/LPV Approach capability in conjunction with this STC the following drawings in Table 11 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 11

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-34-5101	B	Z	No. 1 FMS-6100 WAAS LPV Upgrade	ALL
373-34-5102	B	Z	No. 2 FMS-6100 WAAS LPV Upgrade	ALL

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

For installation new Glareshield Annunciators in conjunction with this STC the following drawings in Table 12 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 12				
Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-33-1200	B	CC	Glareshield Annunciators Wiring Diagram	ALL
Structural Drawings				
NX110-10004	A	EF.09	Glareshield Reg. No Placard	ALL
NX110-10005	G	EF.10	Engine Fire Placard	ALL
NX316-10001	C	EF.09	Assy, Glareshield, New Design (ND)	ALL
NX316-10001-01	C	EF.09	Glareshield, (ND)	ALL
NX316-10001-02	B	Z	Glareshield Main Support	ALL
NX316-10001-03	IR	Z	Strap, Glareshield, Upper	ALL
NX316-10001-04	IR	Z	Strap, Glareshield, (ND), Lower	ALL
NX316-10001-05	IR	Z	Shim, New Glareshield, Top Side Arm, LH	ALL
NX316-10001-06	IR	Z	Shim, Glareshield, ND, Lower	ALL
NX316-10001-07	IR	Z	Shim, New Glareshield, Top Side Arm, RH	ALL
NX315-10005	IR	EF	Placard, Eng Fire Bell Switch	ALL

For installation Midcontinent Instruments MD-302 in conjunction with this STC the following drawings in Table 13 should be used in addition to the drawings included in the base installation and any other optional installations:

Table 13				
Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-24-2005	C	DD	MD302 Standby Instrument Wiring Diagram	ALL
Structural Drawings				
NX312-1211009	A	DD	MD302 Installation with Pitot/Static Lines Retrofit	ALL

Master Drawing List – Upgrade of Collins Proline 4 to 21 Avionics System in the Hawker Beech 400A Aircraft

Supporting Data

Drawings	Rev	Issue	Description	Aircraft applicability
Wiring Diagrams				
373-28-6000	A		W/D Fuel Quantity Cal Box	ALL
373-28-6002	IR		W/D Fuel Quantity Test Box	ALL
815-2680-011	A		Software Accomplishment Summary AFD-3010(E)	FJ44-3AP Equipped Aircraft Only
815-2684-024	A		Software Configuration Index AFD-3010(E)	FJ44-3AP Equipped Aircraft Only
832-5125-511	-		Software Accomplishment Summary IOC-4100	FJ44-3AP Equipped Aircraft Only
818-6437-511	-		Software Configuration Index IOC-4100	FJ44-3AP Equipped Aircraft Only
963-4655-003	-		Software Accomplishment Summary DCU-3001C	FJ44-3AP Equipped Aircraft Only
963-4657-003	-		Software Configuration Index DCU-3001C	FJ44-3AP Equipped Aircraft Only
PT-373-00-0004	C	EF.03	PRODUCTION, GROUND TEST PROCEDURE (PROLINE 21 UPGRADE)	ALL
PT-373-00-0024	B	EF	PRODUCTION, GROUND TEST PROCEDURE (EIS)	ALL

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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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EFTA01261745

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION				1. DATE
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				March 4, 2015
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Constant Aviation	
LIST OF DATA				
6. IDENTIFICATION	7. TITLE			
CA-EO-0315-02 Rev. IR March 4, 2015	LH FWD Circuit Breaker Panel Fabrication.			
NX311-10010 Rev. IR February 20, 2015	Install, MD-302 STBY Instrument Upgrade.			
	-----END-----			
	Notes:			
	1) The structural aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements." 2) This form does not constitute FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration. The electrical requirements are not included in this approval and require separate approval.			
8. PURPOSE OF DATA: Engineering data in support of major alteration to Beechcraft 400A, S/N RK-260.				
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.562(c)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], 25.625(a)(b)(c)[25-23], and 25.789[25-32].				
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.				
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)		
 Paul Hedding	DETR-410115-CE	Structures		

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SDNY_GM_02758165

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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EFTA01261747



ENGINEERING ORDER

CA-EO-0315-02
REV_IR

Model: Nextant 400XT	P/N: N/A
Title: LH FWD Circuit Breaker Panel Fabrication	Drawing(s): N995801
Tail-Serial Number(s): N727KG – RK-260	

DESCRIPTION

<p>The purpose of this Engineering Order is to add proper labeling for the circuit breaker panel on the Avionics Bus. Current circuit breaker panel is not properly labeled.</p> <ol style="list-style-type: none"> 1. Remove Circuit Breaker plug at location shown on page 2. 2. Create Blue circle around location called out on page 2. 3. Label Circuit Breaker Panel with ADVISORY PNL above the location. Follow drawing N995801 for text sizing and color.

APPROVALS

CREATED BY: <u>A. Howard</u>	DATE: <u>03/04/2015</u>	APPROVED BY: <u>D. Buzz</u>	DATE: <u>03/04/2015</u>
CHECKED BY: <u>C. Blackwell</u>	DATE: <u>03/04/2015</u>	FAA APPROVAL REQUIRED? YES <input checked="" type="checkbox"/> NO <input type="checkbox"/>	Change: MAJOR <input checked="" type="checkbox"/> MINOR <input type="checkbox"/>

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EFTA_00245091

EFTA01261751

Nextant Aerospace LLC
355 Richmond Rd.
Richmond Heights OH 44143

Report No.: 373-00-0055
Revision B

Instructions For Continued Airworthiness

Midcontinent Instruments MD-302 Standby Attitude Module (SAM)

Hawker/Beechcraft 400A Aircraft

These instructions must be attached to the approved Airplane Maintenance Manual when a Midcontinent Instruments MD-302 Standby Attitude Module is installed in accordance with STC ST10959SC.

The information contained herein supplements or supersedes the basic Airplane Maintenance Manual only in those areas listed herein. For limitations, procedures, and performance information not contained in this supplement, consult the basic Airplane Maintenance Manual.

Accepted By Nextant Aerospace Date 11-18-2013

1 of 20

SDNY_GM_02758170

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245092

EFTA01261752

LOG OF REVISIONS

REV	PAGES	DESCRIPTION	ACCEPTED BY	DATE
IR	All	Initial Release	Nextant Aerospace	3-25-13
A	4-5	Deleted MD302 Internal Battery	Nextant Aerospace	7-1-13
B	All	Removed additional reference to MD-302 Internal battery. Removed remote baro-set function and Mach readout test as they are no longer available.	Nextant Aerospace	11-18-13

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1.0 Introduction

This document is applicable to the Beech 400A Airplanes, modified by the installation of the Midcontinent Instruments MD-302 Standby Attitude Module (SAM) in accordance with this STC. This document complies with the requirements of 14 CFR Part 25.1529, ICA.

These Maintenance Requirements and Instructions for Continuous Airworthiness (ICA) pertain to the installation of MD-302 Standby Attitude Module (SAM). Modification of an aircraft by this Supplemental Type Certificate obligates the aircraft operator to include the maintenance information provided by this document in the operator's Aircraft Maintenance Manual and the operator's Aircraft Scheduled Maintenance Program.

2.0 System Description

The model MD302 series Standby Attitude Module is a self-contained situational awareness instrument that provides aircraft attitude, altitude, airspeed and slip indication. The MD-302 will replace the three existing electro-mechanical 2" Standby Attitude, Altimeter, and Airspeed.

The MD302 design is built around a solid-state electronic sensor array for high reliability. The dual, high-resolution LCD display uses smooth graphics, daylight-readable brightness, and a configurable lighting response curve to ensure optimal visibility in all conditions.

The MD302 will be supplied emergency backup power by a dedicated existing Standby Emergency Battery. With this backup the MD-302 system meets the operational requirements for aircraft with all electronic instruments.

SYSTEM COMPONENTS

The system consists of the following components:

MD-302 Standby Attitude Module (SAM)
Existing Standby Battery ARM/ON Annunciator Switch

MD-302 IS protected by the following circuit breaker:

C/B Name	Amps	Location	BUS
STBY ATT IND	2	LH FWD CB Panel	Standby Bat Bus

3.0 Control/Special Procedures

Normal operating procedures are identified in the MD-302 Standby Attitude Module (SAM) Installation and Operating Manual, publication number 9017782 Section 3.

4.0 Servicing –

There is no required servicing of systems installed under this STC.

5.0 Maintenance

Maintenance should be referred to person or facility certificated to perform the work being accomplished.

The following manuals contain complete detailed maintenance instructions and should be consulted for all maintenance activities not covered within this ICA:

MD-302 Standby Attitude Module (SAM) Installation and Operating Manual, publication number 9017782

A. ~~Scheduled Inspections and Maintenance~~

Wire/Cable/Racks/Connectors

Inspect system installation components including wires bundles, cables, racks, and connector condition and security in conjunction with "B" Check (Every 400 hours after initial 200 hour "I" inspection) tasks and "C" check (Every 1200 hours after initial 200 hour "I" inspection) per Hawker Beechjet B400 Maintenance Manual 5.20.02 requirements as listed within Scheduled Inspections Section RK-1 and above.

PRESSURE SYSTEM AND ALTIMETER VERIFICATION

Per federal regulation 14 CFR 91.411, it is required that each static pressure system and each altimeter have been tested and inspected within the last twenty-four (24) calendar months.

MD-302 Configuration

The configuration module contains the system configuration as certified under this STC. If the configuration module fails or the configuration is lost the new configuration module must be reconfigured per Nextant Aerospace Ground Test located in Appendix B.

6.0 Troubleshooting

Maintenance and troubleshooting should be referred to person or facility certificated to perform the work being accomplished.

When troubleshooting the system, reference:

Appendix A, Electrical Drawing List.

MD-302 Standby Attitude Module (SAM) Installation and Operating Manual, publication number 9017782, Section 5.

The following Figures and associated descriptions represent warnings or errors and describe the typical reasons and appropriate response action.

FIGURE 6.1

The attitude display has failed due to exceedance of internal rate sensors, loss of airspeed or other various reasons. This will typically self-correct. However, if it does not, immediately have the unit serviced.



FIGURE 6.2

The altimeter instrument has failed, possibly due to exceedance of the pressure sensor range. This may self-correct, but if the error persists, immediately have the unit serviced.



FIGURE 6.3

The airspeed instrument has failed, possibly due to exceedance of the pressure sensor range. This may self-correct, but if the error persists, immediately have the unit serviced.



FIGURE 6.4

The airspeed and altimeter display have both failed due to exceedance of their respective sensor ranges. This may self-correct, but if the error persists, service unit immediately.



FIGURE 6.5

This message is displayed when power is removed at low airspeed (typical ground condition). The pilot has 60 seconds to confirm to remain on, otherwise the unit will automatically power off to conserve battery.



FIGURE 6.6

The configuration module has failed or is not installed properly. Internal settings will be used. If the configuration module is installed, it should be serviced immediately.



FIGURE 6.7

Internal failure. Service unit immediately.



FIGURE 6.8
Internal failure. Service unit immediately.



FIGURE 6.9
Internal (unit) and external settings (configuration module) failure. It is unlikely that both settings would be lost. Try to reconfigure the unit. Service unit if this error persists.



FIGURE 6.10
Internal (unit) settings were found to be invalid. The configuration module settings will be copied to internal memory. Service unit if this error persists.



FIGURE 6.11
External settings (configuration module) were found to be invalid. The internal settings will be copied to the configuration module memory. Service unit if this error persists.



FIGURE 6.12
Internal (unit) and external (configuration module) settings were found to be different. This is typically the result of replacing a unit in an existing installation. The external settings will be copied to replace the settings in internal memory.

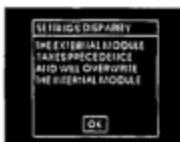


FIGURE 6.13

In configuration mode, this may be the result of lost settings both internally and externally. Try to configure the unit. Seek service if error persists.

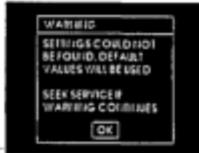


FIGURE 6.14

In configuration mode, this may be the result of lost settings and a failure to initialize memory. Try to restart and configure the unit. Seek service if error persists.

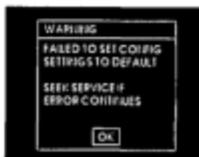


FIGURE 6.15

Internal (unit) and external (configuration module) settings were found to be different. This is typically the result of replacing a unit in an existing installation. In configuration mode, this choice allows selection of which settings to use.



All other discrete inputs and suspected wiring faults should be isolated using basic troubleshooting procedures and the wiring diagrams identified in Appendix A.

7.0 Removal and Replacement

NOTE: Properly protect aircraft electrical connectors if unit is to be removed for extended period of time.

MD-302 (Center Instrument Panel)

Removal of MD-302:

- a. Remove Electrical Power from Aircraft..
- b. Remove 4 mounting screws from MD-302.
- c. Slide out unit from panel and disconnect electrical connector and pitot static lines.

Installation of MD-302.

- a. Secure electrical connector and pitot static lines.
- b. Install MD-302 with four mounting screws.
- c. Apply aircraft power and verify MD-302 powers up and no flags in view. Perform function and pitot static system leak tests per Appendix B Section II.

MD-302 Configuration Module (MD-302 Sub-D Connector)

NOTE: The Configuration Module PC Board Assembly contains sensitive electronics that can be damaged by electrostatic discharge (ESD). Appropriate precautions should be applied prior to handling this component.

Removal of MD-302 Configuration Module:

- a. Remove Electrical Power from Aircraft..
- b. Remove MD-302 as above.
- c. Remove top cover of MD-302 Sub-D connector.
- d. Using appropriate pin removal tool remove the 4 configuration module PC Board pins from the connector and remove module.

Installation of MD-302 Configuration Module: (Refer to Figure 7-1 on next page)

- a. Insert the pins of the Configuration Module PC Board Assembly (Item 3) into their corresponding locations as noted below using an appropriate pin insertion tool.
- b. The wires coming from the Configuration Module PC Board Assembly are marked as follows on the circuit board: TP1, TP2, TP3, TP4.
- c. With the D-Sub oriented up (pin locations 1-5 on top), orient the Configuration Module PC Board Assembly with the electronic parts facing UP prior to pin insertion.
- d. Install each pin into the rear of the D-Sub connector as follows:
 - TP1 = config return = pin 15
 - TP2 = config data = pin 14
 - TP3 = config clock = pin 10
 - TP4 = config power = pin 5
- e. Install the D-Sub Backshell Spring (Item 5) as shown.
- f. Place the D-Sub Slide Lock (Item 6) over the D-Sub connector.
- g. Install the D-Sub connector with Slide Lock and cable harness attached into the Backshell (Item 9) and secure with (2) screws (Item 8). Verify that the Backshell Spring is between the Slide Lock and Backshell. Move the Slide Lock back and forth to verify free movement.

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- h. Route the aircraft wire harness bundle (excluding shield braids) between the two halves of the Cable Strain Relief Clamp (Item 7). The Clamp should be placed over the chafe protection installed in Step 2 (if used).
 - i. Loosely connect the two halves of the Cable Strain Relief Clamp with (2) screws (Item 8).
 - j. Place the Cable Strain Relief Clamp in the Backshell as shown.
 - k. Bend the wires of the Configuration Module PC Board Assembly 180 degrees so that the PC Board has its electrical components facing down as shown. Be careful not to place excess strain on the solder connections between the wires and the PC Board.
-
- l. Capture the Configuration Module PC Board Assembly into the Backshell by placing the Backshell Cover (Item 2) on top of the Backshell.
 - m. Secure the Backshell Cover onto the Backshell using (2) Screws (Item 9).
 - n. The completed assembly should look as shown. Verify that the Slide Lock operates freely and that no wires are pinched, nicked, or otherwise damaged.
 - d. Reinstall MD-302 per the above procedure.
 - e. Apply aircraft power and verify MD-302 powers up and no flags in view. Perform Configuration, Function Test, and pitot static system leak test per Appendix B.

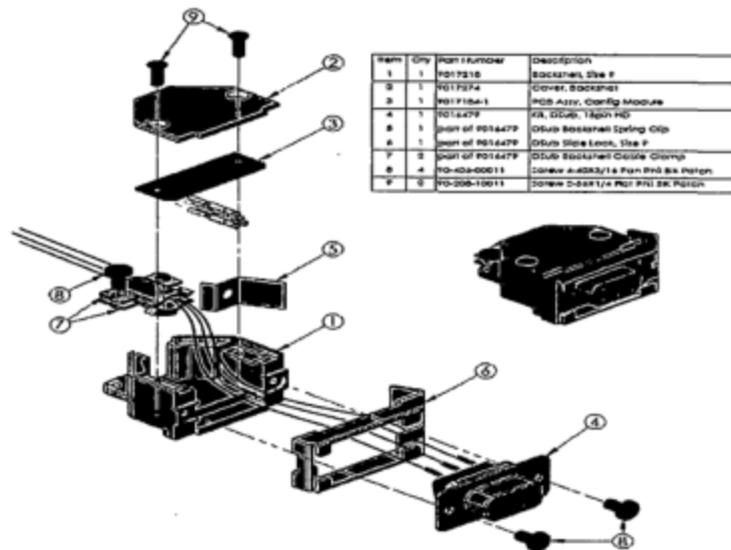


Figure 7-1 Configuration Module Assembly

8.0 Diagrams

The applicable drawings required to perform certain aspects of this ICA are contained in the aircraft drawing package provided to end user and listed in Appendix A.

- 9.0 Special Inspections – None
- 10.0 Protective Treatments – Not Applicable
- 11.0 Fasteners – Not Applicable
- 12.0 Special Tools – None
- 13.0 Commuter – Not Applicable

14.0 Overhaul – Not Applicable

15.0 Airworthiness Limitations

- 15.1 The Airworthiness Limitations section is FAA approved and specifies maintenance required under §§43.16 and 91.403 of the Federal Aviation Regulations unless an alternative program has been FAA approved.

No additional airworthiness limitations are added to the aircraft maintenance procedures with this installation.

16.0 Revisions

Should a revision be required, a letter will be submitted to the Aircraft Certification Office with a copy of the revised ICA. Once a revision is accepted, a copy of the revised ICA will be issued for subsequent installations. Revisions will be sent to operators of previously altered aircraft via US Mail based on STC Authorization Letter address. It is up to the aircraft operator to advise Nextant Aerospace of address changes or change of aircraft ownership.

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End of Report

Nextant Aerospace LLC
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Report No.: 373-00-0055
Revision B

APPENDIX A

APPLICABLE WIRING DIAGRAMS

Drawing Number	Title
373-24-2005	MD302 Standby Instrument Wiring Diagram

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APPENDIX B

Ground Test Procedure

Midcontinent Instruments MD-302 Standby Attitude Module

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I. SYSTEM CONFIGURATION

Configure the MD-302 in accordance with MD-302 Standby Attitude Module (SAM) Installation and Operating Manual, publication number 9017782, Section 4. System configuration shall be as follows:

DISPLAY CONFIGURATION	
Roll Display	Fixed Scale
Airspeed Units	Knots
Display Orientation	Vertical Left
DIMMING CONFIGURATION	
Dimming Control	EXT 5V
Dimming Curve	See Section 4.3.2 of above manual. Set as required to match existing flight displays.
AIRCRAFT CONFIGURATION	
Panel Tilt	11 degrees
Range MMO	.78M
Range Markings	None
ARINC Speed	High Speed

II. GROUND TEST

Test Equipment Required

Pitot Static test equipment Cal Date _____

Initial Power-On

- a. Assure that all avionics systems circuit breakers are set.
- b. Provide ground power to the aircraft and power all aircraft busses.
- c. Assure that all avionics are activated and exhibit no failures or abnormalities.
- d. Verify flags pull from view on MD-302 after initialization is completed (Initialization time may take several minutes).

PASS/FAIL _____

Lighting And Accessibility

- a. While varying the center instrument panel lighting controls, verify lighting of the MD-302 is functioning.

PASS/FAIL _____

MD-302 Emergency Battery Power Test

1. With normal 28 vdc electrical power to the airplane select the Standby Battery switch to ARM. Select cockpit annunciators BRT/DIM switch to BRT. Verify that STBY PWR and ARM annunciators illuminate. Confirm that the MD-302 is powered and with no flags in view.
2. Select aircraft BATTERY switch to OFF. Observe that emergency electrical power activates and that STBY PWR and ON annunciators illuminate.

Verify the MD-302 is powered with no flags in view.

3. Select Standby Power Arm Switch to the OFF position (button released). Switch STBY PWR, ARM, and ON annunciators should be extinguished.
4. Apply normal 28 VDC electrical power to airplane, select the Standby Battery switch to ARM, and verify MD-302 is powered with no flags in view..

Pass ___ Fail ___

MD-302 Static Test

1. Compare the MD-302 display with that of primary flight displays.
 - a. Attitude will be $\pm 0.5^\circ$.
 - b. Airspeed indication will be minimum setting as configured.
 - c. Altitude will indicate field elevation within ± 20 feet. (Ensure proper baro setting)

Pass___ Fail___

MD-302 Air Data Tests

System Leak Test

1. Connect Pitot/Static test set to both Pitot/Static Systems.
2. Evacuate the static pressure system until a pressure differential equivalent to the maximum cabin pressure differential for which the airplane is type certificated is achieved (14,000 feet). Without additional pumping for a period of 1 minute, the loss of indicated altitude must not exceed 2 percent of the equivalent altitude of the maximum cabin differential pressure (280 feet).

Pass ___ Fail ___

NOTE: The following accuracy tests are only required as part of the 24 month 14CRF 91.411 certification.

Altitude Accuracy

Connect Pitot/Static test set to both Pitot/Static Systems.

Using following table verify accuracy of MD-302 Altimeter display.

Note: The MD-302 does not contain a static error correction curve. At certain altitudes significant altitude differences from the primary aircraft displays and the MD-302 may occur.

NOTE: All Tolerances are + or -. Measurements in feet.

Altitude	Tester	MD-302	Tolerance
-1000			20
0			20
500			20
1000			20
1500			25
2000			30
3000			30
4000			35
6000			40
8000			60
10000			80
12000			90
14000			100
16000			110
18000			120
20000			130
22000			140
25000			155
30000			180
35000			205
40000			230
45000			255
50000			280

Continued next page

NOTE: Tests on this page and next page need to be conducted concurrently to assure hysteresis is within tolerance.

Altitude Accuracy (Continued)

Altitude	Tester	MD-302	Tolerance
45000			280
40000			230
35000			205
30000			180
25000			155
22000			140
20000			130
18000			120
16000			110
14000			100
12000			90
10000			80
8000			60
6000			40
4000			35
3000			30
2000			30
1500			25
1000			20
500			20
0			20
-1000			20

NOTE: Assure all altitude readings are within tolerance.

PASS _____ Fail _____

Airspeed/Mach Accuracy

Set altitude at 5,000 ft.

Set Speed as shown in chart and record data for items listed.

SPEED	Tester	MD-302
50kts		
75kts		
100kts		
125kts		
150kts		
175kts		
200kts		
225kts		
250kts		
275kts		
300kts		
325kts		
350kts		

NOTE: Speeds must be within +/- 4 kts of test value

Set altitude at 30,000 ft.

END OF REPORT

Nextant Aerospace LLC
355 Richmond Rd.
Richmond Heights OH 44143

Document No.: 373-00-0048
Revision IR
Hawker Beechcraft 400A

**FAA APPROVED
AIRPLANE FLIGHT MANUAL SUPPLEMENT**

**HAWKER BEECHCRAFT
400A**

Midcontinent Instruments MD-302 Standby Attitude Module (SAM)

This Supplement must be attached to the Approved Airplane Flight Manual when the Midcontinent Instruments MD-302 Standby Attitude Module (SAM) is installed in accordance with STC ST10959SC and Nextant Aerospace Master Document List 373-00-0001.

The information contained herein supplements the information of the basic Airplane Flight Manual. For Limitations, Procedures and Performance information not contained in this Supplement, consult the basic Airplane Flight Manual.

FAA APPROVED

E. Michael Ward

for

Steven L. Lardinois, Manager
Systems and Flight Test Branch, ACE-117C
Chicago Aircraft Certification Office
2300 E. Devon Ave.
Des Plaines, IL 60018
FAA Central Region

Date: DEC 16 2013

Title Page

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Nextant Aerospace LLC
355 Richmond Rd.
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Document No.: 373-00-0048
Revision IR
Hawker Beechcraft 400A

LOG OF REVISIONS

REV	PAGES	DESCRIPTIONS	FAA APPROVED
IR	ALL	Initial Release	 ACE-117C Steven L. Lardinois, Manager Systems and Flight Test Branch, ACE-117C Chicago Aircraft Certification Office 2300 E. Devon Ave. Des Plaines, IL 60018 FAA Central Region Date: DEC 16 2013

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SECTION 1 - GENERAL

SYSTEM DESCRIPTION.

The model MD302 series Standby Attitude Module is a self-contained situational awareness instrument that provides aircraft attitude, altitude, airspeed and slip indication.

The MD302 design is built around a solid-state electronic sensor array for high reliability. The dual, high-resolution LCD display uses smooth graphics, daylight-readable brightness, and a configurable lighting response curve to ensure optimal visibility in all conditions.

The MD302 will be supplied emergency backup power by a dedicated existing Standby Emergency Battery. With this backup the MD-302 system will meet the operational requirements for aircraft with all electronic instruments.

SYSTEM COMPONENTS

The system consists of the following components:
MD-302 Standby Attitude Module (SAM)
Existing Standby Battery ARM/ON Annunciator Switch

MD-302 IS protected by the following circuit breaker:

C/B Name	Amps	Location	BUS
STBY ATT IND	2	LH AFT CB Panel	Standby Bat Bus

SECTION 2 – OPERATING LIMITATIONS

NO CHANGE TO AFM

SECTION 3 - EMERGENCY PROCEDURES

The MD302 is designed to operate reliably and provide the critical situational awareness needed even if the aircraft power systems fail. When this occurs, the unit provides emergency operation by continuing to perform seamlessly and uninterrupted in Flight Mode.

When primary aircraft power to the unit is lost in flight, the unit will immediately begin operating on standby battery power. This is indicated by the STANDBY POWER ON annunciator illuminated.

FAA Approved Date: **DEC 16 2013**

Page 1

SDNY_GM_02758193

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245115

EFTA01261775

SECTION 3A - ABNORMAL PROCEDURES

The following indication may be shown during system failure:

The attitude display has failed. Use remaining aircraft systems for navigation.



The altimeter instrument has failed. Use remaining aircraft systems for altitude.



The airspeed instrument has failed. Use remaining aircraft systems for airspeed.



SECTION 4 - NORMAL PROCEDURES

Pre-Flight

Select Standby Power On using Standby Power ARM/ON switch.
Perform Standby Power test per Airplane Flight Manual (AFM) Section 4.
After power up verify MD-302 enters Flight Mode as shown below.

Operation

In Pre-flight Mode, power is applied to the unit and the Introduction Screen appears during startup.

During Pre-flight Mode, the Introduction Screen will be displayed while the unit conducts an initial Power-up Built-in Test (PBIT) of the system to validate operational readiness. This includes, among others, an internal test to verify software and memory, and confirmation that the internal settings and identification of the unit match the Configuration Module installed in the aircraft cable harness.

The Introduction Screen will be displayed for approximately five seconds and will transition to Flight Mode when complete.



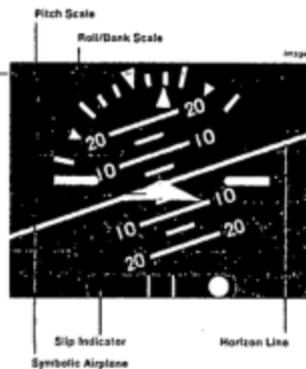
PRE-FLIGHT MODE
INTRODUCTION SCREEN

In Flight Mode, the unit operates normally by displaying four major functions: Attitude, Altitude, Airspeed, and Slip Information



FLIGHT MODE

Attitude



The Attitude Indicator consists of seven parts: Horizon Line, Sky (blue), Ground (brown), Symbolic Airplane, Roll Scale, Pitch Scale and Slip Indicator.

The Roll Scale is depicted as an arc of graduations representing bank angles of 0° (triangle), 10°, 20°, 30°, 45° (small triangle) and 60°.

The unit is operable and usable in a continuous and unlimited roll range of 360°+.

The Roll Pointer is the triangle just below the Roll Scale and represents the aircraft in relation to its bank angle. It is configured to operate conversely to the Roll Scale behavior. That is, a fixed Roll Scale produces a rotating Roll Pointer.

The Pitch Scale is depicted as a series of graduations representing pitch angles every 5°, with every 10° graduation being wider and numbered. The unit is operable and usable in a continuous and unlimited pitch range of 360°+. A series of chevrons (^) will appear overlaid on the Pitch Scale at attitudes greater than ± 45°. This is to indicate to the pilot the direction of the horizon for quick reference when in unusual pitch attitude. The chevrons always point toward the horizon line.



ATTITUDE OPERATION

This image demonstrates 45° pitch up and 15° left turn. Red chevrons point to the horizon line.

The Slip Indicator portion of the display will always appear at the bottom of the Attitude Display. The Slip Indicator is represented by a shaded translucent background with two white lines around center and a white ball. When the ball is maintained between the vertical lines during banking maneuvers, the turn is considered "coordinated" without slip. The Slip Indicator's background becomes semi-transparent if the Roll Scale or Roll Pointer pass behind the indicator. All other elements remain visible.

Altitude

The Altimeter consists of four parts: Altitude Window, Altitude Tape, Barometric Setting Window and optional Altitude Trend Bar.

The Altitude Window displays the current, barometric corrected altitude. The digits of the display are in increments of ten and the window is expanded over this portion of the number to display a minimum spread of twenty units. The numbers will 'roll' or scroll to assist in quick reference to the increasing or decreasing nature of the aircraft's altitude. The hundreds, thousands and ten-thousands digits appear to the left of the tens digits with the thousands and ten-thousand digits slightly larger than the others. The Altitude Pointer (triangle) to the right of the window points to the associated position on the Altitude Tape of the current altitude. Altitude units of measure appear below the Altitude Window and can be changed during Flight Mode using the Options Menu. The pilot may select feet or meters. The Altitude Tape is a vertical scale along the right margin of the display. The current altitude is always in the middle of the tape and indicated by the Altitude Pointer on the right side of the Altitude Window. The tape has numeric indications every one-hundred units with minor graduations every fifty units and sub-graduations every twenty-five units. The tape will 'roll' or scroll to assist in quick reference to the increasing or decreasing nature of the aircraft's altitude.



The Barometric Window shows the currently set barometric pressure. It is identified by the abbreviation BARO and is located at the top center of the Airspeed/Altitude Display. Setting the current barometric pressure compensates the altitude for the appropriate environmental conditions.

The barometric setting can be adjusted by turning the Control Knob while in Flight Mode. When adjusting the barometric pressure, the window will increase in size and the digits will turn green. When finished setting the pressure, the window will return to its original size and color. Barometric pressure units can be selected during Flight Mode using the Options Menu.

The Altitude Trend Bar is located along the right margin of the Altitude Display. This feature is optional and can be turned ON or OFF using the Options Menu. The Altitude Trend Bar is magenta-in-color-and-originate-at-the-current-altitude-on-the-Altitude-Tape-always-from-the-middle-of-the-display, directly across from the Altitude Pointer. The height of the Trend Bar, above or below the current altitude, indicates the altitude of the aircraft on the Altitude Tape if the current vertical speed or 'altitude trend' is maintained over a period of six seconds. For example, as seen in Image 7, the current altitude is approximately 2,420 feet. The Trend Bar is at approximately 2,470 feet, indicating that the aircraft's altitude will be 2,470 feet in six seconds if the current vertical speed or climb is maintained constant. The length of the Trend Bar will increase with increased dive or climb rates and approach zero or disappear entirely as the vertical speed reaches zero in level flight.

Airspeed

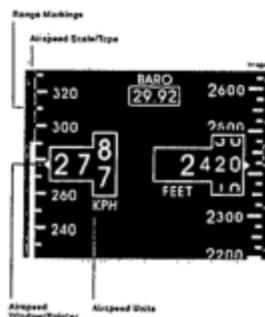
The Airspeed Indicator Display consists of three parts: Airspeed Window, Airspeed Tape and Airspeed Limitations or Range Markings.

The Airspeed Window displays the current Indicated Airspeed (IAS). The digits of the display are enlarged for visibility and increment by one unit. The units will 'roll' or scroll to assist in quick reference as to the increasing or decreasing nature of the aircraft's airspeed.

The Airspeed Pointer (triangle) to the left of the window points to the associated position on the Airspeed Tape of the current airspeed.

The Airspeed Tape is a vertical scale along the left margin of the display. The current airspeed is always in the middle of the tape and indicated by the Airspeed Pointer on the left side of the Airspeed Window. The Airspeed Tape has numeric indications every ten or twenty units depending on the unit of measure selected. Minor graduations appear every five or ten units, respectively.

The Airspeed Tape will 'roll' or scroll to assist in quick reference as to the increasing or decreasing nature of the aircraft's airspeed. A traditional barber pole will be displayed if the aircraft nears Vmo value.



Brightness Control

SAM is configured to adjust its brightness based on the aircraft's manual lighting bus control. This installation uses the center panel dim control in the cockpit overhead.

The pilot or crew member can temporarily override the current brightness and manually increase or decrease brightness.

To adjust brightness, briefly press the Control Knob. The brightness bar will appear overlaid on the Attitude Display and turning the Control Knob will increase or decrease the current setting.

While the unit remains powered, the manual adjustment will remain saved and any change in the lighting bus or photocell sensor will increase or decrease the brightness from the newly set manual adjustment point.

When the unit is powered OFF, the manual adjustment will be reset and default to the lighting response curve programmed into memory per the settings in the Configuration Mode setup by the installer.

SECTION 5 - PERFORMANCE

NO CHANGE TO AFM

SECTION 6 – WEIGHT AND BALANCE/EQUIPMENT

NO CHANGE TO AFM

SECTION 7 – SUPPLEMENTS

NO CHANGE TO AFM

UNITED STATES OF AMERICA
 DEPARTMENT OF TRANSPORTATION-FEDERAL AVIATION ADMINISTRATION
STANDARD AIRWORTHINESS CERTIFICATE

1 NATIONALITY AND REGISTRATION MARKS N727KG	2 MANUFACTURER AND MODEL RAYTHEON AIRCRAFT CO BE-400A	3 AIRCRAFT SERIAL NUMBER RK-260	4 CATEGORY TRANSPORT
--	---	--	--------------------------------

5 AUTHORITY AND BASIS FOR ISSUANCE

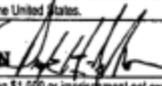
This airworthiness certificate is issued pursuant to 49 U.S.C. § 44704 and certifies that, as of the date of issuance, the aircraft to which issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein.

Exceptions:

NONE

6 TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the FAA, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43, and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United States.

DATE OF ISSUANCE R 12/11/1999	FAA REPRESENTATIVE DEREK H. JACKSON 	DESIGNATION NUMBER AEA-FSDO-68
---	---	--

Any alteration, reproduction, or misuse of this certificate may be punishable by a fine not exceeding \$1,000 or imprisonment not exceeding 3 years or both.
 THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

FAA Form 8100-2 (04-11) Supersedes Previous Edition

SDNY_GM_02758200

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245122

EFTA01261782

SDNY_GM_02758201

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245123

EFTA01261783

UNITED STATES OF AMERICA
 DEPARTMENT OF TRANSPORTATION - FEDERAL AVIATION ADMINISTRATION
STANDARD AIRWORTHINESS CERTIFICATE

1. NATIONALITY AND REGISTRATION MARKS N787TA	2. MANUFACTURER AND MODEL Raytheon Aircraft Co. BE-400A	3. AIRCRAFT SERIAL NUMBER RK-260	4. CATEGORY Transport
--	---	--	---------------------------------

5. AUTHORITY AND BASIS OF ISSUANCE
 This airworthiness certificate is issued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which it is issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein.
 Exceptions:
None

6. TERMS AND CONDITIONS
 Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43, and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United States.

DATE OF ISSUANCE R 04/12/2002	FAA REPRESENTATIVE Donald F. Miller	DESIGNATION NUMBER FAA-PSSD-23
---	---	--

Any alteration, modification, or repair of this aircraft is the responsibility of the holder except to the extent of any comment not exceeding 3 years. On this certificate MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

SDNY_GM_02758202
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758203

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245125

EFTA01261785

REPLACEMENT AIRWORTHINESS CERTIFICATE REQUEST
FAR ORDER 8130.2G, Paragraph 215

I, WILLIAM G. DODRILL, request a replacement
Full Name, please print

Airworthiness Certificate for Aircraft N727KG, RK-260
Registration (N) Number Serial Number

RAYTHEON Aircraft 400A
Make Model

The original certificate is:

Lost

Mutilated

Illegible

"N" number change only

W G Dodrill
Signature

12-29-2017
Date

SDNY_GM_02758204

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245126

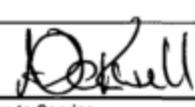
EFTA01261786

SDNY_GM_02758205

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245127

EFTA01261787

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260	
	Make Beechcraft		Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options LLC		Address (As shown on registration certificate) Address: 26180 Curtiss Wright Pkwy City: Richmond Hts State: Oh Zip: 44143 Country: USA	
	3. For FAA Use Only			
3. For FAA Use Only				
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input checked="" type="checkbox"/>	<input type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name: Nextant Aerospace Address: 355 Richmond Rd City: Cleveland State: OH Zip: 44143 Country: USA		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		
		<input type="checkbox"/> Manufacturer C. Certificate No. CRS # 25NR667B		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual  10/9/2012		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual  10/9/2012		

FAA Form 337 (10-06)

SDNY_GM_02758206

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245128

EFTA01261788

SDNY_GM_02758207

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245129

EFTA01261789

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

10/9/2012

Nationality and Registration Mark

Date

Carried out Repair to the Left Pylon Bell Frame Crack.

Repaired Clip P/N 45A96063-7 Crack Damage at Flight Station 329.92 in accordance with US Technical Document EO6217 Rev A, Approved by DERT-605367-NM Darrell A. Meyer. The Clip P/N 45A96063-7 is .063" thick and is made of AL Alloy 2024-T3 CLAD.

See Attached FAA Form 8110-3 Signed by Robert M. Halvorson DERT-605221-NM Dated 10/19/12 Nextant Aerospace EO 108-11 Rev E dated 10/19/12 and US Technical Document U-1-1229 Rev K dated 10/17/12 for Structural Substantiation.

Instructions for Continued Airworthiness.

Follow normal maintenance practices for the left and right engine pylon clip repair as called out in the AMM for the Beechjet 400/400A.

Weight and balance change is negligible.

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758208

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245130

EFTA01261790

SDNY_GM_02758209

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245131

EFTA01261791

DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION			1. DATE October 19, 2012
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A (see note 3)	4. TYPE (Airplane, Radio, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace (see note 4)
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
<p>Nextant Aerospace Document: EO No. 108-11 Rev. E Dated 10/19/12</p> <p>US Technical Document: U1-1229 Rev. K Dated 17 Oct 12</p> <p>—end—</p>	<p>Repair - LH/RH Engine Pylon Clip</p> <p>Structural Substantiation #1 Engine Pylon Clip (P/N 45A96063-7) Crack Damage Repair</p> <p>NOTES:</p> <ol style="list-style-type: none"> Detail design and Structural approval only. This approval is for the engineering design data only. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements". This form does not constitute FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire repair. This approval is for the engineering design and substantiation data. It is not an installation approval. Model: Hawker Beechcraft 400A as defined per TCDS No. A16SW. Applicant: Nextant Aerospace, 26180 Curtiss Wright Pkwy, Cleveland, OH 44143 U.S. LLC 25NR667B. Applicable to the following aircraft S/N & R/N: RK-137 N400AJ, RK-230 N722NK, RK-123 N410FL, RK-225 N415LX, RK-248 N226WC, RK-289 N440LX, RK-093 N493CW, RK-310 N451LX, RK-252 N790TA, RK-195 N718TA, RK-239 N485FL, RK-284 N439LX, RK-334 N442FL, RK-368 N427FL, RK-327 N454LX, RK-146 N746TA, RK-260 N787TA, RK-413 N482LX, RK-397 N473FL, RK-365 N459LX, RK-387 N478LX, RK-189 N715TA, RK-264 N428LX, RK-244 N492LX, RK-180 N709TA. <p>End—end—end—end</p>		
8. PURPOSE OF DATA To show compliance with applicable regulations in support of a Major Repair on Hawker Beechcraft Model 400A as listed in Note 5.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Part 25: 25.301 Amdt. 25-23, 25.303 Amdt. 25-23, 25.305(a)(b) Amdt. 25-86, 25.307(a) Amdt. 25-72, 25.601 Amdt. 25-0, 25.603 Amdt. 25-46, 25.605 Amdt. 25-46, 25.609 Amdt. 25-0, 25.611 Amdt. 25-23, 25.613(a)(b) Amdt. 25-112.			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and attached sheets numbered <u>none</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed.			
I (We) Therefore <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION (S)	
Robert M. Halvorson 	DETR-605221-NM	Structures	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758210

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245132

EFTA01261792

SDNY_GM_02758211

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245133

EFTA01261793

nextant aerospace

ENGINEERING ORDER

EO NO. 108 REV. E

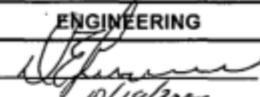
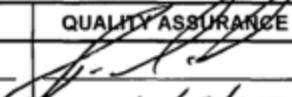
MODEL: BEECH 400A	P/N: 45A96063-7	
TITLE: Repair - LH/RH Engine Pylon Clip	DRAWING NO. n/a	REV: -
Serial Number(s): RK-137, RK-230, RK-123, RK-225, RK-248, RK-289, RK-093, RK-310, RK-252, RK-195, RK-239, RK-284, RK-334, RK-368, RK-327, RK-146, RK-260, RK-413, RK-397, RK-365, RK-387, RK-189, RK-264, RK-244, RK-180		

DESCRIPTION

<p>ORDER DESCRIPTION: Repair, LH/RH Bell Frames: Follow U.S. Technical EO 6217 for repair instructions, document is attached (REF ECR 951)</p> <p>Note: Item 1 in repair bill of materials, section 2.1 of U.S. Technical EO 6217 to be fabricated from .040" thick 2024-T3 CLAD AL PER AMS-QQ-A-250/5 (REF ECR 760)</p>

APPROVALS

CREATED BY: <u>Kevin Himelright</u>	DATE: <u>10/8/2012</u>	APPROVED BY: _____	DATE: _____
CHECKED BY: <u>Kevin Himelright</u>	DATE: <u>10/8/2012</u>	FAA APPROVAL REQUIRED? YES <input checked="" type="checkbox"/> NO <input type="checkbox"/>	Change: MAJOR <input checked="" type="checkbox"/> MINOR <input type="checkbox"/>

ENGINEERING	PRODUCTION	QUALITY ASSURANCE	CUSTOMER / AGENCY
	_____		_____
DATE: <u>10/15/2012</u>	DATE: _____	DATE: <u>2/15/12</u>	DATE: _____

Form No. NXT-426/ Rev IR	Reference Doc: SOP 4.1	Once Printed This Form is Uncontrolled
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SDNY_GM_02758212

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245134

EFTA01261794

SDNY_GM_02758213

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245135

EFTA01261795

nextant aerospace

ENGINEERING CHANGE REQUEST

ECR NO. 951

MODEL: Beech 400A	P/N: 45A96063-7	
TITLE: Repair - LHRH Engine Pylon Clip Revision	DRAWING NO.	REV:

INITIATED BY: Kevin Himmelright	DEPT: Engineering	DATE: 10/8/2012
REASON: Aircraft not in effectivity range need EO repair		

CHANGE DESCRIPTION

REQUESTED CHANGE DESCRIPTION: RK-169, RK-264, RK-244, RK-180	Revise EO 108D TO REV E to include effectivity for S/N RK-260,RK-413, RK-397,RK-365, RK-387, RK-189, RK-264, RK-244, RK-180
--	---

ECR DISPOSITION

ECR: ACCEPT <input checked="" type="checkbox"/>	BY: Kevin Himmelright	REASON FOR REJECTION:
REJECT <input type="checkbox"/>	DATE: 10/8/2012	

APPROVALS

FAA APPROVAL REQUIRED? YES <input checked="" type="checkbox"/> NO <input type="checkbox"/>	Change: MAJOR <input checked="" type="checkbox"/> MINOR <input type="checkbox"/>
--	--

ENGINEERING	PRODUCTION	QUALITY ASSURANCE	SUPPLY CHAIN
DATE: _____	DATE: _____	DATE: _____	DATE: _____

Form No. NXT-425/ Rev A	Reference SOP 4.1	Once Printed this Form is Uncontrolled
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SDNY_GM_02758214

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245136

EFTA01261796

SDNY_GM_02758215

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245137

EFTA01261797



ENGINEERING REPORT

DOC. NO.: E06217

REVISION: C

PAGE: 1 of 10

TITLE:

**Repair – LH/RH Engine Pylon Clip (P/N 45A96063-7)
Crack Damage
Nextant Aerospace – Beechjet 400A**

APPLICATION			BY	DATE
NEXT ASSEMBLY	AIRPLANE	ENGINEER	F. Valerio	20 July 11
N/A	400A	APPROVED	M. Perez	22 July 11

EFFECTIVITY: See Section 1.2

Revisions

REVISION SYMBOL	DESCRIPTION	DATE	APPROVAL
NC	INITIAL RELEASE	22 July 11	Darrell Meyer
A	Updated Figure 3, Section 3.0, and BOM; added Figure 4. Ref UST WO #6231	08 Aug 11	Darrell Meyer
B	Revised EO per customer request for Ground Stud relocation. Added Figure 5.	16 Sep 11	Darrell Meyer
C	Revised EO per customer request to add aircraft RK-295. Ref UST WO #6297	13 Oct 11	Darrell Meyer

List of Active Pages

Rev	C	C	C	C	C	C	C	C	C	C										
PG	1	2	3	4	5	6	7	8	9	10										
Rev																				
PG																				

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Form EA-152

SDNY_GM_02758216

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245138

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SDNY_GM_02758217

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EFTA01261799



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Aviation Products, Services,
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ENGINEERING REPORT

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SDNY_GM_02758218

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 U.S. TECHNICAL Aviation Products, Services, Engineering & Certification	ENGINEERING REPORT	DOC. NO.: E06217
		REVISION: C
		PAGE: 3 of 10
<p>1.0 GENERAL</p> <p>1.1 <u>Scope</u> This document describes the procedure for repair of LH/RH Engine Pylon Clip Crack damage. The Clip P/N 45A96063-7 is .063" thick and is made of Al Alloy 2024-T3 CLAD.</p> <p>1.2 <u>Effectivity</u> Hawker Beechcraft 400A MSN (Registration #) RK-279 (N436FL) and RK-295 (N898TA).</p> <p>1.3 <u>Applicability</u> Damaged area shown in photo pictures in Section 2.</p> <p>1.4 <u>References</u> Beechjet 400A Structural Repair Manual.</p> <p>1.5 <u>Safety Precautions</u> Follow safety practices in Beechjet 400A Structural Repair Manual.</p> <p>1.6 <u>Functional Test Information</u> N/A</p> <p>1.7 <u>Weight and Balance Report</u> N/A</p> <p>1.8 <u>Structural Substantiation Report</u> US Technical Document No. U1-1229.</p> <p>1.9 <u>Electrical Load Data</u> N/A</p>		
<small>Form EA-152</small>		

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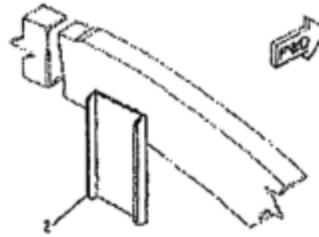
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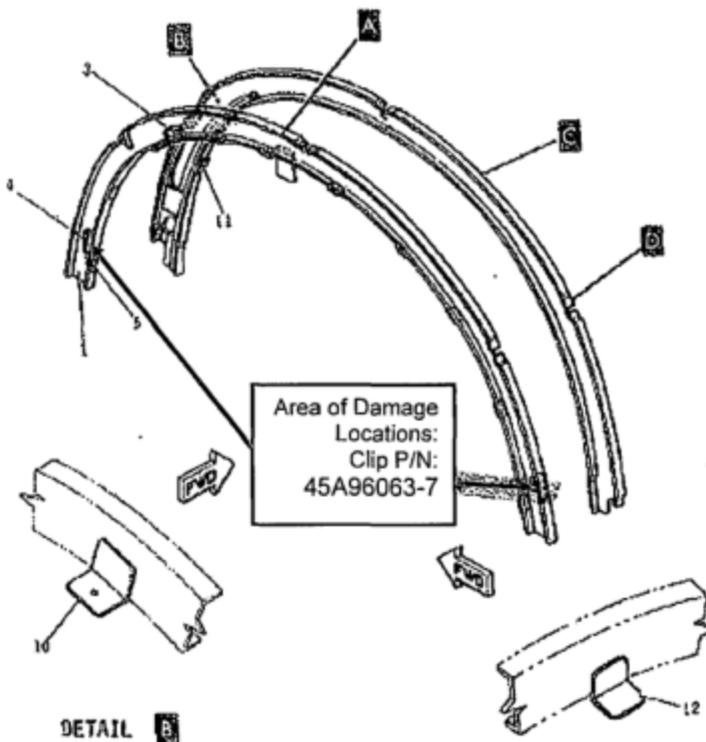
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2.0 DAMAGE & DAMAGE REPAIR



DETAIL A



035338A

Figure 1 – Damage Locations (SRM 53-10-63)

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SDNY_GM_02758223

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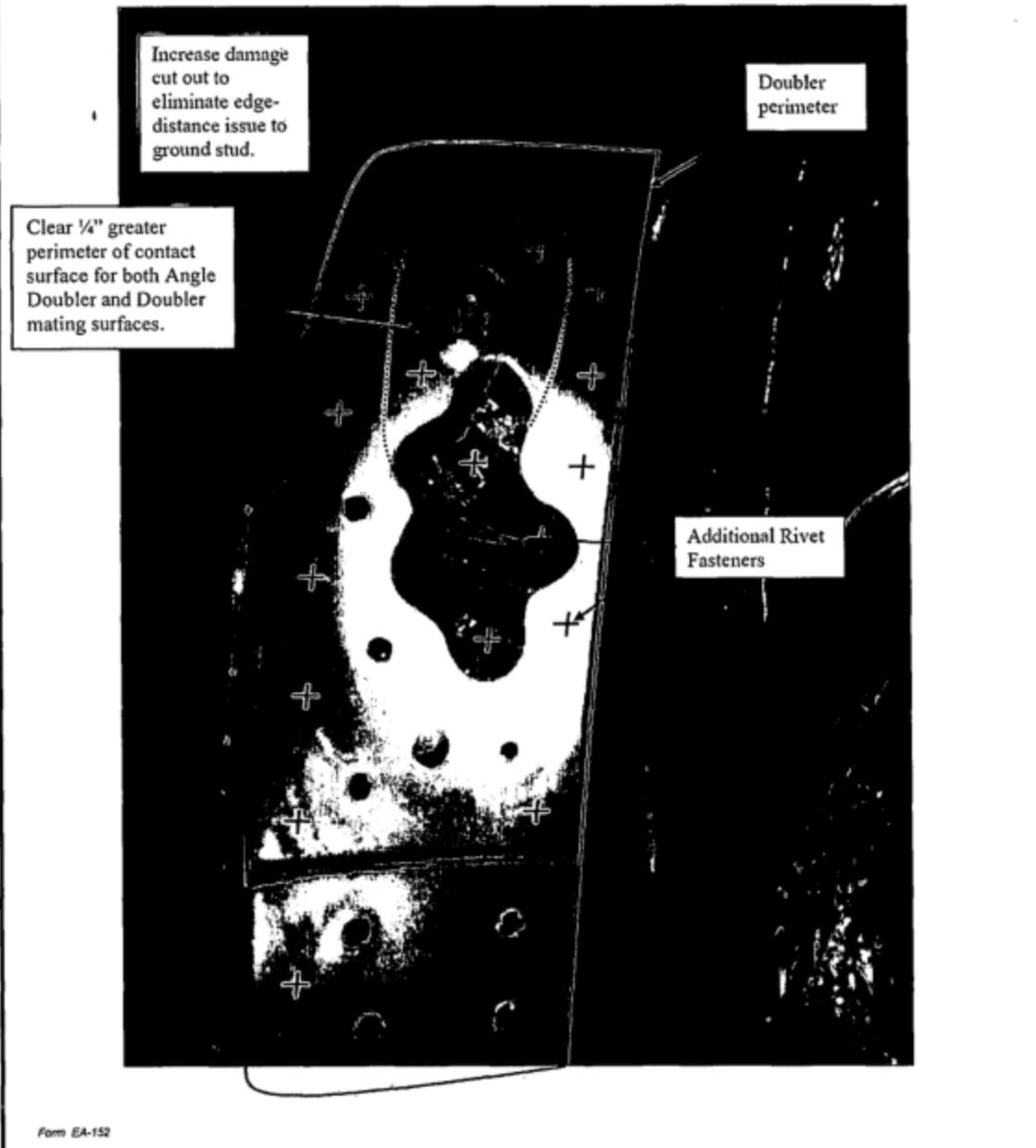
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2.0 DAMAGE & DAMAGE REPAIR – (Continued)

Figure 2 – Photo of LH Clip P/N: 45A96063-7 Crack Damage (RH Opposite)



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2.0 DAMAGE & DAMAGE REPAIR – (Continued)

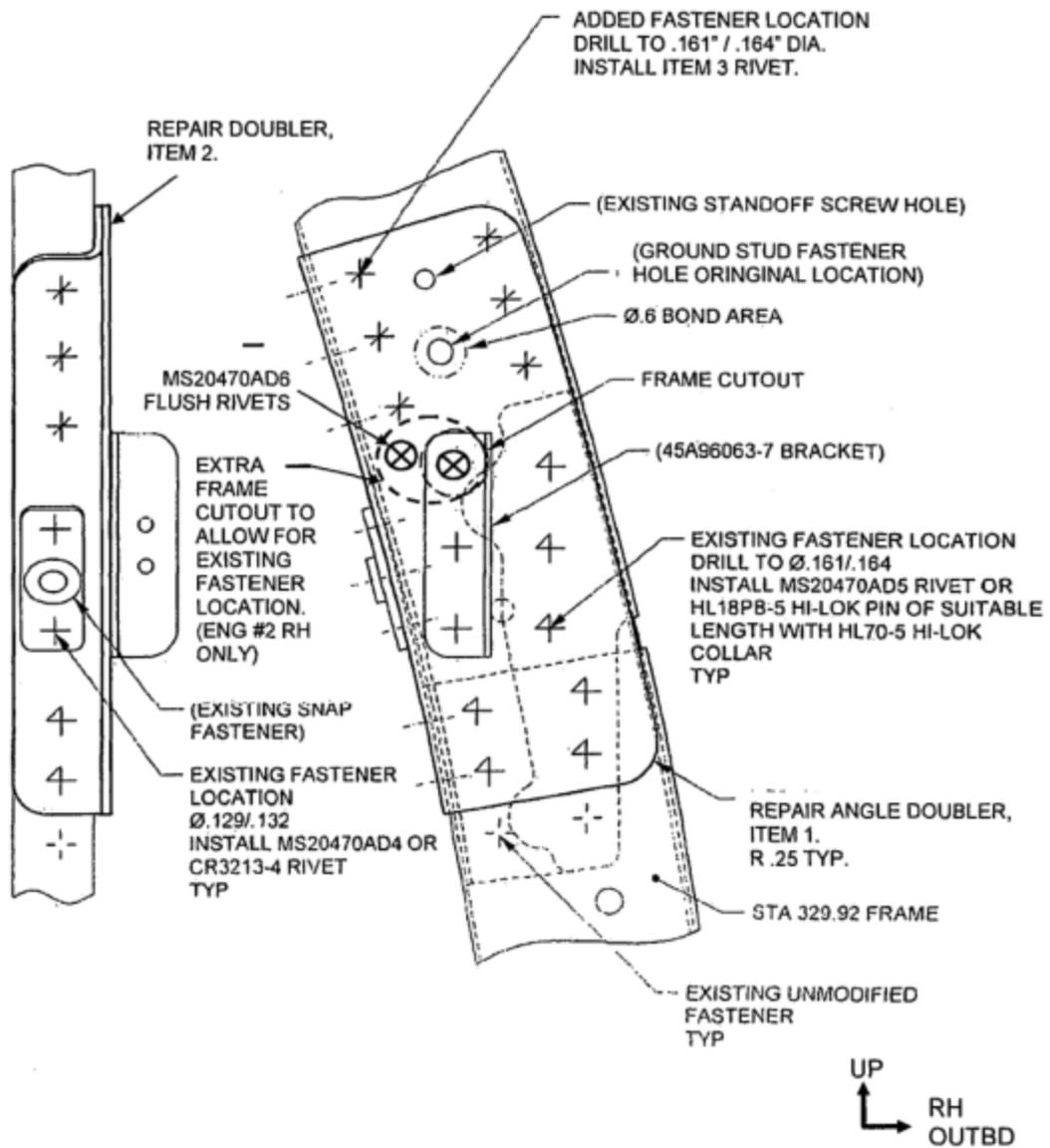


Figure 3 – Sketch of RH Clip Repair (LH Opposite)

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2.0 DAMAGE & DAMAGE REPAIR – (Continued)

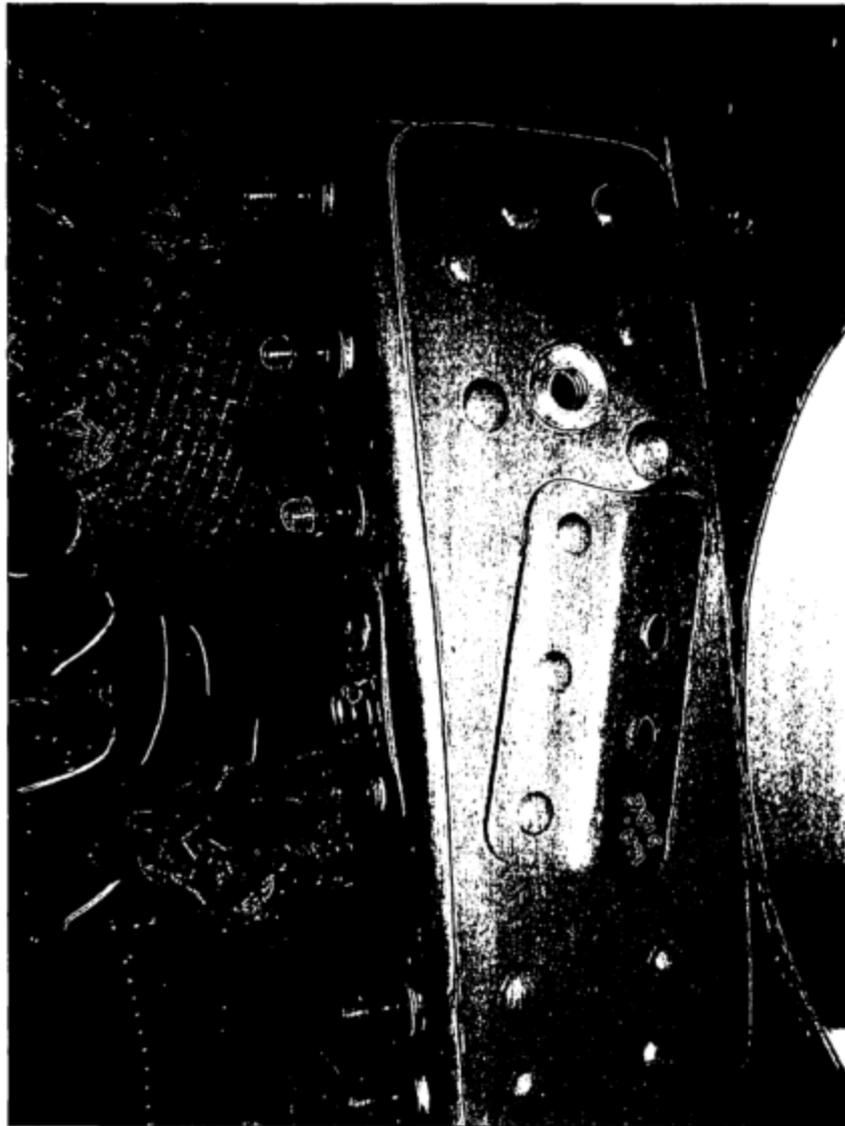


Figure 4 – Photo of RH Clip Final Repair (LH Opposite)

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2.0 DAMAGE & DAMAGE REPAIR – (Continued)



Figure 5 – Photo of Ground Stud Relocation

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2.1 Repair Bill of Materials

Table 1: Repair Parts List

Item No.	Qty	Assy	Description	Specifications	Remarks / Notes
1	2	LH/RH Engine Pylon Clip	Angle Doubler	.040" THICK 2024-T3 CLAD AL PER AMS-QQ-A-250/5	See Figures 1 thru 3
2	2		Doubler	.040" THICK 2024-T3 CLAD AL PER AMS-QQ-A-250/5	
3	AR		Rivet	* MS20470AD5 or CR3213-5-5	
4	AR		Chem Film	MIL-C-5541, CLASS 3	
5	AR		Polyamide Epoxy Primer	MIL-PRE-23377 TYPE 1, CLASS C	
6	AR		Adhesive	HYSOL EA 9309.3NA	
7	2	RH Engine Pylon Clip	Flush Rivet	MS20426AD6	See Figure3

*Note: Permissible to use oversize HL18PB-5 or HL18PB-6-4 and collar Hi-Lok fasteners at limited access locations.

3.0	<u>REPAIR INSTRUCTIONS:</u>	<u>MECH.</u>	<u>INSP.</u>
3.1	CAREFULLY CUT THE DAMAGED AREA FROM THE LH/RH STA 329.92 FRAME AS NEEDED TO REMOVE CRACK DAMAGE. ENSURE ADEQUATE CLEARANCE IS MAINTAINED TO ACCOMMODATE BUTTS OF RIVETS INSTALLED BY THIS FIELD REPAIR (AS SHOWN IN FIGURES 2 & 3). TAKE CARE NOT TO DAMAGE UNDERLYING STRUCTURE.		
3.2	CAREFULLY HAND FORM DAMAGED AREA OF FRAME TO ORIGINAL CONTOUR AS REQUIRED. REFORM 45A96063-7 CLIP AS REQUIRED.		
3.3	CHECK, USING A SUITABLE NDT TECHNIQUE (e.g. FLUORESCENT PENETRANT OR EDDY CURRENT), TO ENSURE NO CRACKS EXIST AND ALL DAMAGE HAS BEEN REMOVED. REPORT ANY ADVERSE FINDINGS TO RAC FOR ASSESSMENT BEFORE CONTINUING WITH THIS REPAIR.		
3.4	FABRICATE REPAIR DOUBLER AND REPAIR ANGLE DOUBLER FROM .040 THICK 2024-T3, RESPECTIVELY, CLAD ALUMINUM SHEET PER AMS-QQ-A-250/5. SIZE AS REQUIRED MAINTAINING A MINIMUM 2D EDGE DISTANCE TO ALL PICKED UP FASTENERS AS SHOWN IN FIGURES 2 & 3, AND FORMED TO NEST OVER INBD FLANGE OF THE STA 329.92 FRAME. BEND RADIUS .06 MIN. HEAT TREAT REPAIR ANGLE DOUBLER TO -T42 CONDITION PER AMS2770 AFTER FORMING COMPLETE.		

Form EA-152

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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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3.0	REPAIR INSTRUCTIONS: (Continued)	<u>MECH.</u>	<u>INSP.</u>
3.5	CAREFULLY REMOVE ALL EXISTING FASTENERS PICKED UP BY THE REPAIR DOUBLER PER THE MODEL 400/400A SRM, 51-40-00.		
3.6	DRILL ALL FASTENER HOLES AS SHOWN IN FIGURES 2 & 3 IN ACCORDANCE WITH THE MODEL 400/400A SRM, 51-41-00 (SOLID RIVETS), 51-43-01 (BLIND RIVETS). MAINTAIN MIN 2D EDGE DISTANCE AT ALL FASTENER LOCATIONS AND 4D-6D SPACING AT ALL ADDED FASTENER LOCATIONS. DRILL REPAIR DOUBLER TO MATCH LOCATION AND SIZE OF EXISTING STANDOFF SCREW HOLES. DRILL GROUND STUD HOLE AT NEW LOCATIONS DEPICTED IN FIGURE 5. DEBURR ALL SHARP EDGES.		
3.7	PREPARE AND BOND THE REPAIR DOUBLER IN PLACE USING HYSOL EA 9309.3NA ADHESIVE IN ACCORDANCE WITH THE ADHESIVE MANUFACTURER'S INSTRUCTIONS. ALLOW ADHESIVE TO FILL GAP CAUSED BY LAP OF INBD FRAME FLANGES.		
3.8	INSTALL ALL FASTENERS AS SHOWN IN FIGURES 2 AND 3 PER THE MODEL 400/400A SRM, 51-40-00 (SOLID RIVETS), 51-43-01 (BLIND RIVETS). NOTE THE UPPER FASTENER SECURING THE 45A96063-7 CLIP WILL BE INSTALLED THROUGH THE REPAIR DOUBLER AND CLIP ONLY.		
3.9	PROTECT ALL BARE ALUMINUM WITH CHEM FILM PER MIL-C-5541, CLASS 3 AND ONE COAT OF MIL-PRE-23377 TYPE 1, CLASS C POLYAMIDE EPOXY PRIMER. ENSURE A 0.6 AREA AROUND THE GROUND STUD HOLE REMAINS FREE OF PRIMER AS SHOWN IN FIGURE 2.		
3.10	REINSTALL ALL OTHER REMOVED HARDWARE WITH EXISTING TYPE AND SIZE FASTENERS OF SUITABLE GRIP LENGTH.		

4.0 INSTRUCTIONS FOR CONTINUED AIRWORTHINESS

FOLLOW NORMAL MAINTENANCE PRACTICES FOR THE LH/RH ENGINE PYLON CLIP REPAIR AS CALLED OUT IN THE AMM FOR BEECHJET 400/400A.

NOTES

1. WEIGHT AND BALANCE CHANGE NEGLIGIBLE.
2. ALL DIMENSIONS ARE IN INCHES UNLESS OTHERWISE SPECIFIED.
3. EMBODIMENT OF THIS FIELD REPAIR MUST BE RECORDED IN THE AIRCRAFT LOG BOOK STATING THIS FIELD REPAIR NUMBER AND REVISION.

Form 6A-152

SDNY_GM_02758234

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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SDNY_GM_02758235

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245157

EFTA01261817

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260	
	Make Raytheon Aircraft Company 400A		Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) 26180 Curtiss Wright PKWY	
			City Richmond Heights	State OH
		Zip 44143-1453	Country USA	
3. For FAA Use Only				
3. For FAA Use Only				
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name Nextant Aerospace		<input type="checkbox"/> U.S. Certified Mechanic		<input type="checkbox"/> Manufacturer
Address 355 Richmond Road		<input type="checkbox"/> Foreign Certified Mechanic		C. Certificate No.
City Cleveland State OH		<input checked="" type="checkbox"/> Certified Repair Station		CRS # 25NR667B
Zip 44143-1453 Country USA		<input type="checkbox"/> Certified Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A. Ockwell 		01/08/2013
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A. Ockwell 		01/08/2013

FAA Form 337 (10-06)

SDNY_GM_02758236

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245158

EFTA01261818

SDNY_GM_02758237

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245159

EFTA01261819

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/08/2013

Date

Installed Bose Headset Mount Panel Connector (LEMO) 6 Pin

Modify Pilots and Co-pilots microphone jack panels per Nextant Aerospace Drawing NX230-10001 & NX230-10001-02.

Wire connected per Nextant Aerospace Drawing NX230-10001-01.

Install circuit breakers in LH AFT Circuit Breaker panel as marked on Nextant Aerospace Drawing N995803 for part Number N995803-4.

This was FAA Approved by Dan Buzz DER-T-230019-CE on a FAA 8110-3 dated 01/08/2013 Using Nextant Aerospace Engineering Document 1037 Dated 12/17/2012

ICA

Continue with Normal Maintenance Manual Inspection Requirements for this area

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-08)

SDNY_GM_02758238

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245160

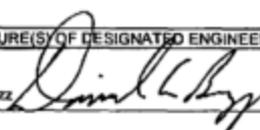
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SDNY_GM_02758239

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245161

EFTA01261821

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE 12/19/2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A S/N RK-260 Only	4. TYPE (Airplane, Radio, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace LLC
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Nextant Aerospace Document EO 1037 dated 12/17/2012	Engineering Order Install Bose Headset Panel Mount Connector (LEMO) 6 Pin		
	END		
	This approval is for the electrical systems aspects only.		
	Note: The above data does not meet the criteria of FAA Part 25 Policy Statement Number ANM-01-04; System Wiring Policy for Certification of Part 25 Airplanes. Additional approvals may be required.		
	This approval is for engineering design data only and is not an installation approval. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "APPLICABLE REQUIREMENTS." (Compliance with additional regulations not listed here may be required). This form does not constitute FAA approval of all the engineering design data necessary for substantiation of compliance to necessary requirements for the entire alteration.		
8. PURPOSE OF DATA Support major alteration to aircraft Hawker Beechcraft 400A RK-260 Only.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 25.1301(a)(b)(c)[ORIG], 25.1307(c)[25-23], 25.1357(a)(c)[ORIG], 25.1431(a)(b)[ORIG], 25.1541[ORIG] Note: Compliance has been found to the amendment levels indicated on Hawker Beechcraft TCDS Revision 26 Part 25 Amendment 25-1 through 25-40 to the above applicable requirements.			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. I (We) Therefore approve these data.			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)	
Daniel L. Buzz 	DETT-230019-CE	Systems and Equipment	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

GPO 901-613

SDNY_GM_02758240

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245162

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SDNY_GM_02758241

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245163

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 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)			Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N787TA			Serial No. RK-260	
	Make Raytheon Aircraft Company 400A			Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC			Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name Nextant Aerospace			<input type="checkbox"/> U.S. Certificated Mechanic	<input type="checkbox"/> Manufacturer	
Address 355 Richmond Road			<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.	
City Cleveland State OH			<input checked="" type="checkbox"/> Certificated Repair Station	CRS # 25NR667B	
Zip 44143-1453 Country USA			<input type="checkbox"/> Certificated Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A. Ockwell 		01/08/2013	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport	
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)	
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A. Ockwell 		01/08/2013	

FAA Form 337 (10-06)

SDNY_GM_02758242

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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SDNY_GM_02758243

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245165

EFTA01261825

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/08/2013

Date

This is a deviation from Nextant Aerospace Document EO-1037

Installation of Bose Headset Panel Mount Connector

Install circuit breakers for Bose Headset per drawing NX230-10001-01 Rev A and mark AFT Circuit breaker panel (N995803) per marked up overlay drawing N995803-4

Test headset connections using following procedure

A.) Optional Equipment:

- 1. Handheld VHF radio
- 2. COM-120 or equivalent VHF Comm test set.

B.) Close the following CB's.

PLT AUDIO, COPLT AUDIO, AURAL WARN, COMM NO1, COMM NO 2, RTU 1, RTU 2, 1MFD, 1MFD HEATER, 2MFD, MFD HEATER, 1PFD, 1PFD HEATER, 2PFD, 2PFD HEATER, PROCU HT NO1, FMS 1, FCS 1, PROCU HT NO2, FMS 2, FCS 2, 1 FSU PRI, 1 FSU BAT, AUX BATT 1 (ON AUX BATT 1 MOUNT) PLT BOSE AND CO-PLT BOSE circuit breakers.

C.) Connect External Power unit.

D.) Plug in BOSE headset into pilots LEMO jack.

E.) Turn on aircraft power.

Test Pilots BOSE:

Test may be accomplished using either the interphone system with speakers or one of the Optional items listed as Optional Equipment (Section A 1-2) Note that if the handheld radio method or the test set method is used, it is necessary to follow all local radio discipline guidelines for authorized operators using established procedures prior to keying up and transmitting on any active operational channel.

For Interphone testing,

- 1.) Set pilots and co-pilots audio panel to interphone. Turn on cabin speakers.
- 2.) Speak into mic and verify audio is present.
- 3.) Speak into opposite side cabin mic and verify audio is heard.

Test Co-Pilots BOSE:

- 4.) Repeat steps 1 - 3 for co-pilots side.

Perform the following (steps 5-8) regardless of headphone jack test method:

- 5.) Set STBY ARM SW to ARM position. Jumper WOW to simulate aircraft in air. Pull LH BUS TIE CB and LOAD BUS TIE CB. Verify pilots audio is inop. Set aircraft power switch to Emergency position.
- 6.) Verify that the Co-Pilots headset is still operational.
- 7.) Pull all Aux Batt and standby battery CB's. Co-pilot audio should be inoperative.
- 8.) Reset all CB's and return WOW switch to normal configuration.
- 9.) TEST COMPLETE

Additional Sheets Are Attached

FAA 337 Form (10-06)

SDNY_GM_02758244

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245166

EFTA01261826

SDNY_GM_02758245

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245167

EFTA01261827

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/08/2013

Date

To test using handheld radio or test set.

- 1.) Tune handheld radio or test set to suitable local VHF radio test frequency. (a suitable channel will be within the range 118.000MHz to 136.975MHz)
- 2.) Tune #1 RTU to frequency of handheld radio or tester.
- 2.) Set pilots and co-pilots audio panel to receive #1 VHF.
- 3.) Generate tone from test set or transmit voice from handheld radio. Audio should be heard on headset.
- 4.) set tester to receiver mode. Key microphone and speak into headset. Audio should be heard on test set or handheld radio.
- 5.) Repeat steps 3 - 4 above to test co-pilots headphone jack.
- 6.) Perform steps 5-9 from interphone testing procedure above.

This was Approved by Dan Buzz DER-T -230019-CE on a FAA 8110-3 Dated 01/08/2013 using Nextant Aerospace Document EO 1140 Dated 01/03/2013 and Nextant Aerospace Drawing NX230-10001-01 Rev A Dated 01/07/2013

.....End.....

Additional Sheets Are Attached

FAA 337 Form (10-06)

SDNY_GM_02758246

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

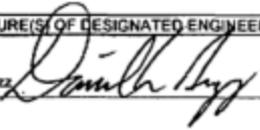
EFTA_00245168

EFTA01261828

SDNY_GM_02758247
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245169

EFTA01261829

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS		1. DATE 1-8-2013	
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A S/N RK-260 Only	4. TYPE (Airplane, Radio, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace LLC
LIST OF DATA			
6. IDENTIFICATION		7. TITLE	
Nextant Aerospace Document EO 1140 dated 1/3/2013 Nextant Aerospace Drawing NX230-10001-01 Rev. A dated 1/7/2013		Engineering Order Deviation to EO-1037 Electrical Drawing W/D Bose Lemo Connector (6 Pin) <p style="text-align: center;">-----END-----</p> This approval is for the electrical systems aspects only. Note: The above data does not meet the criteria of FAA Part 25 Policy Statement Number ANM-01-04; System Wiring Policy for Certification of Part 25 Airplanes. Additional approvals may be required. "This approval is for engineering design data only and is not an installation approval. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "APPLICABLE REQUIREMENTS." (Compliance with additional regulations not listed here may be required). This form does not constitute FAA approval of all the engineering design data necessary for substantiation of compliance to necessary requirements for the entire alteration."	
8. PURPOSE OF DATA Support major alteration to aircraft Hawker Beechcraft 400A RK-260 Only.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 25.1301(a)(b)(c)[ORIG], 25.1307(c)[25-23], 25.1357(a)(c)[ORIG], 25.1431(a)(b)[ORIG], 25.1541[ORIG] Note: Compliance has been found to the amendment levels indicated on Hawker Beechcraft TCDS Revision 26 Part 25 Amendment 25-1 through 25-40 to the above applicable requirements.			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <i>N/A</i> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. I (We) Therefore approve these data.			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)
Daniel L. Buzz 		DERT-230019-CE	Systems and Equipment

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

GPO 901-613

SDNY_GM_02758248

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245170

EFTA01261830

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SDNY_GM_02758249

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245171

EFTA01261831

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SDNY_GM_02758251

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245173

EFTA01261833

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)			Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N787TA			Serial No. RK-260	
	Make Raytheon Aircraft Company 400A			Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC			Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name Nextant Aerospace			<input type="checkbox"/> U.S. Certificated Mechanic		<input type="checkbox"/> Manufacturer
Address 355 Richmond Road			<input type="checkbox"/> Foreign Certificated Mechanic		C. Certificate No.
City Cleveland State OH			<input checked="" type="checkbox"/> Certificated Repair Station		CRS # 25NR667B
Zip 44143-1453 Country USA			<input type="checkbox"/> Certificated Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A. Ockwell 		01/08/2013	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport	
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)	
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A. Ockwell 		01/08/2013	

FAA Form 337 (10-06)

SDNY_GM_02758252

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245174

EFTA01261834

SDNY_GM_02758253

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245175

EFTA01261835

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

1/08/2013

Date

Replaced Dayton Granger Static Wick 16305 with Dayton Grainger Part Number 74001

This was FAA Approved on an 8110-3 by Dan Buzz DER-T 230019-CE on Jan 08 2012 Using Nextant Aerospace Engineering Order 1123 Dated 12/26/2012.

ICA

Changes to weight and Balance are negligible

Continue with Normal Hawker Beech 400A Inspection requirements

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758254

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245176

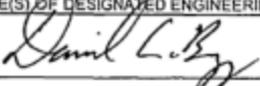
EFTA01261836

SDNY_GM_02758255

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245177

EFTA01261837

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE 1-8-2013
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A S/N RK-260 Only	4. TYPE (Airplane, Radio, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace LLC
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Nextant Aerospace Document EO 1123 dated 12/26/2012	Engineering Order Replace D/G 16305 Static Wick with 740001		
	-----END-----		
	This approval is for the electrical systems aspects only.		
	"This approval is for engineering design data only and is not an installation approval. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "APPLICABLE REQUIREMENTS." (Compliance with additional regulations not listed here may be required). This form does not constitute FAA approval of all the engineering design data necessary for substantiation of compliance to necessary requirements for the entire alteration."		
8. PURPOSE OF DATA Support major alteration to aircraft Hawker Beechcraft 400A RK-260 Only.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 25.1301(a)(c)[ORIG] Note: Compliance has been found to the amendment levels indicated on Hawker Beechcraft TCDS Revision 26 Part 25 Amendment 25-1 through 25-40 to the above applicable requirements.			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. I (We) Therefore approve these data.			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)	
Daniel L. Buzz 	DETR-230019-CE	Systems and Equipment	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

GPO 901-613

SDNY_GM_02758256

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245178

EFTA01261838

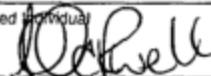
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SDNY_GM_02758257

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245179

EFTA01261839

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260	
	Make Raytheon Aircraft Company 400A		Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only			
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name Nextant Aerospace Address 355 Richmond Road City Cleveland State OH Zip 44143-1453 Country USA		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		
		<input type="checkbox"/> Manufacturer C. Certificate No. CRS # 25NR667B		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A.Ockwell  01/04/2013		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A.Ockwell  01/04/2013		

FAA Form 337 (10-06)

SDNY_GM_02758258

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245180

EFTA01261840

SDNY_GM_02758259
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245181

EFTA01261841

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Title: Ice Detector Master Caution / Master Warning Light on Constantly

Re-Installed the previously removed components (Ref S/B 30-2600, Rev 2)

- 1.) Isolation Relay K255 P/N: 50-380048-1
- 2.) Diode CR654, P/N: 132408-22
- 3.) Ground stud GS151A P/N: MS35206-246 (Stud may be marked GS152)
As shown in chapter 30-01-01-01 of AMM.
- 4.) Wire number corrections on wiring diagram 30-01-01-01 Page 1 correct drawing errors on original drawing. Inspect, verify and correct wiring per EO corrections on EO page 2

See FAA Form 8110-3 signed by Daniel Buzz DERT - 230019-CE. Dated 8-13-2012 and Nextant Aerospace Engineering Order EO 130-12 Rev D. Dated 7-18-2012

ICA

Weight and Balance Changes are Negligible

Continue to Use Existing Beechjet 400/400A Maintenance Manuals Inspection Procedure for this Area

-----END-----

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758260

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245182

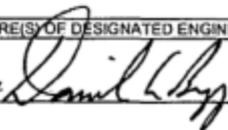
EFTA01261842

SDNY_GM_02758261

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245183

EFTA01261843

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE 8-13-2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A S/N RK-260 Only	4. TYPE (Airplane, Radio, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace LLC
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Nextant Aerospace Document 130-12 dated 7-18-12 Rev. D	Engineering Order Ice Detector MC/MW Light On Constantly		
	END		
	This approval is for the electrical systems aspects only.		
8. PURPOSE OF DATA Support major alteration to aircraft Hawker Beechcraft 400A RK-260 Only.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 25.1301(a) [Orig] Note: Compliance has been found to the amendment levels indicated on Hawker Beechcraft TCDS Revision 26 Part 25 Amendment 25-1 through 25-40 to the above applicable requirements.			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. I (We) Therefore approve these data.			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S) Daniel L. Buzz 	12. DESIGNATION NUMBERS(S) DERT-230019-CE	13. CLASSIFICATION(S) Systems and Equipment	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

GPO 901-613

SDNY_GM_02758262

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245184

EFTA01261844

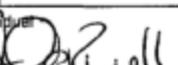


SDNY_GM_02758263

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245185

EFTA01261845

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260	
	Make Raytheon Aircraft Company 400A		Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only			
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name Nextant Aerospace		<input type="checkbox"/> U.S. Certificated Mechanic		<input type="checkbox"/> Manufacturer
Address 355 Richmond Road		<input type="checkbox"/> Foreign Certificated Mechanic		C. Certificate No.
City Cleveland State OH		<input checked="" type="checkbox"/> Certificated Repair Station		CRS # 25NR667B
Zip 44143-1453 Country USA		<input type="checkbox"/> Certificated Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A. Ockwell 		01/04/2013
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A. Ockwell 		01/04/2013

FAA Form 337 (10-06)

SDNY_GM_02758264

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245186

EFTA01261846

SDNY_GM_02758265
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245187

EFTA01261847

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Incorporated the following Cockpit Glareshield Repair Procedure.

Removed the Cockpit Glareshield and installed the following parts to reinforce the Glareshield. REO051-11-06, NXRP45A88736-103, NXRP45A88736-102. Reinstalled the repaired Glareshield IAW HBC 400A Maintenance Manual 31-10-00-201.

See Attached FAA Form 8110-3 signed by DERT-410115-CE Paul Hedding Dated December 05, 2012 using Nextant Aerospace Engineering Order EO118-12. Rev C. Dated January 30, 2012.

ICA

Weight change is Negligible.

Continue with Beechjet 400/400A Maintenance Manual Inspection Procedures

End

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758266

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245188

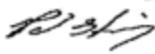
EFTA01261848

SDNY_GM_02758267

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245189

EFTA01261849

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION			1. DATE December 5, 2012
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Nextant Aerospace LLC Richmond Heights, Ohio
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
EO 118-12 Rev. C January 30, 2012	Repair Glaresheild Procedure.		
	-----END-----		
	Notes:		
	<ol style="list-style-type: none"> 1) The structural aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements." 2) This form constitutes FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration. 		
8. PURPOSE OF DATA: Engineering data in support of major alteration to Hawker Beechcraft 400A, S/N RK-260.			
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], and 25.625(a)(b)(c)[25-23].			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> , have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.			
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)	
 Paul Hedding	DETR-410115-CE	Structures	

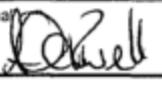
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758269

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245191

EFTA01261851

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)			Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N787TA			Serial No. RK-260	
	Make Raytheon Aircraft Company 400A			Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC			Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name Nextant Aerospace			<input type="checkbox"/> U.S. Certificated Mechanic	<input type="checkbox"/> Manufacturer	
Address 355 Richmond Road			<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.	
City Cleveland State OH			<input checked="" type="checkbox"/> Certificated Repair Station	CRS # 25NR667B	
Zip 44143-1453 Country USA			<input type="checkbox"/> Certificated Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A.Ockwell 		01/04/2013	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport	
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)	
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A.Ockwell 		01/04/2013	

FAA Form 337 (10-06)

SDNY_GM_02758270

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245192

EFTA01261852

SDNY_GM_02758271
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245193

EFTA01261853

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Provisions for J9107 in Cockpit Sidewall at FS122.735

Added mount provisions for J9107 in bracket assy. 45A88588-3 at Flight Station 122.735 provided in Structure Repair Manual Document 53-20-05-001. Use receptacle M53470W18.

See attached FAA Form 8110-3 Signed By DERT-410115-CE Paul Hedding. Dated May 14, 2012 using Nextant Aerospace Engineering Order 115-12 Rev B, Dated April 04, 2012 for details.

ICA

Weight change is Negligible.

Continue with Beechjet 400/400A Maintenance Manual Inspection Procedures.

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-08)

SDNY_GM_02758272

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245194

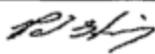
EFTA01261854

SDNY_GM_02758273

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245195

EFTA01261855

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION			1. DATE May 14, 2012
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Nextant Aerospace LLC Richmond Heights, Ohio
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
EO 115-12 Rev. B April 4, 2012	Mount Provisions for J9107.		
	-----END-----		
	Notes:		
	<ol style="list-style-type: none"> 1) The structural aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements." 2) This form constitutes FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration. 		
8. PURPOSE OF DATA: Engineering data in support of major alteration to Hawker Beechcraft 400A, S/N RK-260.			
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], and 25.625(a)(b)(c)[25-23].			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> , have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.			
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)	
 Paul Hedding	DETR-410115-CE	Structures	

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758275

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245197

EFTA01261857

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)			Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N787TA			Serial No. RK-260	
	Make Raytheon Aircraft Company 400A			Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC			Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name Nextant Aerospace Address 355 Richmond Road City Cleveland State OH Zip 44143-1453 Country USA			<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		
			<input type="checkbox"/> Manufacturer C. Certificate No. CRS # 25NR667B		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A.Ockwell  01/04/2013			
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport	
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)	
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A.Ockwell  01/04/2013			

FAA Form 337 (10-06)

SDNY_GM_02758276

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245198

EFTA01261858

SDNY_GM_02758277

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245199

EFTA01261859

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Installed Alternative Fuel Cap Screws in L/H and R/H Fuel Cap

Removed the NAS517-3-3 Screws and Installed MS34694 -C50

This was FAA Approved by Robin Harrison DERT-605234-NM on FAA 8110-3 Dated 06 December 2012 using Nextant Aerospace Engineering Order 083-11. Rev C Dated Nov 09.2011

ICA

Weight and Balance changes are negligible

Continue with Normal Hawker Beechjet400A Maintenance Manual Inspection Requirements for this Area

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758278

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245200

EFTA01261860

SDNY_GM_02758279

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245201

EFTA01261861

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION				1. DATE 06 December 2012
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A (see note 3)	4. TYPE (Aircraft, Engine, Propeller, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace (see note 4)	
LIST OF DATA				
6. IDENTIFICATION	7. TITLE			
Nextant Aerospace Document; EO NO. 083-11 Rev. C Date: 09 Nov. 2011 -----end-----	Alternate Fuel Cap Screws. -----end----- Notes: 1. Detail Design and Structural Approval only. 2. This approval indicates the data listed above demonstrates compliance only with the regulations specified by paragraphs and subparagraphs listed below as "Applicable Requirements." Compliance to additional regulations not listed here may be required. This form DOES NOT constitute FAA approval of all the data necessary for substantiation of compliance to necessary requirements for the entire alteration/repair. This approval is only for the engineering design and/or substantiation data. It is not an installation approval. 3. Model: Hawker Beechcraft 400A (Raytheon Beechjet), identified by FAA TCDS No. A16SW. 4. Applicant: Nextant Aerospace, 26180 Curtiss Wright Pkwy, Cleveland, OH 44143 U.S. LLC 25NR667B. 5. Applicable aircraft: RK-001 thru RK-999, except RK-279, 108, 028, 274, 295, 230, 292, 123, 225, 137, 248, 289, 310, 252, 195, 239, 284, 368, 334, 327, 146, 260 and 413. -----end-----			
8. PURPOSE OF DATA To show compliance with applicable regulations in support of a Major Repair/Alteration FAA Form 337 Approval for Alternate Fuel Cap Screws, Repair for Nextant Aerospace - Hawker Beechcraft 400A Model Aircraft S/N identified by Serial No's listed in Note 5, above.				
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR, PART25 paragraphs: 25.305(a)(b) Amdt. 25-86; 25.307(a) Amdt. 25-72; 25.601 Amdt. 25-0; 25.603(a)(b)(c) Amdt. 25-46; 25.605(a) Amdt. 25-46; and 25.609(a) Amdt. 25-0.				
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>none</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. I (We) Therefore <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)		
 Robin Harrison	DERT-605234-NM	Structures		

6628-1

FAA FORM 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758280

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245202

EFTA01261862

SDNY_GM_02758281

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245203

EFTA01261863

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark <p style="text-align: center;">N787TA</p>		Serial No. RK-260	
	Make <p style="text-align: center;">Raytheon Aircraft Company 400A</p>		Model <p style="text-align: center;">400A</p>	Series
2. Owner	Name (As shown on registration certificate) <p style="text-align: center;">Flight Options .LLC</p>		Address (As shown on registration certificate) Address <u>26180 Curtiss Wright PKWY</u> City <u>Richmond Heights</u> State <u>OH</u> Zip <u>44143-1453</u> Country <u>USA</u>	
	3. For FAA Use Only			
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name <u>Nextant Aerospace</u> Address <u>26180 Curtiss Wright Parkway</u> City <u>Richmond Heights</u> State <u>OH</u> Zip <u>44143-1453</u> Country <u>USA</u>		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		
		<input type="checkbox"/> Manufacturer C. Certificate No. <p style="text-align: center;">CRS # 25NR667B</p>		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual <u>A.Ockwell</u>  <u>01/04/2013</u>		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual <u>A.Ockwell</u>  <u>01/04/2013</u>		

FAA Form 337 (10-06)

SDNY_GM_02758282

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245204

EFTA01261864

SDNY_GM_02758283

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245205

EFTA01261865

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Modify Existing GPS / XM Antenna mounting Doubler plate (P/N 11892-6)

Modified Doubler view of drawing RRK279EO052-11-2.

Modification will be comprised of enlarging the aft portion of existing 2" center hole by drilling 0.625" hole spaced per drawing. Radius blends intersecting edge to 2" hole per drawing.

This was FAA approved by Paul Hedding DERT-410115-CE on 8110-3 Dated December 05, 2012 using Nextant EO 052-11 Rev B Dated November 13, 2012

ICA

Weight and Balance change is negligible.

Continue with Beechjet 400/400A Normal Maintenance Manual Inspections for this Area

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758284

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245206

EFTA01261866

SDNY_GM_02758285

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245207

EFTA01261867

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE December 5, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Nextant Aerospace LLC Richmond Heights, Ohio	
LIST OF DATA				
6. IDENTIFICATION		7. TITLE		
EO 052-11 Rev. B November 13, 2012		GPS / XM Antenna Doubler Engineering Order.		
		-----END-----		
		Notes:		
		<ol style="list-style-type: none"> 1) The structural aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements." 2) This form does not constitute FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration. The electrical requirements are not included in this approval and require separate approval. 		
8. PURPOSE OF DATA: Engineering data in support of major alteration to Hawker Beechcraft 400A, S/N RK-260.				
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.365(a)(b)(c)(d)(f)(g)[25-00], 25.562(c)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], 25.625(a)(b)(c)[25-23], and 25.789[25-32].				
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.				
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)	
 Paul Hedding		DERT-410115-CE	Structures	

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

2

SDNY_GM_02758287

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245209

EFTA01261869

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)			Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N787TA			Serial No. RK-260	
	Make Raytheon Aircraft Company 400A			Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC			Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name Nextant Aerospace			<input type="checkbox"/> U.S. Certificated Mechanic		<input type="checkbox"/> Manufacturer
Address 335 Richmond Road			<input type="checkbox"/> Foreign Certificated Mechanic		C. Certificate No.
City Cleveland OH State OH			<input checked="" type="checkbox"/> Certificated Repair Station		CRS # 25NR667B
Zip 44143-1453 Country USA			<input type="checkbox"/> Certificated Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A.Ockwell 		01/04/2012	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport	
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)	
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A.Ockwell 		01/04/2012	

FAA Form 337 (10-06)

SDNY_GM_02758288

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245210

EFTA01261870

SDNY_GM_02758289

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245211

EFTA01261871

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2012

Date

Modified Baggage Ceiling Panel to Accommodate the Engine FADEC wire harness.

1. Fabricated two panel pieces, N251103-001 and N251103-002.
2. Installed N251103-001 to existing baggage compartment structure 128-440095-1 using two screws and install N251103-002 as shown with a trim to fit honeycomb piece fabricated from interiors using potted inserts.
3. Attach honeycomb baggage panel to N251103-001 in upper aft baggage area using three 40S5-21 Camloc quarter-turn fasteners and 214-16D receptacles. Secured N251203-002 to existing upper aft baggage panel with two Camloc 40S5-21 quarter-turn fasteners and 214-16D receptacles.

See attached FAA Form 8110-3 Signed by DERT-410115-CE Paul Hedding on November 28 2012 using Nextant Aerospace Engineering Order EO 029. Rev C Dated November 15.2012.

ICA

Weight change is Negligible.
Continue with Beechjet 400/400A Maintenance Manual Inspection Procedures.

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758290

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245212

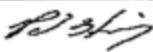
EFTA01261872

SDNY_GM_02758291

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245213

EFTA01261873

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION			1. DATE November 28, 2012	
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Nextant Aerospace LLC Richmond Heights, Ohio	
LIST OF DATA				
6. IDENTIFICATION		7. TITLE		
EO 029 Rev. C November 15, 2012		Upper Aft Baggage Panel Mod.		
		-----END-----		
Notes:		<p>1) The structural aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements."</p> <p>2) This form constitutes FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration.</p>		
8. PURPOSE OF DATA: Engineering data in support of major alteration to Hawker Beechcraft 400A, S/N RK-260.				
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.562(c)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], 25.625(a)(b)(c)[25-23], and 25.789[25-32].				
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> , have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.				
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)	
 Paul Hedding		DERT-410115-CE	Structures	

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758293

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245215

EFTA01261875



**MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)**

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)

1. Aircraft	Nationality and Registration Mark N787TA	Serial No. RK-260	
	Make Raytheon Aircraft Company 400A	Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC	Address (As shown on registration certificate) 26180 Curtiss Wright PKWY	
		City Richmond Heights	State OH
		Zip 44143-1453	Country USA

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency	
Name	Nextant Aerospace	<input type="checkbox"/> U.S. Certificated Mechanic	<input type="checkbox"/> Manufacturer
Address	355 Richmond Road	<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.
City	Cleveland State: OH	<input checked="" type="checkbox"/> Certificated Repair Station	CRS # 25NR667B
Zip	44143-1453 Country: USA	<input type="checkbox"/> Certificated Maintenance Organization	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual A.Ockwell	01/04/2013
---	--------------------------	---	-------------------

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)

Certificate or Designation No. CRS # 25NR667B	Signature/Date of Authorized Individual A.Ockwell	01/04/2013
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SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758295

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245217

EFTA01261877

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

01/04/2013

Nationality and Registration Mark

Date

The Aircraft comes from the Factory equipped with Aero Instruments PH1100-MU-L-1 (R-1) chromium plated Pitot Tubes.

This is to replace them with TSO approved Tubes PH1100-MU-L-EN (R-EN)

The Electrical Aspects of this was approved by Mr. Dan Buzz FAA DERT-230019-CE Systems and Equipment on an FAA 8110-3 Dated 11/7/2012 .Using Nextant Aerospace Engineering Order 024-11 Rev Original Dated 04/11/2011

The Mechanical Aspect was Approved by Paul Hedding FAA DERT-410115-CE Mechanical Systems on a FAA 8110-3 Dated Nov 09.2012 Using Nextant Aerospace Engineering Order 024-11 Rev Original Dated 4/11/2011

ICA.

Continue with Normal Hawker Beech Maintenance Manual Inspections for this Area

There are no Changes to the Weight and Balance

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758296

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245218

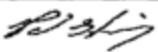
EFTA01261878

SDNY_GM_02758297

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245219

EFTA01261879

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION				1. DATE November 9, 2012
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Nextant Aerospace Cleveland, Ohio	
LIST OF DATA				
6. IDENTIFICATION	7. TITLE			
Engineering Order 024-11 Rev. Original April 11, 2011	Pitot Probe.			
	-----END-----			
	Notes:			
	<ol style="list-style-type: none"> 1) The Mechanical Systems aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements." 2) This form does not constitute FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration. The electrical regulations are not covered by this approval and require separate approval. 			
8. PURPOSE OF DATA: In support of a major alteration to Beech 400A S/N RK-260 aircraft.				
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.601[25-00], 25.603(a)(b)[25-38], 25.605[25-00], 25.607(a)(b)(c)[25-23], and 25.1325(b)(c)[25-12].				
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> , have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.				
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)		
 Paul Hedding	DERT-410115-CE	Mechanical Systems		

FAA Form 8110-3 (03-10) SUPERSEDES PREVIOUS EDITION File Number: H12618-01

SDNY_GM_02758298

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245220

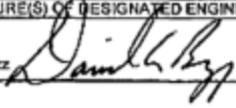
EFTA01261880

SDNY_GM_02758299

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245221

EFTA01261881

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE 11/7/2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A S/N RK-260 Only	4. TYPE (Airplane, Radio, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace LLC
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Nextant Aerospace Document 024-11 dated 4/11/11 Rev. Original	Engineering Order Pitot Probe		
	-----END-----		
	This approval is for the electrical systems aspects only.		
8. PURPOSE OF DATA Support major alteration to aircraft Hawker Beechcraft 400A RK-260 Only.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 25.1301(a)(c) [Orig] Note: Compliance has been found to the amendment levels indicated on Hawker Beechcraft TCDS Revision 26 Part 25 Amendment 25-1 through 25-40 to the above applicable requirements.			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered N/A have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. I (We) Therefore approve these data.			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S) Daniel L. Buzz 	12. DESIGNATION NUMBER(S) DERT-230019-CE	13. CLASSIFICATION(S) Systems and Equipment	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

GPO 901-813

SDNY_GM_02758300

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245222

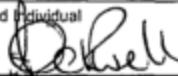
EFTA01261882

SDNY_GM_02758301

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245223

EFTA01261883

 U.S. Department of Transportation Federal Aviation Administration		MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark <div style="text-align: center;">N787TA</div>		Serial No. RK-260		
	Make <div style="text-align: center;">Raytheon Aircraft Company 400A</div>		Model <div style="text-align: center;">400A</div>	Series	
2. Owner	Name (As shown on registration certificate) <div style="text-align: center;">Flight Options .LLC</div>		Address (As shown on registration certificate) Address <u>26180 Curtiss Wright PKWY</u> City <u>Richmond Heights</u> State <u>OH</u> Zip <u>44143-1453</u> Country <u>USA</u>		
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name <u>Nextant Aerospace</u> Address <u>355 Richmond Road</u> City <u>Cleveland</u> State <u>OH</u> Zip <u>44143-1453</u> Country <u>USA</u>			<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		<input type="checkbox"/> Manufacturer C. Certificate No. <u>CRS # 25NR667B</u>
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual <u>A. Ockwell</u> 		01/04/2013	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	<input type="checkbox"/> FAA FIT Standards Inspector	<input type="checkbox"/> Manufacturer	<input type="checkbox"/> Maintenance Organization		<input type="checkbox"/> Person Approved by Canadian Department of Transport
	<input type="checkbox"/> FAA Designee	<input checked="" type="checkbox"/> Repair Station	<input type="checkbox"/> Inspection Authorization		Other (Specify)
Certificate or Designation No. <u>CRS # 25NR667B</u>		Signature/Date of Authorized Individual <u>A. Ockwell</u>		01/04/2013	

FAA Form 337 (10-05)

SDNY_GM_02758302

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245224

EFTA01261884

SDNY_GM_02758303

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245225

EFTA01261885

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Installed the following In-Flight Entertainment System in accordance with STC # ST03960AT and in accordance with Nextant Aerospace Master Drawing List DB1002067. Revision E. Dated January 26 2012.

The Emteq LED cabin lights provide high reliability low power consumption Indirect Cabin Lighting, Galley Task Lighting and Lavatory Vanity Lighting. Installed LED Passenger Safety Signs The cabin lights are interfaced to the existing cabin lighting power distribution and control switches.

Installed Cabin LED Lighting System without the Venue System in accordance with DB1107014. Rev IR Dated January 26 2012.

Installed LED Passenger Safety Signs in accordance with Wiring Diagram DB 1108001 Rev IR January 26 2012

Scheduled Inspection and Maintenance.

Inspect System Installation Component including wire Bundles, Cables Racks and Connector Condition and security in conjunction with the B Check (every 400 hours after Initial 200 Hours "I" inspection) tasks and "C" Check (every 1200 after initial 200 Hr "I" Inspection) Per Hawker Beechjet 400A Maintenance Manual 5.20.02 requirements listed within Scheduled Inspection Section RK-1 and above

ICA for LED Cabin Lighting System Ref to Nextant Aerospace Document No DB1108019 Rev IR Dated September 22 2011 or later FAA Accepted Revisions

Installed Iridium Sat Comm and Broadband Internet System with Aircell Axxess II and ATG 4000 in Accordance with DB 1001013. Rev D

ICA for Aircell System Ref to Nextant Aerospace Document No DB1105003. Rev A dated September 22 2011 or later FAA Accepted Revisions

The following manuals contain complete detailed maintenance instructions and should be consulted for all maintenance activities not covered within this ICA:

- Aircell Axxess II Installation Manual D12004 Revision E, dated March 2010 or later applicable revision
- Aircell ATG-4000 Installation Manual D13485 Revision B Dated November 2009 or later applicable revision
- Aircell Axxess II WLAN AFM Supplement DB1002072 Rev IR dated March 22 2012 for the has been Installed in the AFM

Installed the Astronics DC-AC 115 VAC Power Supply in Accordance with the following Nextant Aerospace Drawings This is Located under the Seat Divan

- NX252-10007-03. Rev IR Dated June 04 2012 Channel Hat Inverter Mount
- NX252-10007-04. Rev IR Dated June 12, 2012. Assy Inverter 115 VAC, W/Heatsink and Hat.
- NX252-10007-05 Rev IR. Dated June 14 2012 Mount Plate ,Single Inverter. Under Divan
- NX252-10007-06 Rev IR Dated. June 13. 2012. Assy, Inverter, Divan Mount
- NX252-10007-12. Rev IR. Dated June 14 . 2012 Assy ,Bracket Plate Mount Inverter ,Under Divan
- NX252-10007-13. Rev IR, Dated June 14 2012. Angle.1x1" .Inverter Mount

ICA for Astronics 115 VAC Cabin Inverter Ref to Nextant Aerospace Report No DB1108018 Rev IR Date 9/22/2011 or later FAA Accepted Revision.

Aircraft Flight Supplement DB1109012 has been installed on the Aircraft

See Current Weight and Balance Dated 12/21/2012

.....End.....

Additional Sheets Are Attached

SDNY_GM_02758305

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245227

EFTA01261887

United States of America
Department of Transportation -- Federal Aviation Administration

Supplemental Type Certificate

Number ST03960AT

This certificate issued to Nextant Aerospace LLC
3800 Southern Blvd.
West Palm Beach, FL 33406

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified herein meets the airworthiness requirements of Part 25 of the Federal Aviation Regulations.

Original Product - Type Certificate Number : A16SW
Make : Hawker Beechcraft Corporation
Model : 400A

Description of Type Design Change:

Installation of Rockwell Collins Venue In-Flight Entertainment(IFE), Aircell Axxess Cabin II Iridium Phone and High Speed Internet System with Wireless Local Area Network (WLAN) Astronics 115VAC 60HZ Cabin Inverter, and Emteq LED Cabin Lighting System in accordance with Nextant Aerospace LLC, Master Drawing List DB1002067, Revision E, dated January 26, 2012, or later FAA approved revision.

Limitations and Conditions: This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated, unless it is determine by the installer that the interrelationship between this change and any other previously approved modifications will produce no adverse effect upon the airworthiness of that airplane. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application : July 28, 2012

Date reissued :

Date of issuance : March 22, 2012

Date amended :



By direction of the Administrator

[Signature]
(Signature)

Melvin D. Taylor
Manager
Atlanta Aircraft Certification Office

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2(11-08)

PAGE 1 of 2 PAGES

This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758306

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245228

EFTA01261888

2

SDNY_GM_02758307

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245229

EFTA01261889

United States of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate
(Continuation Sheet)

Number ST03960AT

When the Rockwell Collins Venue system is installed Airplane Flight Manual Supplement, DB1106015, Revision A, dated March 22, 2012, or later FAA approved revision, is required.

When the Aircell Axxess II system is installed Airplane Flight Manual Supplement, DB1002072, Revision IR, dated March 22, 2012, or later FAA approved revision, is required.

When the Astronics Cabin Inverter system is installed Airplane Flight Manual Supplement, DB1109012, Revision IR, dated March 22, 2012, or later FAA approved revision, is required.

This system is intended to provide internet connection and email services to the airplane's cabin using portable electronic devices (PEDs). Any other intended function of this equipment will require a re-examination of the certification basis.

When the Rockwell Collins Venue system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1002068, Rev IR, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

When the Aircell Axxess II system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1105003, Rev A, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

When the Astronics Cabin Inverter system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1108018, Rev IR, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

When the Emteq LED Cabin Lighting system is installed Instructions for Continued Airworthiness (ICA), Nextant Aerospace Document No. DB1108019, Rev IR, dated September 22, 2011, or later FAA accepted revisions must be made available to the operator at the time of installation.

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2-1 (10-69)

PAGE 2 of 3 PAGES

This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758308

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245230

EFTA01261890

SDNY_GM_02758309

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245231

EFTA01261891

United States of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number ST03960AT

Certification Basis:

The Certification Basis for this installation is per Type Certificate Data Sheet A16SW original Certification Basis, Federal Aviation Regulation (FAR), Part 25 as amended by Amendment 25-1 through 25-40 plus FAR 25.1335, 25.1351(d), 25.1353(c)(5), and 25.1447 of Amendment 25-41; FAR 25.29, FAR 25.255, and FAR 25.1353(c)(6) of Amendment 25-42; and FAR 25.361(b) and 25.1329(h) of Amendment 25-46.

In addition the following specified rules amendment levels were used for this installation.

25.571	25.1353(a)(c)	25.1357	25.1431	25.1529
[25-54]	[25-123]	[25-123]	[25-113]	[25-54]

[-] Indicates amendment level

-----END-----

SDNY_GM_02758311

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245233

EFTA01261893

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number	
	For FAA Use Only				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)					
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260		
	Make Raytheon Aircraft Company 400A		Model 400A	Series	
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) Address: 26180 Curtiss Wright PKWY City: Richmond Heights State: OH Zip: 44143-1453 Country: USA		
	3. For FAA Use Only				
4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT			
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		
6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name Nextant Aerospace 66 355 Richmond Road City Cleveland State: OH Zip 44143-1453 Country USA			<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization		<input type="checkbox"/> Manufacturer C. Certificate No. CRS # 25NR667B
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A. Ockwell 		1/04/2013	
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	<input type="checkbox"/> FAA Fit Standards Inspector	<input type="checkbox"/> Manufacturer	<input type="checkbox"/> Maintenance Organization	Person Approved by Canadian Department of Transport	
	<input checked="" type="checkbox"/> FAA Designee	<input checked="" type="checkbox"/> Repair Station	<input type="checkbox"/> Inspection Authorization	Other (Specify)	
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A. Ockwell 		01/04/2013	

FAA Form 337 (10-06)

SDNY_GM_02758312

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245234

EFTA01261894

SDNY_GM_02758313

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245235

EFTA01261895

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

01/04/2013

Nationality and Registration Mark

Date

Nextant Aerospace Carried out a Repair to the Horizontal Stabilizer Ribs

Using Nextant Aerospace Kit Part Number NXK-128-6200 Rev B. Dated July 16, 2012.
The following Ribs were replaced.

NXR-45A21112-007	Rib – H Stab Cant STA 13.779.
NXR-45A21113-007	Rib – H Stab Cant STA 29.921.
NXR-45A21101-021	Rib – H Stab Cant STA 46.063.
NXR-45A21116-007	Rib – H Stab Cant STA 62.205
NXR-45A21117-007	Rib – H Stab Cant STA 78.346

The following Ribs were not Replaced as there was not Damaged.

NXR128-620020-001
NXR128-620020-002

This Kit was FAA Approved by Paul Hedding DER-T 410115-CE Structures on an FAA 8110-3. Dated December 5, 2012. Using Nextant Aerospace Engineering Drawing NXR-128-6200 Rev B Dated July 16, 2012

ICA

Ref to Current weight and Balance dated 12/06/2012

Continue with Normal Hawker Beech 400A Maintenance Practices for this Area

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758314

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245236

EFTA01261896

SDNY_GM_02758315

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245237

EFTA01261897

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE December 5, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. Horizontal Stabilizer P/N 45A21001-635	4. TYPE (Aircraft, Engine, Propeller, etc.) Component	5. NAME OF APPLICANT Nextant Aerospace LLC Richmond Heights, Ohio
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
NXR-128-6200 Rev. B July 16, 2012	Kit – Horizontal Stabilizer Rib Replacement Installation Instructions.		
NXR-45A21112-007 Rev. C July 13, 2012	Rib – H Stab Cant STA 13.779.		
NXR-45A21113-007 Rev. B July 13, 2012	Rib – H Stab Cant STA 29.921.		
NXR-45A21114-007 Rev. B July 13, 2012	Rib – H Stab Cant STA 46.063.		
NXR-45A21115-007 Rev. C July 13, 2012	Rib – H Stab Cant STA 46.063.		
NXR-45A21101-021 Rev. A July 19, 2012	Rib Assy – H Stab Cant STA 46.063.		
NXR-45A21116-007 Rev. B July 13, 2012	Rib – H Stab Cant STA 62.205.		
NXR-45A21117-007 Rev. B July 13, 2012	Rib – H Stab Cant STA 78.346.		
8. PURPOSE OF DATA: Engineering data in support of major repair to Hawker Beechcraft Horizontal Stabilizer P/N 45A21001-635, S/N RK-260.			
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], and 25.625(a)(b)(c)[25-23].			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>2</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed. <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)	
 Paul Hedding	DERT-410115-CE	Structures	

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758317
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245239

EFTA01261899

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION				1. DATE December 5, 2012
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. Horizontal Stabilizer P/N 45A21001-635	4. TYPE (Aircraft, Engine, Propeller, etc.) Component	5. NAME OF APPLICANT Nextant Aerospace LLC Richmond Heights, Ohio	
LIST OF DATA				
6. IDENTIFICATION	7. TITLE			
	END			
Notes:				
<p>1) The structural aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements."</p> <p>2) This form constitutes FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire repair.</p>				
8. PURPOSE OF DATA: Engineering data in support of major repair to Hawker Beechcraft Horizontal Stabilizer P/N 45A21001-635, S/N RK-260.				
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], and 25.625(a)(b)(c)[25-23].				
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>2</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.				
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)		
 Paul Hedding	DERT-410115-CE	Structures		

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758319

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245241

EFTA01261901

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260	
	Make Raytheon Aircraft Company 400A		Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only			
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name Nextant Aerospace		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Manufacturer		
Address 355 Richmond Road		<input type="checkbox"/> Foreign Certificated Mechanic C. Certificate No.		
City Cleveland State OH		<input checked="" type="checkbox"/> Certificated Repair Station		
Zip 44143-1453 Country USA		<input type="checkbox"/> Certificated Maintenance Organization CRS # 25NR667B		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A. Ockwell  01/04/2013		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A. Ockwell  01/04/2013		

FAA Form 337 (10-05)

SDNY_GM_02758320

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245242

EFTA01261902

2

SDNY_GM_02758321

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245243

EFTA01261903

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/ 04/2013

Date

Nextant Aerospace Install Flight Display Systems Moving Map System.

Install CB panel P/N: NX253-10003-01 and Overlay P/N: NX253-10003-02 using two (2) each brackets, P/N: DB1104014.

Install FD200-7/8 Unit in lower Galley and Drawing NX253-1004-01 wire per drawing NX252-10001-01.(Ref: Flight Display System MAN-FD200CPU-7/8)

Install FD102CV unit on L/H side of Galley. Ref: MAN-FD102CV

Configure and Test per Flight Display System Document MAN-FD200CPU-7/8 Chapter 4

This Installation was FAA Approved on an 8110-3 by Dan Buzz DERT-230019-CE. On Dec 27.2012 using Nextant Aerospace Engineering Order 1121 Dated 12/22/2012 Rev IR

ICA

The FD200CPU-7/8 is a Moving Map Designed not to require regular General Maintenance.

Ref to Weight and Balance Dated 12/21/2012.

.....End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758322

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245244

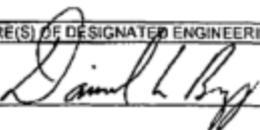
EFTA01261904

SDNY_GM_02758323

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245245

EFTA01261905

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE 12/27/2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft	3. MODEL NO. 400A S/N RK-260 Only	4. TYPE (Airplane, Rotor, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Nextant Aerospace LLC
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Nextant Aerospace Document EO 1121 dated 12/22/2012 Rev. IR	Engineering Order Install Moving Map System		
END			
This approval is for the electrical systems aspects only.			
Note: The above data does not meet the criteria of FAA Part 25 Policy Statement Number ANM-01-04; System Wiring Policy for Certification of Part 25 Airplanes. Additional approvals may be required.			
This approval is for engineering design data only and is not an installation approval. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "APPLICABLE REQUIREMENTS." (Compliance with additional regulations not listed here may be required). This form does not constitute FAA approval of all the engineering design data necessary for substantiation of compliance to necessary requirements for the entire alteration.			
8. PURPOSE OF DATA Support major alteration to aircraft Hawker Beechcraft 400A RK-260 Only.			
9. APPLICABLE REQUIREMENTS (List specific section(s)) 14 CFR 25.1301(a)(b)(c)[ORIG], 25.1307(c)[25-23], 25.1357(a)(c)[ORIG], 25.1431(a)(b)[ORIG], 25.1541[ORIG] Note: Compliance has been found to the amendment levels indicated on Hawker Beechcraft TCDS Revision 26 Part 25 Amendment 25-1 through 25-40 to the above applicable requirements.			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. I (We) Therefore approve these data.			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S) Daniel L. Buzz 	12. DESIGNATION NUMBER(S) DERT-230019-CE	13. CLASSIFICATION(S) Systems and Equipment	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

GPO 901-613

SDNY_GM_02758324

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245246

EFTA01261906

1

SDNY_GM_02758325

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245247

EFTA01261907

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260	
	Make Raytheon Aircraft Company 400A		Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options .LLC		Address (As shown on registration certificate) Address 26180 Curtiss Wright PKWY City Richmond Heights State OH Zip 44143-1453 Country USA	
	3. For FAA Use Only			
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name Nextant Aerospace		<input type="checkbox"/> U.S. Certificated Mechanic	<input type="checkbox"/> Manufacturer	
Address 335 Richmond Road		<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.	
City Cleveland State OH		<input checked="" type="checkbox"/> Certificated Repair Station	CRS # 25NR667B	
Zip 44143-1453 Country USA		<input type="checkbox"/> Certificated Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual A. Ockwell 		01/04/2013
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS # 25NR667B		Signature/Date of Authorized Individual A. Ockwell 		01/04/2013

FAA Form 337 (10-08)

SDNY_GM_02758326

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245248

EFTA01261908

SDNY_GM_02758327

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245249

EFTA01261909

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

Nationality and Registration Mark

01/04/2013

Date

Nextant Aerospace Completed STC # ST02371LA:

Replacement of the Pratt and Whitney JT15D-5 Series Engines with the Williams International FJ44-3AP Engines with Full FADEC and Associated Systems in accordance with Nextant Aerospace MDL -NXT-001 Rev IR Dated 08/22/2011 or later FAA approved revisions, and FAA Approved AFM Supplement NXT-1-AFMS Rev IR dated 9/15/2011 of later FAA approved revisions.

Installed Left Engine Part Number. 111000-202 Serial No. 252745
Installed Right Engine Part Number 111000-202 Serial No. 252746

ICA

AFM Supplement was installed on Aircraft.
Nextant Aerospace Instructions for Continued Airworthiness Document NXT-ICA Rev IR or later FAA accepted revisions are included in the Aircraft Maintenance Records.
Carried out Engine Ground Runs and System OPS checks IAW Ground Test Plan 373-00-0024 Rev F.

See Weight and Balance Dated 12/22/2012

..... End.....

Additional Sheets Are Attached

FAA Form 337 (10-06)

SDNY_GM_02758328

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245250

EFTA01261910

SDNY_GM_02758329

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245251

EFTA01261911

United States Of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

Number ST02371LA

This Certificate issued to Nextant Aerospace, LLC
26180 Curtiss Wright Parkway
Richmond Heights, Ohio 44143

certifies that the change in the type design for the following product with the limitations and conditions, therefore as specified hereon meets the airworthiness requirements of Part 25* of the Federal Aviation Regulations. *Certification basis is set forth on TCDS A16SW and continuation sheets 3 and 4.

Original Product Type Certificate Number: A16SW

Make: Hawker Beechcraft

Model: 400A

Description of Type Design Change: Replacement of Pratt & Whitney JT15D-5 series engine with Williams International FJ44-3AP engine (TCDS E3GL) with Full Authority Digital Engine Control (FADEC) and associated systems, in accordance with Nextant Aerospace Master Drawing List No.: MDL-NXT-001, Revision IR dated 8/22/2011, or later FAA approved revisions and FAA Approved Airplane Flight Manual Supplement, NXT1-AFMS, Rev IR, dated September 15, 2011 or later FAA approved revisions. Nextant Aerospace, LLC Instructions for Continued Airworthiness document NXT1-ICA, Revision IR, or later FAA accepted revisions must be provided with this STC.

Limitations and Conditions: The installation should not be incorporated in any aircraft unless it is determined that the interrelationship between this installation and any previously approved configuration will not introduce any adverse effect upon the airworthiness of that aircraft. The approval of this modification applies to the above-noted airplane models only. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

(CONTINUED on page 3)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration

Date of application: March 24, 2008

Date reissued:

Date of issuance: September 27, 2011

Date amended:



[Handwritten Signature]

(Signature)

ACTING

Manager, Propulsion Branch
Los Angeles Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

SDNY_GM_02758331
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245253

EFTA01261913

INSTRUCTIONS: The transfer endorsement below may be used to notify the appropriate FAA Regional Office of the transfer of this Supplemental Type Certificate.

The FAA will reissue the certificate in the name of the transferee and forward it to him.

TRANSFER ENDORSEMENT

Transfer the ownership of the Supplemental Type Certificate Number _____

to *(Name of transferee)* _____

(Address of transfer) _____
(Number and street)

(City, State, and Zip code)

from *(Name of grantor) (Print or type)* _____

(Address of grantor) _____
(Number and street)

(City, State, and Zip code)

Extent of Authority (if licensing agreement): _____

Date of Transfer: _____

Signature of grantor *(In ink)*: _____

SDNY_GM_02758333

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245255

EFTA01261915

United States Of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number ST02371LA

Limitations and Conditions (continued)

- Nextant Aerospace, LLC, "Installation of Rockwell Collins Pro Line 21," Amended STC ST10959SC, dated September 26, 2011 or later FAA approved revisions is required as part of this alteration.
- The new Vmca and Vmcg values are 94 KIAS and 93 KIAS respectively.
- The new engine minimum rated thrust is 3,031 lb at sea level, flat rated to an ambient temperature of 77°F.
- Approved engine fuels are:

<u>Grade</u>	<u>Specification</u>
Jet A	ASTM-D 1655
Jet A-1	ASTM-D 1655

Certification Basis

The certification basis for this modification is as follows:

The original certification basis for the Hawker Beechcraft Model 400A airplanes identified on TCDS A16SW, Revision 26, dated March 17, 2010:

Part 25 of the Federal Aviation Regulations effective February 1, 1965, as amended by 25-1 through 25-40, plus §§ 25.1335, 25.1351(d), 25.1353(c)(5), and 25.1447 of Amendment 25-41; §§ 25.29, 25.255, and 25.1353(c)(6) of Amendment 25-42; and §§ 25.361(b) and 25.1329(h) of Amendment 25-46. Part 36 of the Federal Aviation Regulations effective December 1, 1969, as amended by 36-1 through 36-17; SFAR 27 effective February 1, 1974, as amended by 27-1 through 27-5; and Special Conditions No. 25-ANM-32 dated February 22, 1990 (High Altitude Operation at 45,000 feet), and Special Conditions No. 25-ANM-33 dated June 18, 1990 (Lightning and Radio Frequency Energy Protection). (See NOTE 12)

Equivalent Safety Items:

- (1) Out-of-trim characteristics § 25.255
 - (2) Pilot compartment view § 25.773(b)(2)
 - (3) Passenger compartment door § 25.813(e)
 - (4) Emergency exit marking §§ 25.811(d)(1) and 25.811(d)(2)
- Application for amended Type Certificate dated February 18, 1988.

Regulation at a later amendments for components and areas affected by the change (Ref. FAA approved Compliance Check List):

Subpart A - General: 25.2(25-99)

Subpart B - Flight: 25.207(25-121), 25.255(25-42)

Subpart C - Structure: 25.571 (25-96)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

• • • •

SDNY_GM_02758335

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245257

EFTA01261917

United States Of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number ST02371LA

Certification Basis (continued)

Subpart D – Design and Construction: 25.603(25-46), 25.605(25-46), 25.611(25-123), 25.613(25-112), 25.625(25-72), 25.832(25-72), 25.863(25-46), 25.869(25-123),

Subpart E – Powerplant: 25.901(25-46), 25.903(25-100), 25.951(25-73), 25.961(25-57), 25.981(25-125), 25.994(25-57), 25.997(25-57), 25.1013(25-72), 25.1019(25-57), 25.1021(25-57), 25.1043(25-42),

25.1045(25-57), 25.1091(25-100), 25.1093(25-72), 25.1103(25-46), 25.1141(25-115), 25.1143(25-57), 25.1163(25-57), 25.1165(25-72), 25.1181(25-115), 25.1183(25-101), 25.1185(25-94), 25.1189(25-57), 25.1195(25-46), 25.1203(25-123), 25.1207(25-46).

Subpart F – Equipment: 25.1301(25-123), 25.1305(25-115), 25.1307(25-72), 25.1309(25-123), 25.1316(25-80), 25.1317(25-122), 25.1321(25-41), 25.1331(25-41), 25.1351(25-72), 25.1353(25-123), 25.1357(25-123), 25.1381(25-72), 25.1431(25-113), 25.1438(25-41),

Subpart G – Operating Limitations and Information: 25.1501(25-42), 25.1521(25-72), 25.1527(25-105), 25.1529(25-54), 25.1543(25-72), 25.1549(25-40), 25.1551(25-72), 25.1557(25-72), 25.1581(25-72), 25.1583(25-105).

Subpart H – Electrical Wiring Interconnection Systems (EWIS): 25.1701(25-123), 25.1703(25-123), 25.1705(25-123), 25.1707(25-123), 25.1709(25-123), 25.1711(25-123), 25.1713(25-123), 25.1715(25-123), 25.1717(25-123), 25.1719(25-123), 25.1721(25-123), 25.1723(25-123), 25.1725(25-123), 25.1727(25-123), 25.1729(25-123), 25.1731(25-123), 25.1733(25-123).

14 CFR Part 34 (34-3): 34.11

14 CFR Part 36 (36-28): 36.1--36.7, 36.101--36.105, 36.1501, 36.1581.

The following Nextant Aerospace Continued Airworthiness Supplements are applicable to airplanes modified in accordance with this STC:

- Master Minimum Equipment List document No. MMEL Rev. 8, dated 02/21/2011 or later FAA approved revisions.
- EWIS (Electrical Wiring Interconnection Systems) Inspection Procedure/Report No. NXT1-EA-007 Rev A dated 8/24/11 or later FAA approved revisions.

-END-

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2 (10-68)

Page 4 of 4

This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758336

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245258

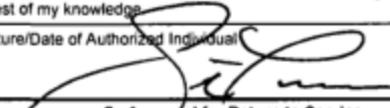
EFTA01261918

SDNY_GM_02758337

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245259

EFTA01261919

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
	For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)				
1. Aircraft	Nationality and Registration Mark N787TA		Serial No. RK-260	
	Make Hawker Beechcraft Corporation		Model 400A	Series
2. Owner	Name (As shown on registration certificate) Flight Options, LLC		Address (As shown on registration certificate) 26180 Curtiss Wright PKWY	
			City Cleveland	State Ohio
		Zip 44143-1453	Country USA	
3. For FAA Use Only				
4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	
			Manufacturer	
6. Conformity Statement				
A. Agency's Name and Address		B. Kind of Agency		
Name Constant Aviation Address 5211 Secondary Road City Cleveland State Ohio Zip 44135 Country United States		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Manufacturer <input type="checkbox"/> Foreign Certificated Mechanic <input type="checkbox"/> Certificate No. <input checked="" type="checkbox"/> Certificated Repair Station CRS#WC7R346J <input type="checkbox"/> Certificated Maintenance Organization		
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.				
Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>		Signature/Date of Authorized Individual  12/26/2012		
7. Approval for Return to Service				
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED				
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. CRS#WC7R346J		Signature/Date of Authorized Individual  26 DEC 12		

FAA Form 337 (10-06)

SDNY_GM_02758338

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245260

EFTA01261920

SDNY_GM_02758339

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245261

EFTA01261921

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

27DEC12

Nationality and Registration Mark

Date

ACCOMPLISHED INTERIOR MODIFICATION AND 3 PLACE DIVAN INSTALLATION

THE FOLLOWING AIRCRAFT INTERIOR ITEMS HAVE BEEN MODIFIED:

PILOT SEAT PART NUMBER TABH100PM0001 SERIAL NUMBER 59, CO-PILOT SEAT PART NUMBER TABH100CM0001 SERIAL NUMBER 56, RIGHT HAND PASSENGER SEAT #1 PART NUMBER 2524.014-530B SERIAL NUMBER 991215-20, RIGHT HAND PASSENGER SEAT #2 PART NUMBER 2524.014-530B SERIAL NUMBER 991215-23, LEFT HAND PASSENGER SEAT #1 PART NUMBER 2524.014-530B SERIAL NUMBER 991215-3, LEFT HAND PASSENGER SEAT #2 PART NUMBER 2524.014-530B SERIAL NUMBER 000925, LAV SEAT CUSHIONS, CONSOLE ASSEMBLY (DADOS), ALL BAGGAGE PANEL MATERIAL, CARPET, AIR STAIR MATERIAL, LEFT HAND GALLEY, RIGHT HAND GALLEY, LEFT HAND AND RIGHT HAND CARD TABLE, LEFT HAND AND RIGHT HAND SIDE LEDGE, RIGHT HAND ARMREST, DIVAN ASSEMBLY, HEADLINER MATERIAL, WINDOW LINE MATERIAL, LOWER SIDE WALL MATERIAL, UPPER AND LOWER ENTRY DOOR MATERIAL, AFT LAV BULKHEAD MATERIAL, AFT BULKHEAD MIRROR, CONTROL COLUMN YOKES MATERIAL, GLARE SHIELD MATERIAL, DIVAN SEAT CUSHIONS, CABINETRY, CONTROL YOKE BOOTS MATERIAL AND RUDDER PEDAL BOOTS MATERIAL.

THE FOLLOWING MANUALS AND APPROVED DATA HAVE BEEN UTILIZED FOR BASIS OF MODIFICATION:

- o HAWKER BEECHCRAFT AMM AND SRM. NEXTANT AER EO 029-11 EXTERIOR BAGGAGE MODIFICATION AS APPROVED ON FAA FORM 8110-3 BY DERT 410115-CE, PAUL HEDDING DATED 28NOV12. DELTA G DESIGN DRAWINGS. LEFT HAND GALLEY ASSEMBLY DGD-CA01-101 REV B DATED 28SEP12, LEFT HAND GALLEY INSTALLATION DGD-CA01-501 REV IR DATED 31JUN12, RIGHT HAND GALLEY INSTALLATION DGD-CA02-501 REV IR DATED 13JUN12, SIDE LEDGE INSTALLATION DGD-CA04-501 REV IR DATED 13JUN12, RIGHT HAND GALLEY ASSEMBLY DGD-CA02-101 REV. A DATED 08AUG12, CABIN HEADLINER DGD-CA06-101 EO. A-4 DATED 10DEC12, CABIN HEADLINER DGD-CA06-101 REV IR, DATED 13JUN12, WINDOW LINER DGD-CA07-101 EO A-4 DATED 10DEC12, WINDOW LINER DGD-CA07-101 REV IR DATED 13JUN12, LAVATORY CABINET DGD-CA10-101 REV IR, DATED 12DEC12, LAVATORY HEADLINER DGD-CA11-101 REV IR, DATED 17DEC12, SIDE LEDGE ASSEMBLY DGD-CA04-101 REV IR DATED 3JUN12, SIDE LEDGE ASSEMBLY DGD-CA04-101 EO: A-5 DATED 20DEC12, AS APPROVED BY DER DONALD GILLESPIE (DERT-750135-SW) ON FAA FORM 8110-3 DATED 19DEC12, CARD TABLE INSTALLATION DGD-CA13-501 REV IR DATED 21DEC12, STRUCTURAL SUBSTANTIATION DGD-CA01-SS-05 REV IR AS APPROVED BY DER DONALD GILLESPIE (DERT-750135-SW) ON FAA FORM 8110-3 DATED 21DEC12, ACES FLAMMABILITY SUBSTANTIATION REPORT A121812-01FSR REV. IR AS APPROVED BY DER KEVIN CAMPBELL (DERT-230220-CE) ON FAA FORM 8110-3 DATED 18DEC12, ACES FLAMMABILITY SUBSTANTIATION REPORT A102612-01FSR REV. IR AS APPROVED BY DER KEVIN CAMPBELL (DERT-230220-CE) ON FAA FORM 8110-3 DATED 26OCT12, ACES FLAMMABILITY TEST PLAN #A120312-01FP REV A AS APPROVED BY DER KEVIN CAMPBELL (DERT-230220-CE) ON FAA FORM 8110-3 DATED 17DEC12, ACES FLAMMABILITY TEST RESULTS REPORT #A122112-02FR REV IR AS APPROVED BY DER KEVIN CAMPBELL (DERT-230220-CE) ON FAA FORM 8110-3 DATED 21DEC12.

THREE (3) PLACE DIVAN INSTALLATION:

INSTALLED THREE (3) PLACE DIVAN PART NUMBER: 32-0396(A60P)T123.k.02, SERIAL NUMBER: 003651 IN ACCORDANCE WITH AVIATION FABRICATORS INC. STC # ST01572WI, COMPLETE UNIT P/N 400-3.

INSTRUCTIONS FOR CONTINUED AIRWORTHINESS:

THE REMOVAL OF THE INTERIOR IS ON THE CONDITION OF FAILURE. THERE ARE NO ADDITIONAL INSPECTIONS REQUIRED. CONTINUE TO INSPECT IAW MANUFACTURERS AMM, SRM, OR CMM AS APPLICABLE

AIRCRAFT EQUIPMENT LIST AND WEIGHT AND BALANCE WERE UPDATED BY THE CUSTOMER

Additional Sheets Are Attached

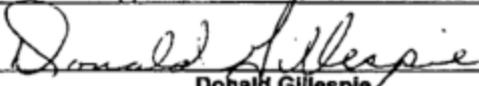
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758341

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245263

EFTA01261923

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS		1. DATE December 19, 2012	
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Beechjet	3. MODEL NO. 400A	4. TYPE (Airplane, Radio, Helicopter, etc.) AIRPLANE	5. NAME OF APPLICANT Constant Aviation Cleveland, Ohio
LIST OF DATA			
6. IDENTIFICATION		7. TITLE	
Document No. DGD-CA10-101 REV: IR Dated: 12/12/12 DGD-CA11-101 REV: IR Dated: 12/17/12		LAVATORY CABINET MOD, LH LAVATORY HEADLINER MOD, LH ----- End of Data ----- NOTES: 1. This approval is for Beechjet model 400A (Nextant 400XT variant) S/N RK-260 only. 2. This approval is for the structural design aspects only on the above listed data and does not constitute an installation approval. 3. Approval is only for regulations specified in the "Applicable Requirements" block. Compliance with additional regulations not listed herein may be required. This form does not constitute FAA approval of all engineering design data for the entire alteration. Additional 8110-3 approvals may be necessary for Damage Tolerance, Flammability, Mechanical Systems, Electrical, Weight & Balance, as well as other disciplines. 4. For certification basis, refer to TC data sheet No. A16SW 5. Structural Approvals are provided separately.	
8. PURPOSE OF DATA To provide type data for FAA approval of structure in support of a return to service on S/N RK-260. This approval is for design data only on a major alteration and is not an installation approval.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Part 25.601 Amdt 25-0, 25.603(a)(b), Amdt 25-38, 25.605 Amdt 25-0, 25.609 Amdt 25-0, 25.611, Amdt 25-23, 25.613 (a)(b)(c) Amdt 25-0, 25.619 Amdt 25-23, 25.625 Amdt 25-23, 25.785 (a)(b)(c)(d)(e)(f) Amdt 25-32, 25.789 Amdt 25-32			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed. <input type="checkbox"/> Recommend approval of these data I (We) Therefore <input checked="" type="checkbox"/> Approve these data			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)  Donald Gillespie		12. DESIGNATION NUMBER(S) DERT-750135-SW	13. CLASSIFICATION(S) STRUCTURAL

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758342

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245264

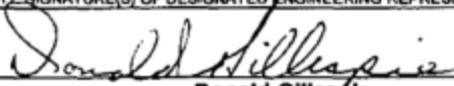
EFTA01261924

SDNY_GM_02758343

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245265

EFTA01261925

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS		1. DATE December 19, 2012	
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Beechjet	3. MODEL NO. 400A	4. TYPE (Airplane, Radio, Helicopter, etc.) AIRPLANE	5. NAME OF APPLICANT Constant Aviation Cleveland, Ohio
LIST OF DATA			
6. IDENTIFICATION		7. TITLE	
Document No. DGD-CA06-101 REV: IR Dated: 6/13/12		CABIN HEADLINER MOD	
DGD-CA06-101 EO:A-4 Dated: 12/10/12		CABIN HEADLINER MOD	
DGD-CA07-101 REV: IR Dated: 6/13/12		WINDOW LINER MOD	
DGD-CA07-101 EO:A-4 Dated: 12/10/12		WINDOW LINER MOD	
----- End of Data -----			
NOTES: <ol style="list-style-type: none"> This approval is for Beechjet model 400A (Nextant 400XT variant) S/N RK-260 only. This approval is for the structural design aspects only on the above listed data and does not constitute an installation approval. Approval is only for regulations specified in the "Applicable Requirements" block. Compliance with additional regulations not listed herein may be required. This form does not constitute FAA approval of all engineering design data for the entire alteration. Additional 8110-3 approvals may be necessary for Damage Tolerance, Flammability, Mechanical Systems, Electrical, Weight & Balance, as well as other disciplines. For certification basis, refer to TC data sheet No. A16SW Structural Approvals are provided separately. 			
8. PURPOSE OF DATA To provide type data for FAA approval of structure in support of a return to service on S/N RK-260. This approval is for design data only on a major alteration and is not an installation approval.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Part 25.601 Amdt 25-0, 25.603(a)(b), Amdt 25-38, 25.605 Amdt 25-0, 25.609 Amdt 25-0, 25.611, Amdt 25-23, 25.613 (a)(b)(c) Amdt 25-0, 25.619 Amdt 25-23, 25.625 Amdt 25-23, 25.785 (a)(b)(c)(d)(e)(f) Amdt 25-32, 25.789 Amdt 25-32			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.			
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data			
I (We) Therefore			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)
 Donald Gillespie		DERT-750135-SW	STRUCTURAL

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758344

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245266

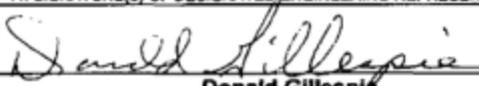
EFTA01261926

SDNY_GM_02758345

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245267

EFTA01261927

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE December 19, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Beechjet	3. MODEL NO. 400A	4. TYPE (Airplane, Radio, Helicopter, etc.) AIRPLANE	5. NAME OF APPLICANT Constant Aviation Cleveland, Ohio
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Document No. DGD-CA01-101 REV: B Dated: 9/28/12 DGD-CA01-501 REV: IR Dated: 6/13/12 DGD-CA02-101 REV: A Dated: 8/08/12 DGD-CA02-501 REV: IR Dated: 6/13/12	LH GALLEY ASSY LH GALLEY INSTL RH GALLEY ASSY RH GALLEY INSTL ----- End of Data ----- NOTES: 1. This approval is for Beechjet model 400A (Nextant 400XT variant) S/N RK-260 only. 2. This approval is for the structural design aspects only on the above listed data and does not constitute an installation approval. 3. Approval is only for regulations specified in the "Applicable Requirements" block. Compliance with additional regulations not listed herein may be required. This form does not constitute FAA approval of all engineering design data for the entire alteration. Additional 8110-3 approvals may be necessary for Damage Tolerance, Flammability, Mechanical Systems, Electrical, Weight & Balance, as well as other disciplines. 4. For certification basis, refer to TC data sheet No. A16SW 5. Structural Approvals are provided separately.		
8. PURPOSE OF DATA To provide type data for FAA approval of structure in support of a return to service on S/N RK-260. This approval is for design data only on a major alteration and is not an installation approval.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Part 25.601 Amdt 25-0, 25.603(a)(b), Amdt 25-38, 25.605 Amdt 25-0, 25.609 Amdt 25-0, 25.611, Amdt 25-23, 25.613 (a)(b)(c) Amdt 25-0, 25.619 Amdt 25-23, 25.625 Amdt 25-23, 25.785 (a)(b)(c)(d)(e)(f) Amdt 25-32, 25.789 Amdt 25-32			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed. <input type="checkbox"/> Recommend approval of these data I (We) Therefore <input checked="" type="checkbox"/> Approve these data			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)  Donald Gillespie	12. DESIGNATION NUMBERS(S) DERT-750135-SW	13. CLASSIFICATION(S) STRUCTURAL	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758346

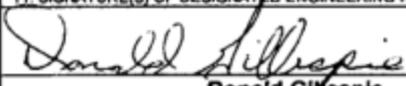
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758347

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245269

EFTA01261929

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE December 19, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Beechjet	3. MODEL NO. 400A	4. TYPE (Airplane, Radio, Helicopter, etc.) AIRPLANE	5. NAME OF APPLICANT Constant Aviation Cleveland, Ohio
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Document No. DGD-CA04-101 REV: IR Dated: 6/13/12	SIDE LEDGE ASSY		
DGD-CA04-101 EO:A-4 Dated: 12/11/12	SIDE LEDGE ASSY		
DGD-CA04-501 REV: IR Dated: 6/13/12	SIDE LEDGE INSTL		
----- End of Data -----			
NOTES:			
1. This approval is for Beechjet model 400A (Nextant 400XT variant) S/N RK-260 only.			
2. This approval is for the structural design aspects only on the above listed data and does not constitute an installation approval.			
3. Approval is only for regulations specified in the "Applicable Requirements" block. Compliance with additional regulations not listed herein may be required. This form does not constitute FAA approval of all engineering design data for the entire alteration. Additional 8110-3 approvals may be necessary for Damage Tolerance, Flammability, Mechanical Systems, Electrical, Weight & Balance, as well as other disciplines.			
4. For certification basis, refer to TC data sheet No. A16SW			
5. Structural Approvals are provided separately.			
8. PURPOSE OF DATA To provide type data for FAA approval of structure in support of a return to service on S/N RK-260. This approval is for design data only on a major alteration and is not an installation approval.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Part 25.601 Amdt 25-0, 25.603(a)(b), Amdt 25-38, 25.605 Amdt 25-0, 25.609 Amdt 25-0, 25.611, Amdt 25-23, 25.613 (a)(b)(c) Amdt 25-0, 25.619 Amdt 25-23, 25.625 Amdt 25-23, 25.785 (a)(b)(c)(d)(e)(f) Amdt 25-32, 25.789 Amdt 25-32			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed. <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)	
 Donald Gillespie	DERT-750135-SW	STRUCTURAL	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758348

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245270

EFTA01261930

SDNY_GM_02758349

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245271

EFTA01261931



(660) 885-8317
(660) 885-3743 Fax
www.avfab.com
e-mail: sales@avfab.com

AUTHORIZATION OF STC USE:

DATE: 12-17-12

TO: Nextant Aerospace LLC
26180 Curtiss Wright Parkway
Cleveland, OH 44143

Reference your order number: 003793

Authorization is hereby given for one time usage of Supplemental Type Certificate

Number SA001572WI install Aviation Fabricators' Beechcraft

3 Place Divan w/ Tabs

P/N 32-0396(A60P)T123.K-02 S/N 003651

This kit is approved for installation on Beechcraft 400A
(aircraft make) (aircraft model)

RK-260
(serial number of aircraft)

Aviation Fabricators
805 North Fourth Street
Clinton, Missouri, 64735


S. Todd Pogue
Inspector

(Note: This must be original type on colored Aviation Fabricators letterhead)

Aviation Fabricators • Central Airmotive
805 North Fourth Street, Clinton, Missouri 64735

SDNY_GM_02758350

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245272

EFTA01261932

SDNY_GM_02758351

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245273

EFTA01261933

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE October 9, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE	3. MODEL NO.	4. TYPE (Aircraft, Engine, Propeller, etc.)	5. NAME OF APPLICANT	
Hawker Beechcraft Corporation	400A Beechjet	Aircraft	Constant Aviation, LLC	
LIST OF DATA				
6. IDENTIFICATION		7. TITLE		
Report: A100912-01FP REV IR ===== END =====		FLAMMABILITY TEST PLAN – YOKE COLUMNS – HAWKER BEECHCRAFT CORPORATION MODEL 400A BEECHJET SN RK-327 Note: 1. Approval of test results is required. ===== END =====		
PURPOSE OF DATA Document the flammability test requirements for the installation of new coverings to the yoke columns on Hawker Beechcraft Corporation Model 400A Beechjet SN RK-327. Test plan approval only, in support of a major alteration.				
8. APPLICABLE REQUIREMENTS (List specific sections) 14CFR Part 25.853(a) at Amendment 25-116				
10. CERTIFICATION -Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>None</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)	
Aviation Consulting & Engineering Solutions, Inc. K.C. Campbell <i>K.C. Campbell</i>		DERT-230220-CE	Structures FAR 23 & 25	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758352

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245274

EFTA01261934

SDNY_GM_02758353

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245275

EFTA01261935

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE October 10, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A Beechjet	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Constant Aviation, LLC	
LIST OF DATA				
6. IDENTIFICATION		7. TITLE		
Report: A101012-01FR REV IR		FLAMMABILITY TEST RESULTS – YOKE COLUMNS- HAWKER BEECHCRAFT CORPORATION MODEL 400A BEECHJET SN RK-327		
Notes:		<p>1. The flammability aspects only of the above listed data are approved herein. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements."</p> <p>2. This form does not constitute FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration/repair. Compliance with additional regulations may be required.</p>		
=====		=====		
END		END		
=====		=====		
PURPOSE OF DATA Document the flammability test results for the installation of new coverings to the yoke columns on Hawker Beechcraft Corporation Model 400A Beechjet SN RK-327 in support of a major alteration.				
8. APPLICABLE REQUIREMENTS (List specific sections) 14CFR Part 25.853(a) at Amendment 25-116				
10. CERTIFICATION -Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>None</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed.				
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)	
Aviation Consulting & Engineering Solutions, Inc. K.C. Campbell <i>K.C. Campbell</i>		DETT-230220-CE	Structures FAR 23 & 25	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758354

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245276

EFTA01261936

SDNY_GM_02758355

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245277

EFTA01261937

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE December 17, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE	3. MODEL NO.	4. TYPE (Aircraft, Engine, Propeller, etc.)	5. NAME OF APPLICANT	
Hawker Beechcraft Corporation	400A Beechjet	Aircraft	Constant Aviation, LLC	
LIST OF DATA				
6. IDENTIFICATION	7. TITLE			
Report: A120312-01FP REV A =====	FLAMMABILITY AND FIREBLOCK TEST PLAN – INTERIOR REFURBISHMENT – HAWKER BEECHCRAFT CORPORATION MODEL 400A BEECHJET SN RK-260			
END	Note: 1. Approval of test results is required. =====			
=====	END			
PURPOSE OF DATA Document the flammability and fireblock test requirements for the interior refurbishment on Hawker Beechcraft Corporation Model 400A Beechjet SN RK-260. Test plan approval only, in support of a major alteration.				
8. APPLICABLE REQUIREMENTS (List specific sections) 14CFR Parts 25.853(a)(c) at Amendment 25-116				
10. CERTIFICATION -Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>None</u> , have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)		
Aviation Consulting & Engineering Solutions, Inc. K.C. Campbell <i>Kevin Campbell</i>	DERT-230220-CE	Structures FAR 23 & 25		

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758356

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245278

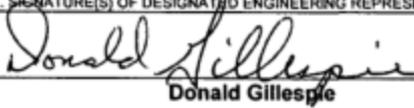
EFTA01261938

SDNY_GM_02758357

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245279

EFTA01261939

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			1. DATE December 21, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Beechjet	3. MODEL NO. 400A	4. TYPE (Airplane, Rado, Helicopter, etc.) AIRPLANE	5. NAME OF APPLICANT Constant Aviation Cleveland, Ohio
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
Document No. DGD-CA01-SS-05 REV: IR Dated: 12/21/12	SUBSTANTIATION ANALYSIS OF THE INTERIOR ON AN HAWKER BEECHCRAFT 400A AIRCRAFT ----- End of Data ----- NOTES: 1. This approval is for Beechjet model 400A (Nextant 400XT variant) S/N RK-260 only. 2. This approval is for the structural design aspects only on the above listed data and does not constitute an installation approval. 3. Approval is only for regulations specified in the "Applicable Requirements" block. Compliance with additional regulations not listed herein may be required. This form does not constitute FAA approval of all engineering design data for the entire alteration. Additional 8110-3 approvals may be necessary for Damage Tolerance, Flammability, Mechanical Systems, Electrical, Weight & Balance, as well as other disciplines. 4. For certification basis, refer to TC data sheet No. A16SW		
8. PURPOSE OF DATA Structural substantiation in support of FAA Form 337 for aircraft S/N RK-260. This approval is for design data only on a major alteration and is not an installation approval.			
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Part 25.301[Amdt. 25-23], .303[Amdt. 25-23], .305(a)(b)[Amdt. 25-23], .307[Amdt. 25-23], .341[Amdt. 25-0], .561[Amdt. 25-23], .601[Amdt. 25-0], .603(a)(b)[Amdt. 25-38], .605[Amdt. 25-0], .609[Amdt. 25-0], .611[Amdt. 25-23], .613(a)(b)(c)[Amdt. 25-0], .625 [Amdt. 25-23], .785(a)(b)(c)(d)(e)(f)[Amdt. 25-32], .787(a)[Amdt. 25-32], .789[Amdt. 25-32]			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed. <input type="checkbox"/> Recommend approval of these data I (We) Therefore <input checked="" type="checkbox"/> Approve these data			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBER(S)	13. CLASSIFICATION(S)	
 Donald Gillespie	DERT-750135-SW	STRUCTURAL	

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758358

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758359

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245281

EFTA01261941

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE December 21, 2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A Beechjet	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Constant Aviation, LLC	
LIST OF DATA				
6. IDENTIFICATION		7. TITLE		
Report: A122112-02FR Rev IR		FLAMMABILITY AND FIREBLOCK TEST RESULTS – INTERIOR REFURBISHMENT – HAWKER BEECHCRAFT CORPORATION MODEL 400A BEECHJET SN RK-260 Notes: 1. The flammability aspects only of the above listed data are approved herein. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements." 2. This form does not constitute FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration/repair. Compliance with additional regulations may be required.		
=====		=====		
END		END		
=====		=====		
PURPOSE OF DATA Document the flammability and fireblock test results for interior refurbishment on Hawker Beechcraft Corporation Model 400A Beechjet RK-260 in support of a major alteration.				
8. APPLICABLE REQUIREMENTS (List specific sections) 14CFR Parts 25.853(a)(c) at Amendment 25-116				
10. CERTIFICATION -Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>None</u> , have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed. <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S) Aviation Consulting & Engineering Solutions, Inc. K.C. Campbell <i>K.C. Campbell</i>		12. DESIGNATION NUMBER(S) DERT-230220-CE		13. CLASSIFICATION(S) Structures FAR 23 & 25

FAA Form 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION

SDNY_GM_02758360

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245282

EFTA01261942

SDNY_GM_02758361
SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245283

EFTA01261943

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION			1. DATE November 27, 2012
STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS			
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
2. MAKE Hawker Beechcraft Corporation	3. MODEL NO. 400A	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Nextant Aerospace LLC Cleveland, Ohio
LIST OF DATA			
6. IDENTIFICATION	7. TITLE		
EO 1018 Rev. A December 26, 2012	Pedestal Cover Extension Mod.		
	-----END-----		
	Notes:		
	<ol style="list-style-type: none"> 1) The structural aspects only of the above listed data are approved herein. This approval is only for the engineering data. It indicates the data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements." 2) This form constitutes FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration. 		
8. PURPOSE OF DATA: Engineering data in support of major alteration to Hawker Beechcraft 400A, S/N RK-260.			
9. APPLICABLE REQUIREMENTS (List specific sections) CFR Title 14 Part 25.301[25-23], 25.303[25-23], 25.305(a)(b)(c)[25-23], 25.307(a)[25-23], 25.562(c)[25-23], 25.601[25-00], 25.603(a)(b)(c)[25-38], 25.605(a)[25-00], 25.607[25-23], 25.609(a)(b)[25-00], 25.611[25-23], 25.613(a)(b)(c)[25-00], 25.619[25-23], 25.625(a)(b)(c)[25-23], and 25.789[25-32].			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> , have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards Listed.			
<input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data I (We) Therefore			
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)	
 Paul Hedding	DERT-410115-CE	Structures	

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

SDNY_GM_02758363

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245285

EFTA01261945

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)				Form Approved OMB No. 2120-0020 11/30/2007		Electronic Tracking Number	
				For FAA Use Only			
INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a)).							
1. Aircraft	Nationality and Registration Mark N787TA			Serial No. RK-260			
	Make Raytheon Aircraft Company			Model 400A		Series	
2. Owner	Name (As shown on registration certificate) Flight Options LLC			Address (As shown on registration certificate) 26180 Curtiss Wright Pkwy Richmond Heights, Ohio 44143-1453			
3. For FAA Use Only							
4. Type		5. Unit Identification					
Repair	Alteration	Unit	Make	Model		Serial No.	
	✓	AIRFRAME	-----	(As described in Item 1 above)		-----	
		POWERPLANT					
		PROPELLER					
		APPLIANCE	Type				
			Manufacturer				
6. Conformity Statement							
A. Agency's Name and Address				B. Kind of Agency			
Flying Colours Corp. R.R. #5, Peterborough Airport Peterborough, Ontario K9J 6X6 CANADA				U.S. Certificated Mechanic		Manufacturer	
				✓ Foreign Certificated Mechanic		C. Certificate No.	
				Certificated Repair Station		Canadian AMO 45-08	
				Certificated Maintenance Organization			
D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.							
Extended range fuel per 14 CFR Part 43 App. B (If yes, place 1 copy on board)			Signature / Date of Authorized Individual December 13, 2012			  Tim Pickett	
7. Approval for Return to Service							
Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED							
B Y	FAA Fit. Standards Inspector	Manufacturer	Maintenance Organization	✓ Persons Approved by Canadian Department of Transportation			
	FAA Designee	Repair Station	Inspection Authorization	Other (Specify)			
Certificate or Designation No. Canadian AMO 45-08			Signature / Date of Authorized Individual December 13, 2012			  Tim Pickett	

FAA Form 337 (10-06)

SDNY_GM_02758364

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245286

EFTA01261946

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N787TA

December 13, 2012

Nationality and Registration Mark

Date

USE OF JETGLO PAINT ON WINDSHIELD / WINDOW RETAINERS

- 1) U.S. Technical Engineering Order EO 6546-001 'Protective Material Substitution - Use of JetGlo Paint on Outer Windshield and Side Window Retainer' applied.
- 2) This alteration is approved under FAA Form 8110-3 issued by Robert M. Halvorson, DERT-605221-NM, dated October 12, 2012.
- 3) There are no additional continuing airworthiness inspections required as a result of this work. The inspection criteria and frequency of these areas is covered in the existing Beechjet 400A inspection program.
- 4) This installation has no effect on aircraft weight and balance.
- 5) Details of work performed are maintained on file at Flying Colours under work order 125842.

***** END *****

No Additional Sheets Are Attached

SDNY_GM_02758365

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245287

EFTA01261947

RW
3

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)		Form Approved OMB No. 2120-0020
U.S. Department of Transportation Federal Aviation Administration		For FAA Use Only Office Identification 62757
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)		
1. Aircraft	Make BAYTHEON Serial No. BK-260	Model BEECHJET 400A Nationality and Registration Mark N787TA
2. Owner	Name (As shown on registration certificate) AIR GATSLAINE INC. ET AL	Address (As shown on registration certificate) FLIGHT OPTIONS LLC 20180 CURTIS WRIGHT PARKWAY RICHMOND HEIGHTS, OH 44143
3. For FAA Use Only		
4. Unit Identification		5. Type
Unit	Make	Model
(As described in item 1 above)		
AIRFRAME		
POWERPLANT		
PROPELLER		
APPLIANCE	Type	
	Manufacturer	
6. Conformity Statement		
A. Agency's Name and Address	B. Kind of Agency	C. Certificate No.
BAYTHEON - RICHARD ERWIN 20180 CURTIS WRIGHT PARKWAY RICHMOND HEIGHTS, OH 44143	<input checked="" type="checkbox"/> U.S. Certified Mechanic <input type="checkbox"/> Foreign Certified Mechanic <input type="checkbox"/> Certified Repair Station <input type="checkbox"/> Manufacturer	AMP 304146
D. I certify that the repair and/or alteration made to the unit identified in item 4 above and described on the request or attachment hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.		
Date 11-10-05	Signature of Authorized Individual <i>[Signature]</i>	
7. Approval for Return to Service		
Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED		
BY	Manufacturer	Inspection Authorization
<input type="checkbox"/> FAA Part Standards Inspector <input type="checkbox"/> FAA Designee	<input checked="" type="checkbox"/> Repair Station	Other (Specify)
Date of Approval or Rejection 11/10/05	Certificate of Designation No. FA2125820	Signature of Authorized Individual <i>[Signature]</i> DOUGLAS M. ALBERT

RECEIVED
NOV 14 2005
CIE-GL25

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

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[Faint, illegible text, possibly a list or table]

[Faint, illegible text, possibly a signature block or closing]

SDNY_GM_02758367

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

B. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

REMOVED MAIN BATTERY AND INSTALLED
SECURAPLANE BATTERY P/N 9750W0370
IN ACCORDANCE WITH STC ST01794LA.
WEIGHT AND BALANCE INFORMATION WAS
UPDATED IN THE AIRCRAFT RECORD.

Additional Sheets Are Attached

U.S. GPO: 1990-0-603-171

SDNY_GM_02758369

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020

For FAA Use Only

Office Identification
FEA-23 ME

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Make Raytheon Aircraft Company	Model 400A
	Serial No. RK-260	Nationality and Registration Mark N787TA
2. Owner	Name (As shown on registration certificate) Grand/Sakwa Transportation LLC c/o Flight Options LLC	Address (As shown on registration certificate) 26180 Curtiss-Wright Parkway Richmond Heights, OH 44143-1453

3. For FAA Use Only

4. Unit Identification

Unit	Make	Model	Serial No.	5. Type	
				Repair	Alteration
AIRFRAME	----- (As described in Item 1 above) -----				✓
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement

A. Agency's Name and Address Rapid Aircraft Repair, Inc. R.R. #5, Peterborough Airport Peterborough, Ontario K9J 6X6 CANADA	B. Kind of Agency <input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station - Canadian <input type="checkbox"/> Manufacturer	C. Certificate No. AMO 73-92
--	--	--

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date June 1, 2004	Signature of Authorized Individual
----------------------	--

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit Standards Inspector	Manufacturer	Inspection Authorization	Other (Specify)
	FAA Designee	Repair Station	<input checked="" type="checkbox"/> Person Approved by Transport Canada Airworthiness Group	
Date of Approval or Rejection June 1, 2004		Certificate or Designation No. AMO 73-92	Signature of Authorized Individual 	

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

GRAPHIC FILM 220 APPLICATION

- 1) Applied 3M Graphic Film 220 and 7-Year Metallic Series Film to exterior of aircraft per Flight Options Engineering Directive EDU91-1 Rev B dated 3/19/04 and EDU91-2 Rev A dated 3/19/04; and FAA 8110-3 form issued by William B. Cotney, Jr., DERT-510080-CE, dated March 19, 2004.
- 2) There are no additional continuing airworthiness inspections required as a result of this modification. Inspection criteria and frequency for this item is contained in the existing Beechjet 400A airframe inspection program.
- 3) This modification has no effect on aircraft weight and balance.
- 4) Details of work performed are maintained on file at Rapid Aircraft Repair under work order 042771.

***** END *****

1 Additional Sheets Are Attached

SDNY_GM_02758371

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245293

EFTA01261953

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION			DATE
STATEMENT OF COMPLIANCE WITH THE FEDERAL AVIATION REGULATIONS			March 19, 2004
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION			
MAKE	MODEL NO.	TYPE (Airplane, Radio, Helicopter, etc.)	NAME OF APPLICANT
Raytheon Aircraft Co.	400A	Airplane	Flight Options, LLC Richmond Heights, OH 44143
LIST OF DATA			
IDENTIFICATION	TITLE		
Flight Options, LLC Dwg: EDU91-1 Rev: B Dated: 3/19/04	GRAPHIC FILM 220 APPLICATION		
Flight Options, LLC Dwg: EDU91-2 Rev: A Dated: 3/19/04	7-YEAR METALLIC SERIES FILM APPLICATION		
-----END-----			
NOTES: Note: 1) Approval for structural aspects of the above listed data only. 2) This approval is valid only for Raytheon Aircraft Co.400A S/N: RK-260 (N787TA)			
PURPOSE OF DATA To show compliance with the regulations listed below in support of exterior graphic film application. This is design data approval only, not installation approval.			
APPLICABLE REQUIREMENTS (List specific sections) FAR 25.601, 25.603, and 25.605			
CERTIFICATION - Under the authority vested by direction of the Administrator and in accordance with the conditions and limitations of appointment under Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <u>AS ABOVE</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Federal Aviation Regulations.			
I <input checked="" type="checkbox"/> Therefore <input type="checkbox"/> Recommend approval of these data <input checked="" type="checkbox"/> Approve these data			
SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	DESIGNATION NUMBERS(S)	CLASSIFICATION(S)	
William B. Cotney, Jr. <i>William B. Cotney</i>	DETR-510080-CE	-STRUCTURES-	

FAA Form 8110-3 (11-70) SUPERSEDES PREVIOUS EDITION

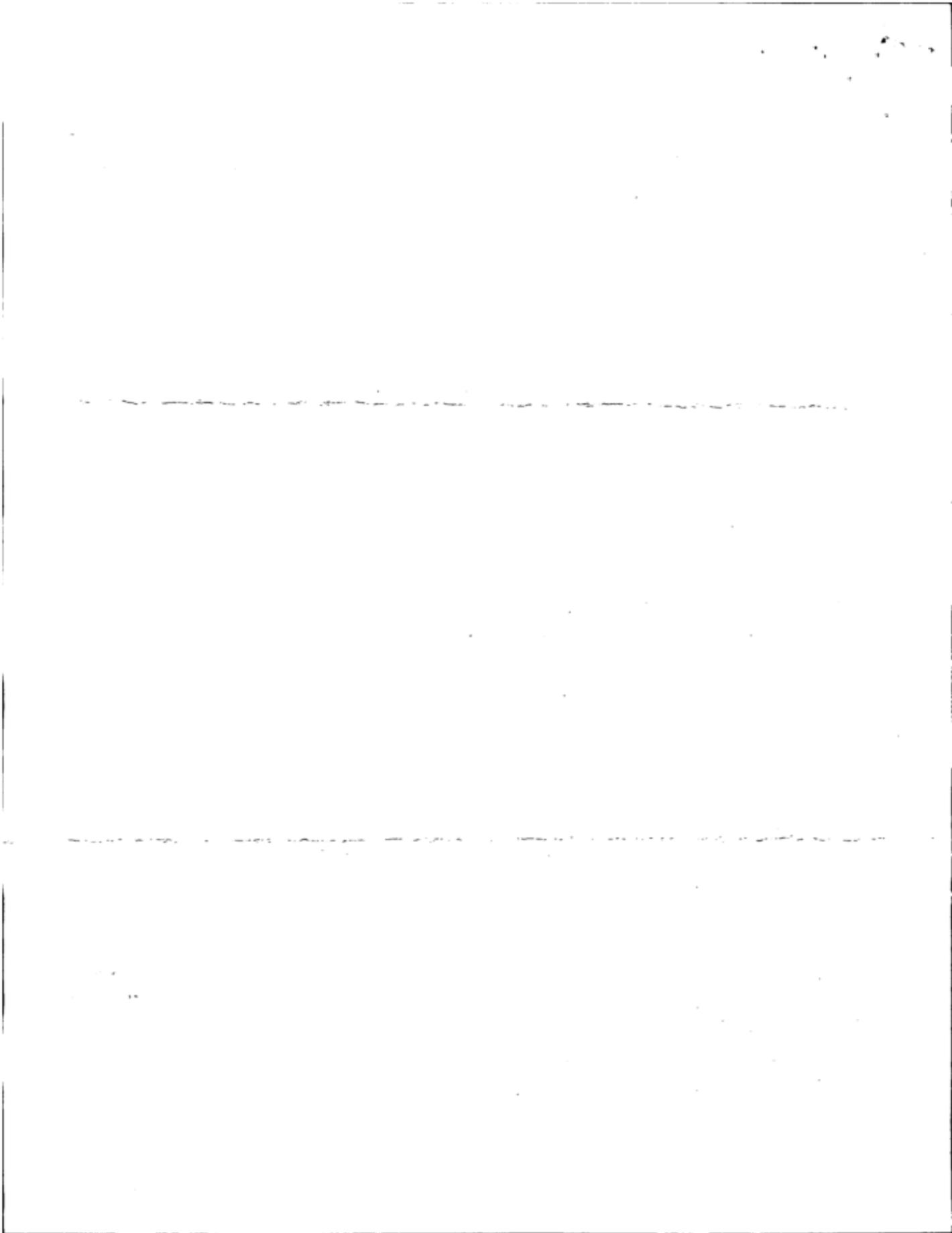
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SDNY_GM_02758372

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245294

EFTA01261954



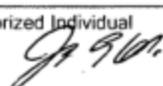
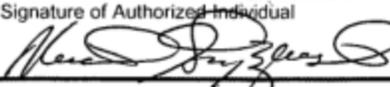
SDNY_GM_02758373

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245295

EFTA01261955

APR 01 2004

 <p>US Department of Federal Aviation Administration</p>		<p>MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)</p>		<p>Form Approved OMB No. 2120-0020</p>	
				<p>For FAA Use Only</p>	
				<p>Office Identification AEA-FSDO 23</p>	
<p>INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).</p>					
1. Aircraft	Make RAYTHEON		Model 400A		
	Serial No. RK-260		Nationality and Registration Mark USA N787TA		
2. Owner	Name (As shown on registration certificate) KIRK AIR LLC ET-AL		Address (As shown on registration certificate) C/O FLIGHT OPTIONS LLC 26180 CURTISS WRIGHT PARKWAY RICHMOND HEIGHTS OH 44143		
	3. For FAA Use Only				
4. Unit Identification					
Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	----- (As described in item 1 above) -----				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				
6. Conformity Statement					
A. Agency's Name and Address		B. Kind of Agency		C. Certificate No.	
JAY /EMERSON DAVIS C/O FLIGHT OPTIONS LLC 26180 CURTISS WRIGHT PARKWAY RICHMOND HEIGHTS, OH 44143		<input checked="" type="checkbox"/> U.S. Certificated Mechanic		AP 2788753	
		<input type="checkbox"/> Foreign Certificated Mechanic			
		<input type="checkbox"/> Certificated Repair Station			
		<input type="checkbox"/> Manufacturer			
<p>D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.</p>					
Date March 29, 2004		Signature of Authorized Individual 			
7. Approval for Return To Service					
<p>Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED</p>					
BY	FAA Fit. Standards Inspector	Manufacturer	Inspection Authorization	Other (Specify) FLIGHT OPTIONS LLC, AIR CARRIER CERTIFICATE	
	FAA Designee	Repair Station	Person Approved by Transport Canada Airworthiness Group		
Date of Approval or Rejection March 29, 2004		Certificate or Designation No. DJFA206D	Signature of Authorized Individual  KEITH GAWSYSZAWSKI		

FAA Form 337 (12-88)

SDNY_GM_02758374

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245296

EFTA01261956

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

INSTALLED ROCKWELL COLLINS, INC., SUPPLEMENTAL TYPE CERTIFICATE #ST00881WI-D, MODIFICATION OF COLLINS TCAS II SYSTEM INSTALLATION WITH COLLINS TTR-920 CHANGE 6.04a RECEIVER TRANSMITTER TO A COLLINS TCAS II SYSTEM INSTALLATION WITH COLLINS TTR-920 CHANGE 7 TCAS II RECEIVER TRANSMITTER IN ACCORDANCE WITH COLLINS INSTALLATION INSTRUCTIONS AS LISTED ON AML # ST00881WI-D.

AIRCRAFT FLIGHT MANUAL SUPPLEMENT INSERTED INTO AIRCRAFT FLIGHT MANUAL.

NO CHANGE TO WEIGHT AND BALANCE.

-----END-----

Additional Sheets Are Attached

*U.S.GPO:1994-568-012/00019

SDNY_GM_02758375

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245297

EFTA01261957

United States of America
Department of Transportation -- Federal Aviation Administration

Supplemental Type Certificate

Number ST00881W1-D

This certificate, issued to Rockwell Collins, Inc.
400 Collins Road NE, MS 164-100
Cedar Rapids, IA 52498

*certifies that the change in the type design for the following product with the limitations and conditions
therefor as specified hereon meets the airworthiness requirements of Part 21 of the
Regulations.*

Original Product - Type Certificate Number : * *See attached FAA Approved Model List (AML)
Make : * No. ST00881W1-D for list of approved airplane models
Model : * and applicable airworthiness regulations.

Description of Type Design Change: Modification of Collins TCAS II or TCAS 94 (TCAS II) system installation
with Collins TTR-920 Change 6.04a Receiver Transmitter to a Collins TCAS II or TCAS 94 (TCAS II) system
installation with Collins TTR-920 or Collins TTR-921 or Collins TTR-4000 Change 7.0 TCAS II Receiver Transmitter
and optional TPR-901 in accordance with Collins Installation Instructions as listed on AML No. ST00881W1-D,
Revision -, or later FAA approved revisions.

Limitations and Conditions: (1) Approval of this change in type design applies to the above model aircraft only.
(2) Compatibility of this design change with previously approved modifications must be determined by the installer.
(3) A copy of this Certificate and FAA Approved Model List (AML) No. ST00881W1-D issued June 30, 2000, or later
FAA approved revision, must be maintained as part of the permanent records for the modified aircraft. (4) Prior
installation and approval of a Collins TTR-920 Change 6.04a Receiver Transmitter is a prerequisite to this STC. (5) If
the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other
person written evidence of that permission.

*This certificate and the supporting data which is the basis for approval shall remain in effect until
surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the
Federal Aviation Administration*

Date of application : June 29, 2000

Date reissued :

Date of issuance : June 30, 2000

Date amended : July 14, 2000, December 1, 2000



By direction of the Administrator

John S. Bohm
(Signature)

John S. Bohm

Administrator, DAS-500864-CE

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2(10-99)

PAGE 1

This certificate may be transferred in accordance with FAR 21.47.

SDNY_GM_02758376

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245298

EFTA01261958

SDNY_GM_02758377

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245299

EFTA01261959



Avionics Certification Center
400 Collins Road NE



**Rockwell
Collins**

May 14, 2003

WO030737.DOC

Flight Options
26180 Curtiss-Wright Parkway
Cleveland, OH 44143
Tel: (216) 797-3500

Attention: Mr. Jody Kreamsreiter/
Mr. Richard Fine (Collins on-site)

Subject: STC ST00881WI-D Modification of Collins TCAS II System Installation with Collins TTR-920 Change 6.04a Receiver Transmitter and TPR-900 Mode S to a Collins TCAS II System Installation with Collins TTR-920, Collins TTR-921 or Collins TTR-4000 Change 7.0 TCAS II Receiver Transmitter and optional TPR-901 in Part 25 Aircraft

Dear Mr. Kreamsreiter:

Enclosed is a compact disk with one copy each of the following FAA approved STC data:

1. STC ST00881WI-D Certificate, dated December 1, 2000.
2. FAA Approved Model List (AML) Number ST00881WI-D, amended April 15, 2003.
3. Drawing Index 992-4358-001, Revision M.
4. One copy of each drawing listed on Drawing Index 992-4358-001.
5. Airplane Flight Manual Supplement, ACC-00-105, Revision -.

This letter is our authorization for Flight Options to use this STC as basis for the installation of our equipment in your airplanes based on the following model and serial numbers provided by Mr. Richard Fine of Rockwell Collins.

SDNY_GM_02758378

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245300

EFTA01261960

SDNY_GM_02758379

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245301

EFTA01261961

Flight Options
STC ST00881WI-D
May 14, 2003
Page 2

WO030737.DOC

STC ST00881WI-D

<u>Model</u>	<u>MSN</u>	<u>MSN</u>
400A	RK006	RK248
	RK016	RK252
	RK022	RK257
	RK027	RK260
	RK030	RK264
	RK031	RK265
	RK045	RK268
	RK062	RK271
	RK093	RK273
	RK146	RK274
	RK161	RK276
	RK168	RK279
	RK178	RK282
	RK180	RK284
	RK186	RK289
	RK189	RK292
	RK195	RK295
	RK198	RK297
	RK201	RK301
	RK202	RK305
	RK209	RK307
	RK222	RK310
	RK230	RK317
	RK234	RK324
	RK237	RK327
	RK239	RK328
	RK244	RK334
	RK245	

Please return a record of any additional aircraft serial number(s) to me for installations performed using this STC. My fax number is (319) 295-0337. Collins needs to receive this data before we can authorize additional installations.

SDNY_GM_02758380

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245302

EFTA01261962

SDNY_GM_02758381

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245303

EFTA01261963

Flight Options
STC ST00881WI-D
May 14, 2003
Page 3

WO030737.DOC

Flight Options is responsible for the regulatory approval of any deviations from STC ST00881WI-D.

Instructions explaining how to access the data on the CD have been inserted inside the jewel case.

Customer satisfaction is important to us, and we would appreciate your input. Please complete and return the enclosed questionnaire or email me at dwjessen@rockwellcollins.com and let us know how we can best serve you.

We trust this data will fulfill your installation requirements. However, if you have any questions, please call me at (319) 295-0618.

Sincerely,



Douglas W. Jessen
Collins Avionics Certification Center
Mail Station 164-100

/pah

Enclosure

SDNY_GM_02758382

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245304

EFTA01261964

SDNY_GM_02758383

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245305

EFTA01261965

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020

For FAA Use Only

Office Identification
DAE AEA-23

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Make Raytheon Aircraft Company	Model 400A
	Serial No. RK-260	Nationality and Registration Mark N787TA
2. Owner	Name (As shown on registration certificate) Grand/Sakwa Transportation LLC c/o Flight Options LLC	Address (As shown on registration certificate) 26180 Curtiss-Wright Parkway Richmond Heights, OH 44143

3. For FAA Use Only

4. Unit Identification

Unit	Make	Model	Serial No.	5. Type	
				Repair	Alteration
AIRFRAME	~~~~~ (As described in Item 1 above) ~~~~~			✓	
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement

A. Agency's Name and Address	B. Kind of Agency	C. Certificate No.
Rapid Aircraft Repair, Inc. R.R. #5, Peterborough Airport Peterborough, Ontario K9J 6X6 CANADA	<input type="checkbox"/> U.S. Certificated Mechanic	AMO 73-92
	<input type="checkbox"/> Foreign Certificated Mechanic	
	<input checked="" type="checkbox"/> Certificated Repair Station - Canadian	
	<input type="checkbox"/> Manufacturer	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date May 28, 2004	Signature of Authorized Individual im Pickett
----------------------	---

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit Standards Inspector	Manufacturer	Inspection Authorization	Other (Specify)
	FAA Designee	Repair Station	<input checked="" type="checkbox"/> Person Approved by Transport Canada Airworthiness Group	

Date of Approval or Rejection May 28, 2004	Certificate or Designation No. AMO 73-92	Signature of Authorized Individual im Pickett
---	---	---

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

REPAIR OF FUSELAGE CHAFF

- 1) Chaff mark in fuselage just aft of nose gear opening repaired by installation of doubler (2024T3 .040") per Raytheon Aircraft drawing SR-BJ-00015 and FAA form 8110-3 issued by Larry G. Young, DERY-230322-CE, dated July 3, 2003.
- 2) There are no additional continuing airworthiness inspections required as a result of this repair. Inspection criteria and frequency for this item is contained in the existing Beechjet 400A inspection program.
- 3) This repair has no effect on aircraft weight and balance.
- 4) Details of work performed are maintained on file at Rapid Aircraft Repair under work order 042772.

***** END *****

1 Additional Sheets Are Attached

SDNY_GM_02758385

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245307

EFTA01261967

U.S. DEPARTMENT OF TRANSPORTATION
 FEDERAL AVIATION ADMINISTRATION
 STATEMENT OF COMPLIANCE WITH THE FEDERAL AVIATION REGULATIONS

DATE
 July 3, 2003

AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION

MAKE MHI Beechcraft	MODEL NO. MU 300 400, 400A	TYPE (Airplane, Rotor, Helicopter, etc.) Airplane	NAME OF APPLICANT Raytheon Aircraft Company
------------------------	-------------------------------	--	--

Reference Letter No. 840-2003-07-066

LIST OF DATA

IDENTIFICATION

TITLE

SR-BJ-00015
 Rev. (-)

"Standard Repair- Chafing, Aft Nose Cabin Door on Fuselage Skin".

Notes:

1. This data approval is in support of and constitutes approval of the data listed herein and is not valid for any other purpose or application.
2. This approval is for Structural details only. This approval is only for the engineering design data.
3. Operator should obtain final approval from their local regulatory authority.

PURPOSE OF DATA To approve field repair.

APPLICABLE REQUIREMENTS (List specific sections)

14 CFR Part 25.25, 25.301, 25.303, 25.305, 25.307, 25.601, 25.603 and 25.605.

CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under Part 1X3 of the Federal Aviation Regulations, data listed above and on attached sheets numbered _____ have been examined in accordance with established procedures and found to comply with applicable requirements of the Federal Aviation Regulations.

I (We) Therefore Recommend approval of these data
 Approve these data

SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	DESIGNATION NUMBER(S)	CLASSIFICATION(S)
Larry G. Young <i>Larry Gene Young</i>	DERY-230322-CE	Structural

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

10-10-10

SDNY_GM_02758387

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245309

EFTA01261969



US Department of Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020

For FAA Use Only

Office Identification

JLL AEA-FSDO
23

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Make Beechcraft	Model 400A
	Serial No. RK-260	Nationality and Registration Mark N787TA
2. Owner	Name (As shown on registration certificate) Flight Options LLC	Address (As shown on registration certificate) 26180 Curtiss-Wright Pkwy Richmond Heights OH 44143-1494

3. For FAA Use Only
THE TECHNICAL DATA IDENTIFIED HEREIN HAS BEEN FOUND TO COMPLY WITH APPLICABLE AIRWORTHINESS REQUIREMENTS AND IS HEREBY APPROVED FOR USE ONLY ON THE ABOVE DESCRIBED AIRCRAFT, SUBJECT TO CONFORMITY INSPECTION BY A PERSON IN 14 CFR PART 43.7.

DATE 2/26/04 APPROVING INSPECTOR [Signature]
AEA FSDO 23

4. Unit Identification				5. Type	
Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	----- (As described in item 1 above) -----				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement		
A. Agency's Name and Address	B. Kind of Agency	C. Certificate No.
Lynn H. Huyen C/O Flight Options LLC 26180 Curtiss Wright Pkwy. Richmond Heights, OH 44143	<input checked="" type="checkbox"/> U.S. Certificated Mechanic	A&P 278825845
	<input type="checkbox"/> Foreign Certificated Mechanic	
	<input type="checkbox"/> Certificated Repair Station	
	<input type="checkbox"/> Manufacturer	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date February 24, 2004	Signature of Authorized Individual <u>[Signature]</u>
----------------------------------	--

7. Approval for Return To Service
Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit. Standards Inspector	Manufacturer	<input checked="" type="checkbox"/> Inspection Authorization	Other (Specify)
	FAA Designee	Repair Station	Person Approved by Transport Canada Airworthiness Group	
Date of Approval or Rejection 02-27-04		Certificate or Designation No. 105261968	Signature of Authorized Individual <u>[Signature]</u>	

FAA Form 337 (12-88)

SDNY_GM_02758388

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Installed the following equipment in Beechcraft 400A S/N RK-260:

Equipment	Install Manual	Maintenance Manual	Pilot's Guide
TAWS 8000 (Class B)	009-18001-001 Rev D Dated 20 March 2003	N/A	009-18000-001 Rev B Dated 11 June 2002

Installed Goodrich Avionics Systems Landmark TAWS8000 system (TSO-C151a Class B) using Flight Options LLC electrical drawing 400-3446-20-01 and applicable manufacturers installation manuals listed above. This installation includes the optional Terrain Display function.

FAA Approved Flight Manual Supplement 400-3446-20 must be on board the aircraft when the Class B TAWS 8000 system is in use.

The TAWS 8000 system has been previously approved for multiple Beechjet aircraft under STC ST01881CH-D, with the Terrain Display function enabled. This installation has been inspected and found to conform to the operational and technical characteristics of STC ST01881CH-D with minor deviations in location of switching relays and wiring interface points as depicted on Flight Options LLC drawing 400-3446-20-01. Similarity to STC ST01881CH-D is the basis for this request for field approval.

Using the respective manufacturer's instructions for post-installation ground tests, performed operational checks on all systems. The systems performed to manufacturer's specifications and no interference to existing aircraft systems was noted.

In addition to specific instructions contained in the manufacturer's installation manual(s) noted above, all physical mounting considerations were made with reference to AC 43.13-2A, Chapter 1, Chapter 2, Chapter 3, and Chapter 11 as they apply to this installation. Regarding attachment to additional structure all alterations were inspected to comply with the aircraft manufacturer's structural repair and alteration specifications.

In addition to specific instructions contained in the manufacturer's installation manual(s) noted above, wire bundles were fabricated using MIL-SPEC, Tefzel wire, that meets the acceptability requirements of AC43.13-1B, Chapter 11, Section 7, Paragraph 11-85 (a) and/or (b), paragraph 11-88, and paragraph 11-89. All wiring was marked in accordance with AC43.13-1B, Chapter 11, Section 16, Para. 11-206, 11-207, 11-208 (a) and/or (b) and 11-210 (a) and (b). Individual wires were formed into bundles by lacing and tying using the techniques described in AC43.13-1B, Chapter 11, Section 12, paragraph 11-158 (a) and (b). Cables were routed and secured in the aircraft using AC43.13-1B, Chapter 11, Section 8, Para. 11-96, (a) through (gg) as a guide.

An electrical load analysis was performed; the electrical load does not exceed 75% of the rated generator capacity. All compasses aboard this aircraft were checked for accuracy on completion of this installation. The aircraft will be weighed on completion of the interior modifications and new data will be entered into the Flight Manual. The aircraft equipment list and weight and balance were updated.

INSTRUCTIONS FOR CONTINUED AIRWORTHINESS IAW HBAW 98-18 DATED 10-07-98:

Instructions for continued airworthiness (ICA) for this aircraft alteration and interfacing components are as follows;

- (1) Introduction to the aircraft altered is explained above.
- (2) Description of the major alteration is explained above.
- (3) Operation information is described in the documents listed above for each particular product.
- (4) Servicing, (8) Diagrams and access plates, (9) Special inspection requirements, (10) Application of protective treatments, (11) Data, (12) List of Special Tools, (14) Recommended Overhaul Periods, and (15) Airworthiness limitation section are not applicable.
- (5) Maintenance, (6) troubleshooting information, (7) Removal and Replacement, and (13) for commuter category aircraft of the products listed, will be IAW the appropriate and current manufacturers installation manuals.
- (16) Revision will be in accordance with the manufacturers' maintenance manuals and submitted to the local FSDO.

Antennas and other parts and materials installed such as wiring, circuit breakers, switches, annunciators, clamps, doublers, shelves, and racks will be inspected in conjunction with the Raytheon Beechjet 400A Phase C inspection program, for condition, security of attachment, evidence of damage, and normal operation IAW FAR 25.1529 and/or applicable manufacturers service instructions.

Additional Sheets Are Attached

*U.S.GPO:1994-568-012/00019

SDNY_GM_02758389

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245311

EFTA01261971



US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
For FAA Use Only
Office Identification
5017 TWH

INSTRUCTIONS: Print or type all entries. See FAR 43.9 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Make Raytheon Aircraft Company	Model 400A
	Serial No. RK-260	Nationality and Registration Mark N787TA
2. Owner	Name (As shown on registration certificate) Bloomfield Investment, Wilero LLC, Maka of Turtland, East Penn Manufacturing, Et-Al, Flight Options LLC	Address (As shown on registration certificate) 26180 Curtiss Wright Parkway Richmond Heights OH 44143

3. For FAA Use Only

4. Unit Identification				5. Type	
Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	----- (As described in Item 1 above) -----				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement

A. Agency's Name and Address Pedro Vachier, 1017 Big Torch Street West Palm Beach, Fl 33407		B. Kind of Agency <input checked="" type="checkbox"/> U.S. Certified Mechanic <input type="checkbox"/> Foreign Certified Mechanic <input type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Manufacturer	C. Certificate No. A&P 582392447
D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.			
Date 11/07/03	Signature of Authorized Individual 		

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit. Standards Inspector	Manufacturer	<input checked="" type="checkbox"/> Inspection Authorization	Other (Specify)
	FAA Designee	Repair Station	Person Approved by Transport Canada Airworthiness Group	
Date of Approval or Rejection 11/07/03	Certificate or Designation No. 48463004 IA	Signature of Authorized Individual 		

FAA Form 337 (12-88)

SDNY_GM_02758390

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245312

EFTA01261972

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets, identify with aircraft nationality and registration mark and date work accomplished.)

N787TA , Beechjet 400A, S/N RK-260, TT 3247.1, 11/07/2003

Installed an Artex ELT-110-4 ELT, S/N 70130 with mount and remote switch in cockpit I/A/W Raytheon Aircraft DWG 128-340597, DWG 128-340582 and Beechjet 400A maintenance manual chapter 25-61-00-201 using installation kit # 455-7004. Used existing Raytheon wiring provisions.

Instructions for Continued Airworthiness are contained in the Beechjet 400 A maintenance manual.

Revised the weight and balance / equipment list and inserted into the flight manual.

-----END-----

Additional Sheets Are Attached

© U.S. G.P.O. 1990-761-753

SDNY_GM_02758391

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245313

EFTA01261973

 <p style="text-align: center;">MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)</p>		Form Approved OMB No. 2120-0020			
		For FAA Use Only			
		Office Identification <i>SW17 PA</i>			
<p>INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-I (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).</p>					
1. Aircraft	Make RAYTHEON AIRCRAFT Serial No. RK-260	Model 400A Nationality and Registration Mark USA N787TA			
2. Owner	Name (As shown on registration certificate) RAYTHEON TRAVEL AIR COMPANY, FLY AWAY LLC, BLOOMFIELD INVESTMENT COMPANY LLC, WILERO LLC, MAKAF OF TURRTLELAND L.L.C., EAST PENN MANUFACTURING CO. INC, FC CORPORATE AIR TRAVEL, INC, SAMOLOT LLC, ALCON INC, PILGRIM COVE AIR LLC, LEONARD M. RAND AND BARBARA E. RAND, AIR GHISLAINE INC, NASSAU HOLDINGS INC, BERGEN INDUSTRIES INC.	Address (As shown on registration certificate) 101 SOUTH WEBB ROAD WICHITA, KS 67207			
3. For FAA Use Only					
4. Unit Identification		5. Type			
Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	(As described in Item 1 above)				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				
6. Conformity Statement					
A. Agency's Name and Address		B. Kind of Agency		C. Certificate No.	
RAYTHEON AIRCRAFT SERVICES 1115 PAUL WILKINS ROAD SAN ANTONIO TEXAS 78216		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station Manufacturer		CRS-XA14605K LIMITED AIRFRAME	
D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Date 02-25-02		Signature of Authorized Individual MONTY CARROLL <i>Monty Carroll</i>			
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Flt. Standards Inspector	Manufacturer	Inspection Authorization	Other (Specify)	
	FAA Designee	Repair Station	Person Approved by Transport Canada Airworthiness Group		
Date of Approval or Rejection 02-25-02		Certificate or Designation No. CRS-XA14605K	Signature of Authorized Individual MONTY CARROLL <i>Monty Carroll</i>		

FAA Form 337 (12-88)

SDNY_GM_02758392

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245314

EFTA01261974

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

MAKE: RAYTHEON 400A S/N: RK-260 REG: N787TA DATE: 02-25-02

ALL SEVEN CABIN SEATS AND THE LAVATORY SEAT WERE FIREBLOCKED IN ACCORDANCE WITH FAR 25.853 (a) APPENDIX F PART I (a) (1) (ii) AND FAR 25.853 (c) APPENDIX F PART II AMENDMENT 25-83. THE FIREBLOCKING WAS ACCOMPLISHED IN ACCORDANCE WITH THE SKANDIA INC. TEST PLAN #8497 REV. A DATED 02-25-02 FOR P/N's 8497-1 AND 8497-2 ONLY, REFERENCE THE FORM 8110-3 DATED 02-25-02 APPROVED BY DONNA J. PARRISH, DESIGNATED ENGINEERING REPRESENTATIVE, #DERY-410100-CE, STRUCTURAL SPECIAL.

-----END-----

Additional Sheets Are Attached

★ U.S. GOVERNMENT PRINTING OFFICE: 1992-769-012/60157

SDNY_GM_02758393

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245315

EFTA01261975

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)				Form Approved OMB No. 2120-0020	
					For FAA Use Only	
					Office Identification <i>Sw17 M</i>	
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).						
1. Aircraft	Make RAYTHEON AIRCRAFT			Model 400A		
	Serial No. RK-260			Nationality and Registration Mark USA N787TA		
2. Owner	Name (As shown on registration certificate) RAYTHEON TRAVEL AIR COMPANY, FLY AWAY LLC, BLOOMFIELD INVESTMENT COMPANY LLC, WILERO LLC, MAKA OF TURRTLELAND L.L.C., EAST PENN MANUFACTURING CO. INC, FC CORPORATE AIR TRAVEL, INC, SAMOLOT LLC, ALCON INC, PILGRM COVE AIR LLC, LEONARD M. RAND AND BARBARA E. RAND, AIR GHISLAINE INC, NASSAU HOLDINGS INC, BERGEN INDUSTRIES INC.			Address (As shown on registration certificate) 101 SOUTH WEBB ROAD WICHITA, KS 67207		
3. For FAA Use Only						
4. Unit Identification				5. Type		
Unit	Make	Model	Serial No.	Repair	Alteration	
AIRFRAME	(As described in Item 1 above)				X	
POWERPLANT						
PROPELLER						
APPLIANCE	Type					
	Manufacturer					
6. Conformity Statement						
A. Agency's Name and Address RAYTHEON AIRCRAFT SERVICES 1115 PAUL WILKINS ROAD SAN ANTONIO TEXAS 78216			B. Kind of Agency <input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Manufacturer		C. Certificate No. CRS-XA14605K LIMITED AIRFRAME RADIO CLASS I,II,III	
D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.						
Date 02-25-02			Signature of Authorized Individual MONTY CARROLL <i>Monty Carroll</i>			
7. Approval for Return to Service						
Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED						
BY	FAA Fit. Standards Inspector		Manufacturer		Inspection Authorization	Other (Specify)
	FAA Designee	X	Repair Station		Person Approved by Transport Canada Airworthiness Group	
Date of Approval or Rejection 02-25-02			Certificate or Designation No. CRS-XA14605K		Signature of Authorized Individual MONTY CARROLL <i>Monty Carroll</i>	

FAA Form 337 (12-88)

SDNY_GM_02758394

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245316

EFTA01261976

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

MAKE: RAYTHEON 400A S/N: RK-260 REG: N787TA DATE: 02-25-02

MODIFIED THE EXISTING RIGHT HAND FORWARD GALLEY/CLOSET CABINET IN ACCORDANCE WITH THE RAYTHEON AIRCRAFT SERVICES DWG. #528-2526-001 REV. A DATED 07-27-00, REFERENCE THE FORM 8110-3 DATED 02-22-02 APPROVED KAMALA J. MEADER, DESIGNATED ENGINEERING REPRESENTATIVE, #DERT-405126-CE, STRUCTURAL. THE MODIFICATION CONSISTED OF INSTALLING ELECTRICAL PROVISIONS FOR A MAPCO, CUP HOLDERS AND LIGHTING IN THE FORWARD SECTION OF THE BAGGAGE CABINET. THE ELECTRICAL WIRING FOR THE GALLEY/CLOSET CABINET WAS INSTALLED IN ACCORDANCE WITH THE RAYTHEON AIRCRAFT SERVICES DWG. #RAS-73-0203 REV. IR DATED 02-22-02, REFERENCE THE FORM 8110-3 DATED 02-22-02 APPROVED BY ROBERT M. HURLEY, DESIGNATED ENGINEERING REPRESENTATIVE, #DERT-710137-SW, SYSTEMS AND EQUIPMENT. THE WEIGHT AND BALANCE FORM AND THE EQUIPMENT LIST HAVE BEEN REVISED AND BOTH FORMS HAVE BEEN INSERTED IN THE AIRCRAFT FLIGHT MANUAL.

-----END-----

Additional Sheets Are Attached

★ U.S. GOVERNMENT PRINTING OFFICE: 1992-769-012/60157

SDNY_GM_02758395

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245317

EFTA01261977

 U.S. Department of Transportation Federal Aviation Administration	MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)			Form Approved OMB No. 2120-0020	
	For FAA Use Only				
	Office Identification 3417 <i>h</i>				
INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).					
1. Aircraft	Make	RAYTHEON AIRCRAFT		Model	400A
	Serial No.	RK-260		Nationality and Registration Mark	USA N787TA
2. Owner	Name (As shown on registration certificate)			Address (As shown on registration certificate)	
	RAYTHEON TRAVEL AIR COMPANY, FLY AWAY LLC, BLOOMFIELD INVESTMENT COMPANY LLC, WILERO LLC, MAKA OF TURRTLELAND L.L.C., EAST PENN MANUFACTURING CO. INC, FC CORPORATE AIR TRAVEL, INC, SAMOLOT LLC, ALCON INC, PILGRIM COVE AIR LLC, LEONARD M. RAND AND BARBARA E. RAND, AIR GHISLAINE INC, NASSAU HOLDINGS INC, BERGEN INDUSTRIES INC.			101 SOUTH WEBB ROAD WICHITA, KS 67207	
3. For FAA Use Only					
4. Unit Identification					
				5. Type	
Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	(As described in Item 1 above)				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				
6. Conformity Statement					
A. Agency's Name and Address		B. Kind of Agency		C. Certificate No.	
RAYTHEON AIRCRAFT SERVICES 1115 PAUL WILKINS ROAD SAN ANTONIO TEXAS 78216		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Manufacturer		CRS-XA14605K LIMITED AIRFRAME	
D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
Date		Signature of Authorized Individual			
02-25-02		MONTY CARROLL <i>Monty Carroll</i>			
7. Approval for Return to Service					
Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Flt. Standards Inspector	Manufacturer	Inspection Authorization	Other (Specify)	
	FAA Designee	X Repair Station	Person Approved by Transport Canada Airworthiness Group		
Date of Approval or Rejection		Certificate or Designation No.	Signature of Authorized Individual		
02-25-02		CRS-XA14605K	MONTY CARROLL <i>Monty Carroll</i>		

FAA Form 337 (12-88)

SDNY_GM_02758396

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245318

EFTA01261978

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

MAKE: RAYTHEON 400A S/N: RK-260 REG: N787TA DATE: 02-25-02

FABRICATED AND INSTALLED A NEW THE LIFE VEST CONTAINER IN THE TOILET CABINET IN ACCORDANCE WITH THE RAYTHEON AIRCRAFT SERVICES DWG. #562-3832-001 REV. IR DATED 08-22-00, REFERENCE THE FORM 8110-3 DATED 02-22-02 APPROVED BY KAMALA J. MEADER, DESIGNATED ENGINEERING REPRESENTATIVE, #DERT-405126-CE, STRUCTURAL. THE LIFE VEST CONTAINER WAS FABRICATED USING AAR COMPOSITES FLAT NOMEX PANEL P/N ATR-FP-251F2, REFERENCE THE FORM 8110-3 DATED 07-01-99 APPROVED BY JUDY BOGGS, DESIGNATED ENGINEERING REPRESENTATIVE, #CHI-410, STRUCTURAL SPECIAL. THE MODIFICATION OF THE CONTAINER WAS FOR THE INSTALLATION OF A NEW EASTERN AERO MARINE LIFE VEST, P/N PO1074-113. THE WEIGHT CHANGE IS NEGLIGIBLE. THE EQUIPMENT LIST HAS BEEN REVISED TO REFLECT THE NEW LIFE VEST AND HAS BEEN INSERTED IN THE AIRCRAFT FLIGHT MANUAL.

-----END-----

Additional Sheets Are Attached

★ U.S. GOVERNMENT PRINTING OFFICE: 1992-769-012/60157

SDNY_GM_02758397

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245319

EFTA01261979



US Department of Transportation
Federal Aviation Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
For FAA Use Only
Office Identification
File F800 on SAW

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each violation (Section 901 of Federal Aviation Act of 1958).

1. Aircraft	Make Raytheon	Model 400A
	Serial No. RK-260	Nationality and Registration Mark N787TA
2. Owner	Name (As shown on registration certificate) See Attached Copy (2 Pages)	Address (As shown on registration certificate) 101 S. Webb Rd. Wichita, KS 67201

3. For FAA Use Only

4. Unit Identification

5. Type

Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	----- (As described in Item 1 above) -----				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

RECEIVED
 DO NOT WRITE
 WICHITA FSDO
 JUN 8 11:05 AM '00

6. Conformity Statement

A. Agency's Name and Address Kenneth W. Chitty 101 S. Webb P.O. Box 2902 Wichita, Ks 67201-2902	B. Kind of Agency <input checked="" type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input type="checkbox"/> Certified Repair Station <input type="checkbox"/> Manufacturer	C. Certificate No. 454944057
---	---	--

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date 03-06-00	Signature of Authorized Individual <i>Kenneth W. Chitty</i>
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7. Approval for Return To Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Flt. Standards Inspector	Manufacturer	<input checked="" type="checkbox"/>	Inspection Authorization	Other (Specify)
	FAA Designee	Repair Station		Person Approved by Transport Canada Airworthiness Group	
Date of Approval or Rejection 03-06-00	Certificate or Designation No. 454944057	Signature of Authorized Individual <i>Kenneth W. Chitty</i>			

FAA Form 337 (12-88)

SDNY_GM_02758398

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245320

EFTA01261980



MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020

For FAA Use Only

Office Identification
SW FSDO 17 *BRR*

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Make RAYTHEON	Model 400A
	Serial No. RK-260	Nationality and Registration Mark USA N787TA
2. Owner	Name (As shown on registration certificate) RAYTHEON AIRCRAFT COMPANY	Address (As shown on registration certificate) 9709 E CENTRAL AVE WICHITA, KANSAS 67206-2507

3. For FAA Use Only

4. Unit Identification				5. Type	
Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	***** (As described in Item 1 above) *****				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement

A. Agency's Name and Address RAYTHEON AIRCRAFT SERVICES 1115 PAUL WILKINS ROAD SAN ANTONIO TEXAS 78216	B. Kind of Agency <input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Manufacturer	C. Certificate No. CRS-XA14605K LIMITED AIRFRAME RADIO CLASS I,II,III
---	--	--

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date 01-28-00	Signature of Authorized Individual MONTY CARROLL <i>Monty Carroll</i>
------------------	--

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit. Standards Inspector	Manufacturer	Inspection Authorization	Other (Specify)
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Person Approved by Transport Canada Airworthiness Group	
Date of Approval or Rejection 01-28-00		Certificate or Designation No. CRS-XA14605K	Signature of Authorized Individual MONTY CARROLL <i>Monty Carroll</i>	

FAA Form 337 (12-88)

SDNY_GM_02758400

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245322

EFTA01261982

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

MAKE: RAYTHEON 400A S/N: RK-260 REG: N787TA DATE: 01-28-00

INSTALLED A SUNDSTRAND MK-VI GROUND PROXIMITY WARNING SYSTEM CONSISTING OF THE FOLLOWING EQUIPMENT:

MOD/MFG	DESCRIPTION	P/N	WEIGHT	ARM
MK-VI	GPWS COMPUTER	965-0686-020	4.2	33.10
ADC-500	CONVERTER	500-8020	2.25	38.17
	BRACKET & PLATE		0.5	36.14
	WIRING & CONNECTOR		3.5	54.0

THE SUNDSTRAND MK-VI GPWS SYSTEM WAS INSTALLED IN ACCORDANCE WITH THE STC #SA3322SO ISSUED 09-18-92 AND REISSUED 10-04-95 AND THE HANGER ONE MASTER DRAWING LIST #11892-GPWS, REV. D DATED 03-22-93. THE ELECTRICAL WIRING WAS INSTALLED IN ACCORDANCE WITH THE RAYTHEON AIRCRAFT SERVICES DWG. #RAS-67-9915 REV. IR DATED 09-29-99, REFERENCE THE FORM 8110-3 DATED 12-30-99 APPROVED BY ROBERT M. HURLEY, DESIGNATED ENGINEERING REPRESENTATIVE #DERT-710137-SW, SYSTEMS AND EQUIPMENT.

A FUNCTIONAL TEST OF THE EQUIPMENT HAS BEEN PERFORMED IN ACCORDANCE WITH FAR 25.1301, FAR 25.1309 AND CHECKED IN ACCORDANCE WITH FAR 25.1431 FOR OPERATING SATISFACTORILY AND DID NOT ADVERSELY AFFECT OTHER COMPONENTS INSTALLED IN THE AIRCRAFT.

THE SUNDSTRAND MK-VI GPWS APPROVED FLIGHT MANUAL SUPPLEMENT, REV. B DATED 03-30-93 HAS BEEN INSERTED IN THE SUPPLEMENT SECTION OF THE AIRCRAFT FLIGHT MANUAL AND THE GPWS PILOT'S GUIDE P/N 060-4087-000 REV. A DATED 10-96 HAS BEEN INSERTED IN THE AIRCRAFT AS PART OF THE AIRCRAFT'S PERMANENT RECORDS.

THE AIRCRAFT WEIGHT AND BALANCE FORM AND THE EQUIPMENT LIST WERE REVISED TO REFLECT THIS MODIFICATION AND BOTH FORMS WERE INSERTED IN THE AIRCRAFT FLIGHT MANUAL.

-----END-----

Additional Sheets Are Attached

★ U.S. GOVERNMENT PRINTING OFFICE: 1992-769-012/60157

SDNY_GM_02758401

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245323

EFTA01261983

United States of America
Department of Transportation — Federal Aviation Administration
Supplemental Type Certificate

Number SA3322SO

This certificate, issued to Raytheon Aircraft Services, Inc.
3992 Aviation Circle
Fulton County Airport-Brown Field
Atlanta, Georgia 30336

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 25 of the Federal Aviation Regulations

Original Product — Type Certificate Number: A16SW
Make: Beech
Model: 400A

Description of Type Design Change: Installation of a Sundstrand MK VI GPWS in accordance with Jangar One. Drawing List No. 11892-GPWS, Rev. B, dated September 17, 1992, or later FAA Approved revision.

Limitations and Conditions: This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications will produce no adverse effect upon the airworthiness of that aircraft. FAA Approved Airplane Flight Manual Supplement, dated September 18, 1992, is a required part of this STC.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration

Date of application: August 5, 1992

Date issued: October 4, 1995

Date of issuance: September 18, 1992

Date amended:



By direction of the Administrator

David P. West (Signature)

Associate Manager, ACE-116A

Atlanta Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

FAA Form 0110-2 (10-68)

SDNY_GM_02758402

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245324

EFTA01261984

SDNY_GM_02758403

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245325

EFTA01261985

SDNY_GM_02758405

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245327

EFTA01261987

SDNY_GM_02758407

SUBJECT TO PROTECTIVE ORDER PARAGRAPHS 7, 8, 9, 10, 15, and 17

EFTA_00245329

EFTA01261989